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Environmental Assessment for the Satellite Power System- Concept Development and Evaluation Program- Atmospheric Effects

November 1980

Prepared for:
U.S. Department of Energy
Office of Energy Research
Solar Power Satellite Projects Division
Under Contract No. 31-109-ENG -38

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DOE/NASA
Satellite Power System
Concept Development
and
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Argonne National Laboratory
Argonne, IL 60439
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EXECUTIVE SUMMARY

This document constitutes the final technical report of the Atmospheric Effects Assessment Program that has been conducted under the auspices of the Satellite Power Systems (SPS) Project Office of the U.S. Department of Energy as a part of the SPS Concept Development and Evaluation Program.

The objectives of the Atmospheric Effects task were to identify potentially significant effects, along with the possible mitigating strategies, of SPS-related operations on the entire Earth's atmosphere from the ground to geosynchronous orbit (GEO); to predict the nature, extent, and import of those effects within program constraints; and to assess the reliability of those predictions.

Programmatic constraints (program duration and funding levels) have limited the assessment, with a few exceptions, to use of archived data and existing assessment techniques, including mathematical/computer simulation models. However, some additional basic scientific data were acquired through measurement programs associated with certain targets of opportunity. These data have helped resolve some of the important questions that arose during the course of the assessment. Also, as a result of comparisons between theoretical model simulations and both the archived and newly acquired data, some basic improvements were made in our understanding of and ability to predict SPS-related effects.

An additional programmatic constraint was that the assessment was focused on a Reference System description of the SPS. This description, while adequate for identification and assessment of potentially significant atmospheric effects, is certain to undergo considerable change if such a system eventually is built. Selection of a specific launch site(s) and a more complete specification of the final operations system design would undoubtedly require some updating of the present atmospheric effects assessment. With the exception of some example calculations making use of the Kennedy Space Flight Center meteorological and air quality conditions, all the analyses in the present assessment are independent of launch site.

The initial task of identifying potential atmospheric effects was undertaken in a series of workshops held in 1978. The proceedings of those workshops and a follow-up workshop in 1979 provided fairly extensive lists of possible effects, and summarized our state of knowledge at that time regarding the effects and their impacts on man's environment. Recommendations for both near- and long-term research needs and approaches also resulted from those workshops. The general finding of the workshops was that essentially every level of the Earth's atmosphere is potentially affected by the construction and operation of the SPS. The principal SPS activities and their associated potential effects were initially identified as follows:

1. Rectenna waste heat releases and changes in the albedo and surface roughness of the terrain could lead to local weather modification.
2. Rocket effluents injected into the troposphere, especially near ground level, could lead to air quality degradation and weather modification.

3. Rocket effluents injected into the upper atmosphere could result in climatic changes, ozone depletion, ionospheric depletion, enhanced airglow emissions, structural changes in the plasmasphere and magnetosphere, including Van Allen Belt modifications and changes in response of the Earth's atmosphere to magnetic storms, and changes in atmospheric properties that could affect the performance of telecommunications, navigation, and remote sensing systems and other space-borne facilities.
4. SPS stationkeeping, attitude control, and emissions from space structures could cause an alteration of the satellite environment.

Noise-related effects are dealt with in the Nonmicrowave Effects task.

In addition to elucidating these cause-effect relationships, an important function of the Atmospheric Effects task has been to provide basic information on changes in atmospheric properties to other program task areas that are responsible for evaluating the ultimate impacts on health, ecosystems, telecommunications, etc. On the other hand, in the case of ionospheric heating due to absorption of energy from the microwave power beam, the Ionospheric Heating task is responsible for providing basic data that serve as input to the Atmospheric Effects task's assessment of possible climatic or other atmospheric effects outside the beam cross section.

The assessed atmospheric effects are summarized below:

Heavy lift launch vehicle (HLLV) launches have been predicted to have a significant potential for inadvertent weather modifications on a local, short-term scale. Under selected meteorological conditions such launches can affect convective patterns, alter cloud populations, and induce detectable precipitation. None of these effects are judged to be serious. Long-term, large-scale effects are unlikely, but have not been examined in detail.

Air quality impacts of HLLV launches are predicted to be very small except possibly with respect to nitrogen dioxide. If a short-term air quality standard is set as anticipated, then ground-level concentrations of nitrogen dioxide (NO₂) due to rocket launches could exacerbate existing problems. However, the launches by themselves are not expected to cause an exceedance of the expected standard. The production of NO₂ can also lead to slight increases in acidity of precipitation on a local, intermittent basis. It seems unlikely that the enhancement would be great enough to cause noticeable environmental effects.

Stratospheric carbon dioxide (CO₂) and water injections are estimated to be completely negligible from both the ozone depletion and greenhouse effect points of view. The fractional increase in the CO₂ concentration would be only 1-2 parts per 10⁵, an amount too small to affect the radiation balance. Small predicted reductions (0.01-0.02%) in the total ozone column (and therefore a corresponding increase in ultraviolet radiation at ground level) due to water injections would not be detectable.

The production of nitric oxide (NO) during reentry of the HLLV second stage is predicted theoretically to lead to an increase of about 0.02% in the

globally averaged ozone column density, with latitude-dependent variations of a factor of 2 around this value. The uncertainty in the calculations is such that no increase or even a small decrease is possible.

Repeated injections of water in the mesosphere would result in an increase of about 8% (globally averaged) in the water vapor content at 80 km. Over a narrow region surrounding the launch latitude at the same height, the increase may be as much as 15%. At lower altitudes, the buildup is smaller and no corridor effects are expected. At higher altitudes, preliminary estimates suggest that the buildup in water vapor content will be greater, increasing the globally averaged natural content by as much as a factor of 100 or more at 120 km and above. On the short-term, smaller scale basis associated with individual rocket launches, significant increases in vapor concentrations are also expected. On the same basis, small-scale clouds (noctilucent) are expected to be formed but are not expected to reach global scale and, therefore, are not expected to cause climatic impacts. Temperature effects, not fully understood at present, may alter the likelihood of cloud formation and possibly increase cloud persistence. Significant visual displays at low- to mid-latitudes may result near sunrise or sunset.

The injection of water vapor and hydrogen molecules above 80 km will lead to an enhancement in the upward flux of hydrogen through the thermosphere. It has been estimated that the natural total flux of hydrogen (in all chemical forms) will be approximately doubled. The consequences of such an enhancement have not been examined in detail as yet. Possible consequences include an increase in escape rate of hydrogen to outer space and/or an increase in atmospheric density above 800 km.

Several mechanisms that are not completely independent of each other have been estimated to produce both large positive and negative changes in D- and E-region plasma densities on a local, short-term scale. It is not clear how large or persistent the net plasma density changes would be or even what sign they would take. Long-term, global-scale D-region free electron density changes appear to be quite small and probably unimportant although the predictions are uncertain at this time.

In the F-region, however, the picture is much clearer. We can predict with considerable confidence the scale of, and with somewhat less confidence the duration of, short-term F-region holes caused by various engine burns of SPS-space vehicles. Each personnel orbit transfer vehicle (POTV) injection burn would result in an ionospheric hole two to three times as large as the continental United States. HLLV circularization burns would produce holes one-tenth the size. (Personnel orbit transfer vehicle and HLLV burns occur once or twice per month and once or twice per day, respectively.) More speculatively, the periodic engine burns may lead to a chronic partial depletion in a ring-shaped global region centered around the launch latitude. Crude estimates suggest a 10% plasma reduction in this region superimposed on periodic, small-scale, but much more complete, depletions. Confirmation with more detailed model calculations, taking into account global transport processes, is required. Probably the major consequences of this partially depleted ring would be perturbations of very low frequency (VLF), high frequency (HF), and possibly very high frequency (VHF) wave propagation. According to present information such perturbations are not expected to be large. However, uncertainties exist with the current data and theoretical

modeling studies and warrant considerable caution in attempting to scale up radio wave propagation effects. Another impact of potential importance is the enhancement of airglow emissions. The greatest enhancements observed to date are on the order of 10 kilorayleighs for certain emissions in the visible and near infrared band. The significance of these effects in terms of their impacts on astronomical observations, remote sensing and satellite-borne surveillance systems should be determined in the future, perhaps with the aid of the remote sensing and astronomy communities.

The effect of POTV and cargo orbit transfer vehicle (COTV) emissions in the magnetosphere is perhaps least well understood at present and least well supported by observations. This area of investigation could benefit by field and laboratory experiments and further improvements in theory. The masses and energies injected are unquestionably large compared to naturally occurring values and, therefore, cause concern. Heating by the argon plasma beam and the addition of cold, heavy ions may lead to enhancement of dosage of trapped relativistic electrons in the outer Van Allen Belt and perhaps in the GEO environment. Production of artificial ionospheric electric currents similar to those associated with naturally occurring magnetic storms could be driven by the magnetosphere-argon ion beam interactions. Such currents could trip circuit-breakers of long telephone and power lines and enhance pipeline corrosion. The presence of large quantities of neutrals (from the POTV) could lead to substantial depletion of energetic, trapped, charged particles and modification of auroral response to solar activity. The possibility also exists for the triggering of plasma instabilities that could cause satellite communications interference. Finally, an appreciable amount of airglow emission may be generated, especially near LEO, by the impingement of the dense argon ion beam on the thermosphere. Again, the import of the artificial airglow should be judged by the remote sensing and astronomy communities.

The sun-weather effect that has been widely debated in the recent climate literature, if it is real, is likely to involve a coupling of the upper and lower atmosphere. Several mechanisms have been proposed to explain such a coupling. It has been conjectured that if indeed modifications in the solar wind can influence climate through some such coupling scheme, then perhaps the SPS upper atmospheric effects that have been discussed herein could disturb the coupling scheme. In that sense, upper atmospheric SPS effects may be causally connected to possible climate effects. However, until the sun-weather effect is placed on a firmer footing and until the SPS upper atmospheric effects themselves are better understood, the potential effect of SPS on climate and weather cannot be evaluated except in terms of the direct tropospheric effects discussed earlier. Based on present knowledge, significant influence of SPS-induced upper atmospheric effects on tropospheric weather is regarded as unlikely.

One final effect that has been investigated in the current assessment is the inadvertent weather modification likely to be caused by the rectenna structure coupled with its release of waste heat. From the weather and climate points of view, the rectenna seems likely to have effects comparable with those due to other nonindustrial land-use changes covering the same surface area; that is, local, detectable meteorological effects that probably have little consequence.

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ENVIRONMENTAL ASSESSMENT FOR THE SATELLITE POWER SYSTEM CONCEPT
DEVELOPMENT AND EVALUATION PROGRAM: ATMOSPHERIC EFFECTS

by

Donald M. Rote, Kenneth L. Brubaker, Jiin-Lang Lee

ABSTRACT

The U.S. Department of Energy (DOE) has undertaken a preliminary, three-year program to investigate the impacts of the construction and operation of a satellite power system, of unprecedented scale. The Department of Energy's program, titled The Concept Development and Evaluation Program, focused its investigations on a Reference System description that calls for the use of either silicon (Si) or gallium aluminum-arsenide (GaAlAs) photovoltaic cells on 60 satellites to be constructed in GEO over a 30-yr period. Rectennas would be constructed on the ground to receive microwave energy from the satellites. Each satellite-rectenna pair is designed to produce 5 GW of power on an essentially continuous basis for use as a baseload power source for an electric power distribution system.

The environmental assessment part of the program was divided into five interdependent task areas. The present document constitutes the final technical report on one of the five task areas, the Assessment of the Atmospheric Effects, and as such presents an in-depth summary of work performed during the assessment program.

The issues associated with SPS activities in the troposphere are examined. These include tropospheric weather modification related to rectenna operations and rocket launches, and air quality impacts related to rocket-launch ground clouds. Then progressing upward through the various levels of the atmosphere, the principal middle and upper atmospheric effects associated with rocket effluents are analyzed. Finally, all of the potential SPS atmospheric effects are summarized.

1 INTRODUCTION

The unique properties of the Earth's space environment make it an attractive place for a broad spectrum of scientific, commercial, industrial, and military activities. The proposal, originally due to Glaser,¹ to use space as an alternative renewable energy source, as epitomized by the Satellite Power System (SPS), adds yet another dimension to this exploitation of space. The SPS would take advantage of the fact that a satellite placed in geosynchronous orbit (GEO) would be continuously (except for seasonally

dependent short occultation periods) exposed to unattenuated solar radiation while remaining stationary relative to a point below on the Earth's surface. The Reference System² calls for 60 satellite-rectenna pairs, an example of which is illustrated in Fig. 1. Each pair produces 5 GW of power on an essentially continuous basis for use as a baseload power source for an electric power distribution system. As shown in the figure each satellite continuously converts solar radiation into microwave radiation in GEO, then transmits it to Earth where it is received and converted to DC current by a rectenna.

The reference system calls for the use of either silicon (Si) or gallium aluminum-arsenide (GaAlAs) photovoltaic cells spread over a low mass-density structure about 5 x 10 km in area. The microwave transmission antenna is about 1 km in diameter and rotates relative to the solar cell array structure to permit simultaneous exposure to the sun and transmission to a fixed point (rectenna site) on the Earth's surface. For mid-latitude sites, the elliptically shaped rectenna has dimensions of about 10 x 13 km. The peak microwave power density at the rectenna center would be 230 W/m².

As is characteristic of solar powered systems designed to provide a substantial fraction of the U.S. total electrical energy demand (~20% by the time frame 2020-2040), the scale of the SPS and related resource requirements is very large. In addition, the SPS also has space transportation requirements of unprecedented proportions. The 30-year construction program calls for the completion of two 5-GW satellites per year. Corresponding space transportation system requirements for both photovoltaic cell options are outlined in Fig. 2. The number of flights does not include those required for maintenance operations, which could become substantial after several satellites have been in operation for a number of years. Some of the pertinent details of the four vehicle types composing the system, including their operating altitude ranges are listed in Table 1. The masses of the principal rocket exhaust products, as a function of altitude, are given in Table 2. Note that both in terms of size and launch frequency, the heavy lift launch vehicle (HLLV) dominates the system. It has a payload roughly five times that of the Saturn V vehicle used in the Apollo Program. Allowing for maintenance operations, one or two HLLVs would be launched per day. The trip from the ground to low Earth orbit (LEO) will involve the use of a Hohman transfer ellipse, in which the HLLV will climb to 124 km and then slightly lose altitude while gaining speed. Cutoff of the second stage occurs near 111 km, from which point the vehicle travels into an elliptic trajectory up to 477 km altitude, where a circularization burn puts it into parking orbit. After depositing its payload, the HLLV makes a de-orbit burn and returns to the surface for re-use. Reentry heating near 60-90 km will inject a significant quantity of nitric oxide (NO) into the atmosphere. The HLLV ascent trajectory is shown in Fig. 3, and exhaust emission data are presented in Table 3.

In view of the program scale as well as the potential costs and benefits, DOE, in cooperation with NASA, decided to undertake a three-year program to investigate the technical feasibility, economic viability, and social, political, institutional, and environmental impacts of the SPS option and the comparison of those impacts with other energy options. The Concept Development and Evaluation Program, as it became known, focused these investigations, among others, on the Reference System design² cited above. The environmental assessment part of that program was divided into the following five interdependent major task areas:

Solar Power Satellite
FUELED BY THE SUN'S RAYS

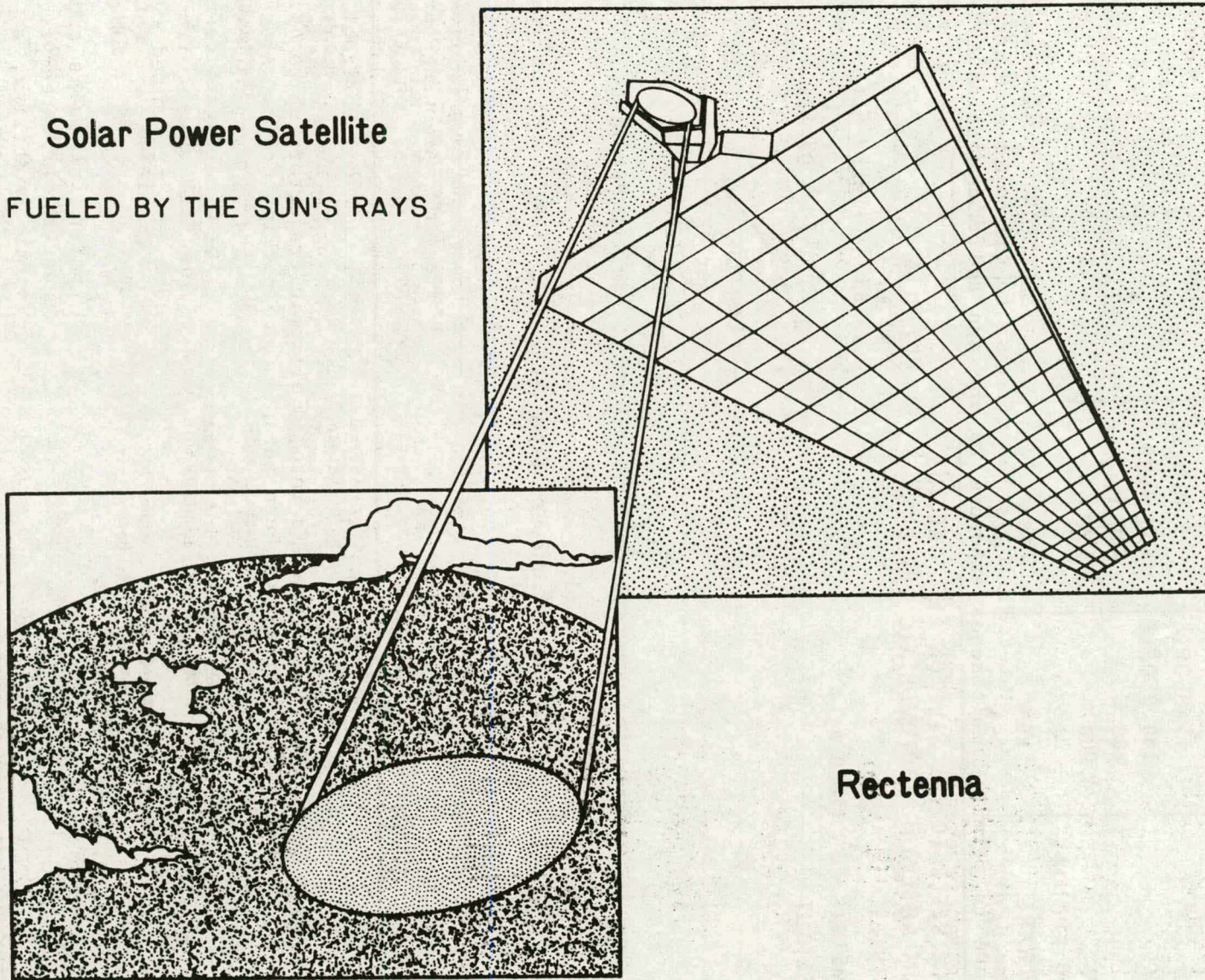


Fig. 1 SPS Satellite in Geosynchronous Earth Orbit and Surface Rectenna

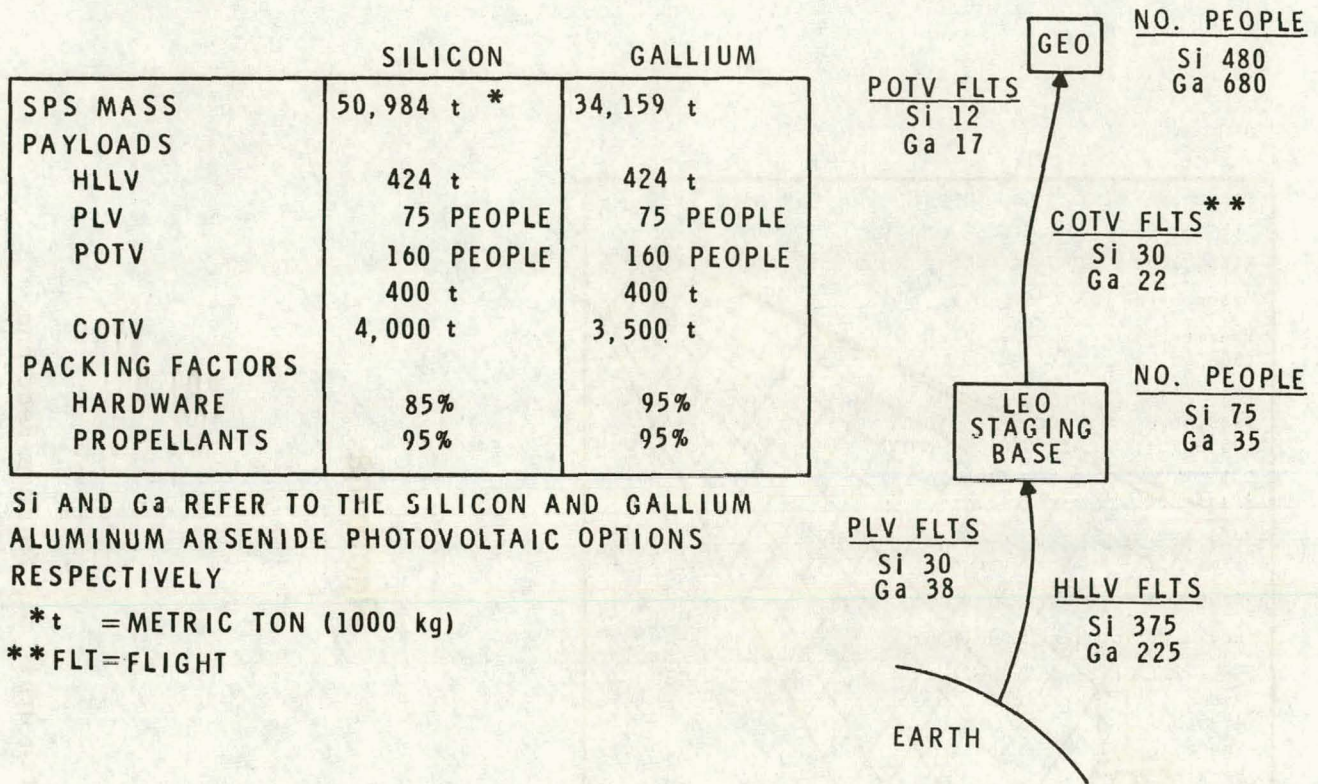


Fig. 2 Scenario for Construction of Two 5-GW Satellites Per Year

Table 1 SPS Space Transportation Vehicles

Name	Abbreviation	Function	Propellants	Launch ^a Rate (year ⁻¹)	Operating Altitude (km)	Main Exhaust Products
Heavy-lift launch vehicle	HLLV	Transport of materials	CH ₄ /O ₂ (Stage 1)	375	0-57	CO ₂ ,H ₂ O
			H ₂ /O ₂ (Stage 2)	375	57-120	H ₂ O,H ₂
			H ₂ /O ₂ (circular ization/deorbit)	375	450-500	H ₂ O,H ₂
Personnel launch vehicle	PLV	Transport of personnel earth surface to LEO	Details not available (probably same as HLLV)	30	0-500	CO ₂ ,H ₂ O,H ₂
Cargo-orbit	COTV	Transport of materials LEO to GEO	Argon H ₂ /O ₂	30	500-35,800	Ar ion plasma H ₂ O,H ₂
Personnel-orbit transfer vehicle	POTV	Transport of personnel LEO to GEO	H ₂ /O ₂	12	500-35,800	H ₂ O,H ₂

^aAssuming construction of two (silicon option) 5-GW satellites/year.

Table 2 Exhaust Products of SPS Transportation Vehicles^a

Atmospheric Region	Altitude Range (km)	Source ^b	Total Mass (kg)	Mass (kg) of Specific Emission Products				
				CO ₂	CO	H ₂ O	H ₂	Ar
Troposphere (0-13 km)	0-0.5	HLLV, PLV	6.50 (5) ^c	2.60 (5)	1.17 (5)	2.60 (5)	1.30 (4)	-----
	0.5-13	HLLV, PLV	2.85 (6)	1.14 (6)	5.13 (5)	1.14 (6)	5.70 (4)	-----
Stratosphere	13-50	HLLV, PLV	3.03 (6)	1.21 (6)	5.46 (5)	1.21 (6)	6.06 (4)	-----
Mesosphere	50-80	HLLV, PLV	7.58 (5)	1.99 (5)	8.96 (4)	4.50 (5)	1.91 (4)	-----
Thermosphere (80-500 km)	80-125	HLLV, PLV	2.03 (6)	--	--	1.96 (6)	7.10 (4)	-----
	477 ^d	HLLV, PLV	3.29 (4)	--	--	3.17 (4)	1.15 (3)	-----
	477	POTV	4.60 (5)	--	--	4.43 (5)	1.12 (4)	-----
Exosphere (>500 km)	GEO ^d	POTV	1.53 (5)	--	--	1.47 (5)	4.69 (3)	-----
	477-GEO	COTV ^e	9.85 (5)	--	--	--	--	9.85 (5)

^aMass emissions per flight.

^bPLV emissions are chemically similar to those of the HLLV, but are not otherwise determined at this time. The numbers shown are emissions of the HLLV only.

^c6.50 (5) denotes 6.50×10^5

^dLow Earth orbit (LEO) is at 477 km, geosynchronous orbit (GEO) is at 35,800 km.

^eIn addition to mass emissions, the argon plasma engines of the COTV will inject a significant amount of energy into this altitude range. Also, argon plasma engines will be used for satellite altitude control at GEO; these emissions are unknown at present and have not been included. The mass of COTV propellant given is for the Si option. For the GaAlAs option, the Ar⁺ mass would be 2.12 (5).

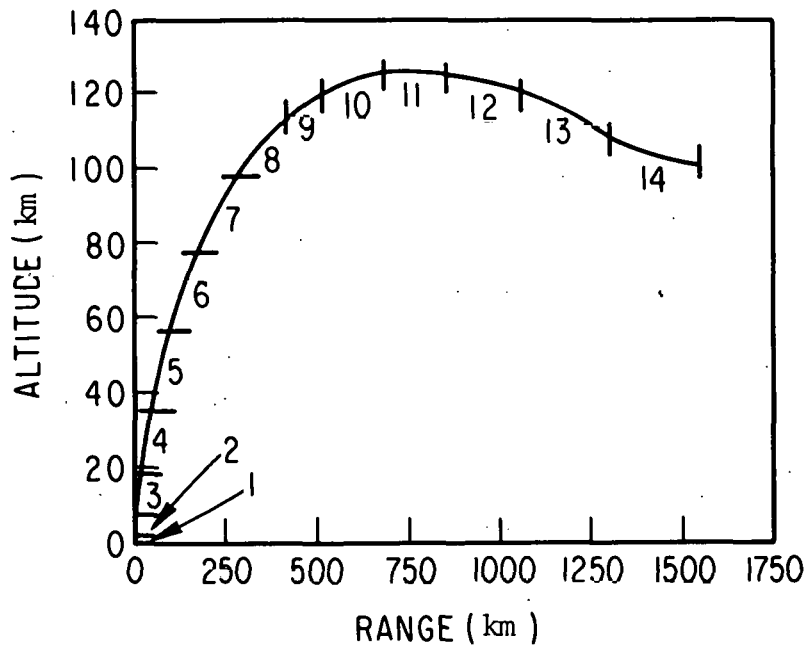


Fig. 3 HLLV Main Burn Trajectory

Table 3 HLLV Main Burn Exhaust Emissions

Step	Time (sec)	Δ (Altitude) (km)	Δ (Range) (km)	Emissions (10^3 kg)			
				H ₂ O	H ₂	CO ₂	CO
1	30.8	1.71	0.00	553.2	27.7	553.2	249.0
2	29.2	6.00	2.09	524.5	26.2	524.5	236.0
3	30.0	11.29	12.24	538.9	26.9	538.9	242.5
4	30.0	15.73	25.02	538.9	26.9	538.9	242.5
5	35.2	22.06	57.26	577.3	28.9	577.3	259.8
6	34.0	20.64	77.43	219.9	8.0	0	0
7	40.0	19.15	99.55	253.1	9.2	0	0
8	40.0	14.20	116.4	253.1	9.2	0	0
9	40.0	9.25	133.2	253.1	9.2	0	0
10	40.0	4.33	153.6	253.1	9.2	0	0
11	40.0	-0.32	177.7	253.1	9.2	0	0
12	40.0	-4.02	205.2	253.1	9.2	0	0
13	40.0	-5.90	238.0	253.1	9.2	0	0
14	35.5	-3.15	244.6	217.1	7.9	0	0

1. Microwave Effects on Health and Ecosystems
2. Nonmicrowave Effects on Health and Ecosystems
3. Atmospheric Effects
4. Effects of Ionospheric Heating on Telecommunications.
5. Electromagnetic Systems Compatibility

The initial assessment covering these five areas was published by DOE in October 1978.^{3,4} The first revision to that preliminary environmental assessment report was subsequently prepared in the summer of 1979 and appeared in print in January of 1980.^{5,6} The second and final revision of that report is published in six volumes: one for each of the five tasks listed above and one that provides an overview of the whole environmental assessment. This report covers Task 3.

The objectives of the Atmospheric Assessment task area are to identify potentially significant effects of SPS-related operations on the entire Earth's atmosphere from the ground to GEO; to predict the nature, extent and significance of those effects within program constraints; and to assess the reliability of those predictions. If effects are found that are judged to be significant, an additional objective is to identify possible mitigating strategies.

Programmatic constraints (program duration and funding levels) have limited the assessment (with a few exceptions) to use of archived data and existing assessment techniques, including mathematical/computer simulation models. Some additional basic scientific data have been acquired through measurement programs associated with certain targets of opportunity. This data has helped resolve some of the important questions that have arisen during the course of the assessment. Moreover, as a result of comparisons between theoretical model simulations and both archived and newly acquired data, some basic improvements have been made in our understanding and ability to predict SPS-related effects.

Every level of the Earth's atmosphere is potentially affected by the construction and operation of the SPS. The principal SPS activities and their associated potential effects were initially identified as:

1. Rectenna waste heat releases and changes in the albedo and surface roughness of the terrain could lead to local weather modification.
2. Rocket effluents injected into the troposphere, especially near ground level, could lead to air quality degradation and weather modification.
3. Rocket effluents injected into the upper atmosphere could result in climatic changes, partial ozone depletion, partial ionospheric depletion, enhanced airglow emissions, structural changes in the plasmasphere and magnetosphere, including Van Allen Belt modifications and changes in the response of the Earth's atmosphere to magnetic storms, as well as changes in atmospheric properties that could affect the performance of telecommunications, navigation, and remote sensing systems.
4. SPS stationkeeping, attitude control and emissions from space structures could cause an alteration of the satellite environment.

In addition to elucidating these cause-effect relationships, an important function of Task 3 has been to provide basic information on changes in atmospheric properties to other task areas that are responsible for evaluating the ultimate impacts on health, ecosystems, telecommunications, etc. On the other hand, in the case of ionospheric heating due to absorption of energy from the microwave power beam, Task 4 is responsible for providing basic data that serves as input to Task 3's assessment of possible climatic or other atmospheric effects outside the beam cross section.

The initial task of identifying potential atmospheric effects was undertaken in a series of workshops held in 1978. The proceedings of these workshops⁷⁻¹⁰ and a follow-up workshop in 1979¹¹ provide fairly extensive lists of possible effects, and summarize our state of knowledge at that time regarding the effects and their impacts on man's environment. Recommendations for both near- and long-term research needs and approaches also resulted from these workshops.

The issues associated with SPS activities in the troposphere, including inadvertent weather modification there, through rectenna operations and rocket launches, are taken up first, followed by a discussion of air quality impacts associated with rocket launch ground clouds. Then, progressing upward through the various levels of the atmosphere illustrated in Fig. 4, the principal middle and upper atmospheric effects associated with rocket effluents are addressed. For convenience, all of the potential SPS atmospheric effects are summarized in the conclusions.

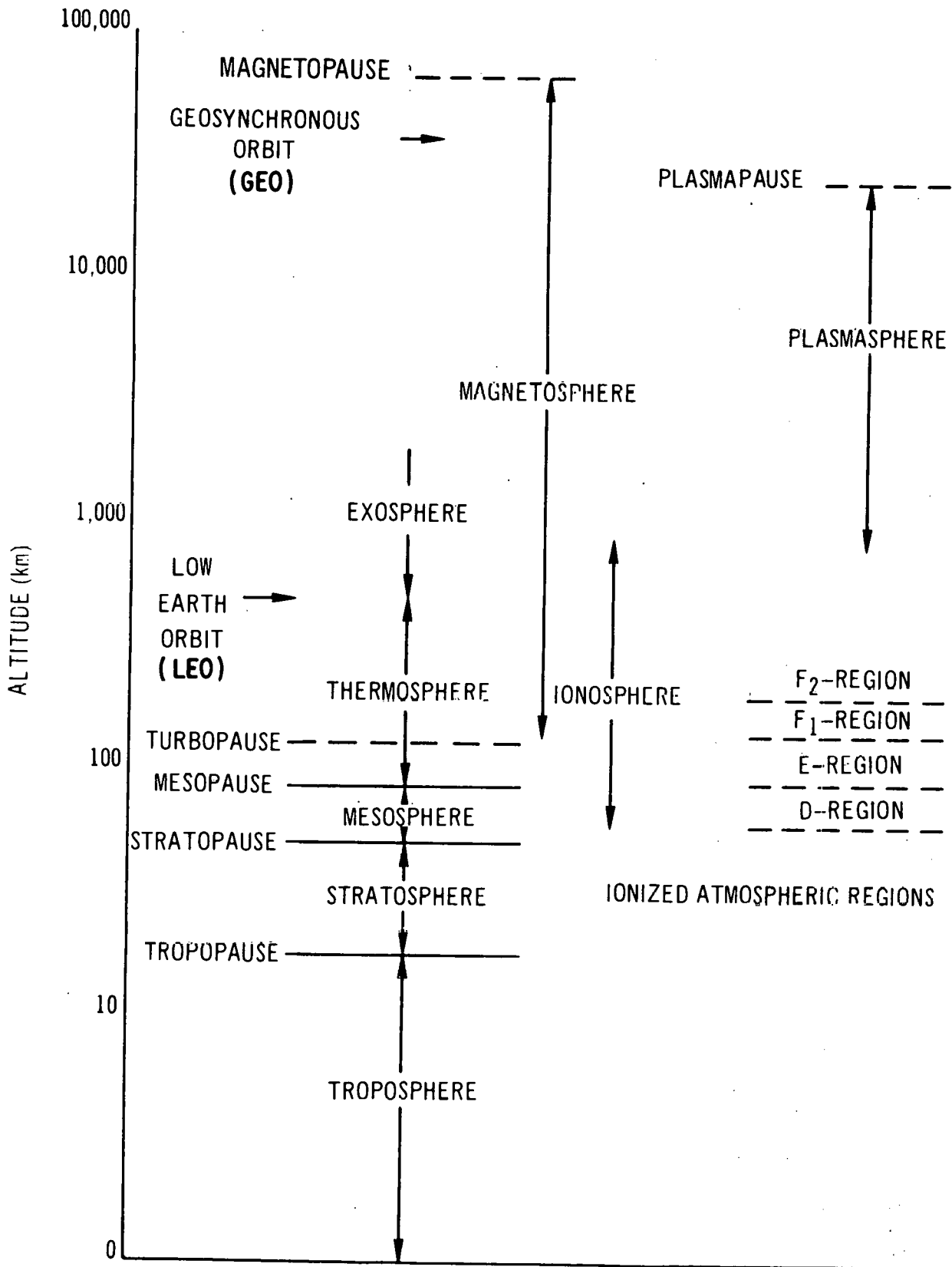


Fig. 4 Regions of the Atmosphere

2 ASSESSMENT OF LOWER ATMOSPHERIC EFFECTS

The term "lower atmospheric effects" refers here to the SPS-related effects occurring in the troposphere, the lowest region of the atmosphere, corresponding to altitudes below about 10 km at high latitudes and 16-17 km near the equator. Tropospheric effects are of particular interest, since all plant and animal life, including human beings, are susceptible to changes in this region.

The composition of the tropospheric atmosphere is one of its most important characteristics from the environmental point of view. The major constituents are molecular nitrogen, molecular oxygen, and argon, comprising 78.08, 20.95 and 0.93% by volume of dry air, respectively; these percentages change only over geologic periods of time, and may be taken as constant for most purposes. The remaining fraction consists primarily of a variable and steadily increasing quantity of carbon dioxide currently approaching 335 ppmv (parts per million by volume),¹² trace (i.e., part-per-million level or less) quantities of other inert gases, methane, and substances such as ozone and various oxides of nitrogen that are generally regarded as air pollutants. In addition to these components of dry air, the atmosphere contains a variable amount of water vapor, ranging up to several percent (water also appears in the atmosphere in its liquid form as fog, clouds, and rain, and in solid form as snow, hail, and ice crystals). As to weather processes, water is by far the most important constituent of the atmosphere. Large quantities of energy are released or absorbed during condensation and evaporation, with profound consequences for atmospheric phenomena. Water vapor, like carbon dioxide, also affects atmospheric radiation properties, and accordingly helps govern the thermal conditions in the atmosphere.

The planetary boundary layer comprises the lowest 1-2 km of the troposphere, the area within which friction between the Earth's surface and the atmosphere significantly affects atmospheric motions induced by large-scale pressure variations and by the diurnal solar heating cycle. The wind and temperature structure in the planetary boundary layer determine the meteorological conditions in which most human activities take place, and its depth and turbulent characteristics affect the ground-level concentrations of various pollutants emitted into the atmosphere by these same human activities.

The construction and operation of a satellite power system (SPS) will directly affect the troposphere in several ways, the most important of which are the effect of the SPS rectennas on local meteorological variables, the effect of SPS-related rocket launches on cloud formation and weather in the vicinity of the launch site, and the effect of SPS-related rocket launches on air quality and the acidity of precipitation.

The presence of a rectenna covering an area of approximately 100 km² would be expected to alter the aerodynamic characteristics of the surface in its immediate vicinity. A change in surface roughness affects the vertical fluxes of momentum and thermal energy, and a change in radiative properties (albedo for solar radiation and emissivity for terrestrial radiation) affects the surface energy budget. The operation of a rectenna would also add an additional heat source at the surface. The primary consequences of these changes could be to alter the wind velocity profile and stability of the planetary boundary layer and hence to alter the local cloud population. These effects are considered in Sec. 2.1.

The possible influence of microwave transmission through the troposphere would arise from the absorption, especially in clouds, of microwave energy along the beam path, causing local heating. On the other hand, the presence of convective or turbulent air motions and the existence of hydrometeors* cause refraction, scattering, and absorption of microwave energy and can lead to beam wandering and spreading. (Many issues of concern regarding the effects of atmospheric conditions on SPS beam propagation can be found in the proceedings of a workshop held in August 1978,⁷ and are not discussed here).

The proposed heavy lift launch vehicle (HLLV) and personnel launch vehicle (PLV) would emit a large amount of exhaust effluent into the atmospheric boundary layer. These rocket effluents may affect the local environment of the launch site in several ways, all related to the development and subsequent evolution of the rocket ground cloud. The ground cloud consists of the exhaust emitted by the rocket during the first 15-25 s following ignition and liftoff, together with a large quantity of entrained air, cooling water, dust, and other debris. After the rocket has accelerated to an appreciable velocity, the exhaust products are deposited in a thin column that disperses quickly in the ambient air and would have no significant adverse environmental effects.

Immediately after formation, the ground cloud, due to the buoyant effect of its high thermal energy content, rises to an altitude of between several hundred and several thousand meters, depending upon ambient meteorological conditions. As the cloud rises, much of the surface dust and debris falls out, the distance over which the fallout occurs being determined by the wind speed, by the nature of the turbulence within the cloud, and by the size of the particles. This distance may be as great as a few kilometers. At the same time, the upward convective motion of the ground cloud and the surrounding air may result in the formation of a water-saturated cloud (a phenomenon observed after several previous rocket launches) or even a convective storm. In addition, cloud microphysical processes may be affected by the production in the rocket exhaust of both cloud condensation nuclei (CCN) and ice-forming nuclei (IN), and by the entrainment of giant nuclei during lift-off. The meteorological effects of such a ground cloud are discussed in Sec. 2.2.

The ground cloud also represents a source of air pollution and associated effects. The cloud disperses over a period of time, the rate of dispersion being determined by the level of turbulence both in the cloud itself and in the ambient atmosphere. Depending on the chemical nature of the rocket exhaust products, adverse environmental effects may or may not be produced at ground level. The Reference System design² for both the HLLV and PLV calls for the use of liquid methane and liquid oxygen in the first stage, instead of a solid propellant such as is used in most current rockets. A major exhaust product of solid-fueled rockets is hydrochloric acid. In contrast, the major exhaust products of the HLLV and PLV boosters are carbon dioxide and water. Smaller quantities of nitrogen oxides, primarily nitric oxide and nitrogen dioxide, are expected to be produced from a possible molecular nitrogen impurity in the liquid oxygen, from a nitrogen impurity in

*Cloud water droplets, raindrops, hail, ice crystals, etc.

the fuel, or from entrainment and heating of ambient air in the hot rocket exhaust. In addition, possible impurities, such as sulfur, in the fuel would give rise to a corresponding amount of oxidation products, such as sulfur dioxide. As described below, sulfur dioxide has been measured in at least one ground cloud because of a sulfur impurity in the hydrocarbon fuels currently in use. Thus, for liquid-fueled rockets like the HLLV and PLV, the potential air quality impacts must arise from substances present in relatively small amounts, the principal exhaust products being environmentally innocuous. The exhaust products of a solid-fueled booster would cause much greater environmental problems. Two types of air quality effects are considered in Sec. 2.3. One, is the direct increase of the atmospheric concentration of gaseous air pollutants; the other, is the effect of rocket exhaust constituents on the acidity of precipitation. The potential tropospheric effects of rocket launches are illustrated in Fig. 5. Many of the effects shown were discussed in a workshop held in September 1978.⁸

2.1 INADVERTENT WEATHER MODIFICATION CAUSED BY RECTENNA STRUCTURE AND WASTE HEAT RELEASE

Human activities have altered the surface of the Earth long before the present era.¹³ The first major change started when man developed agriculture. Agriculture and lumbering have led to mesoscale climatic changes

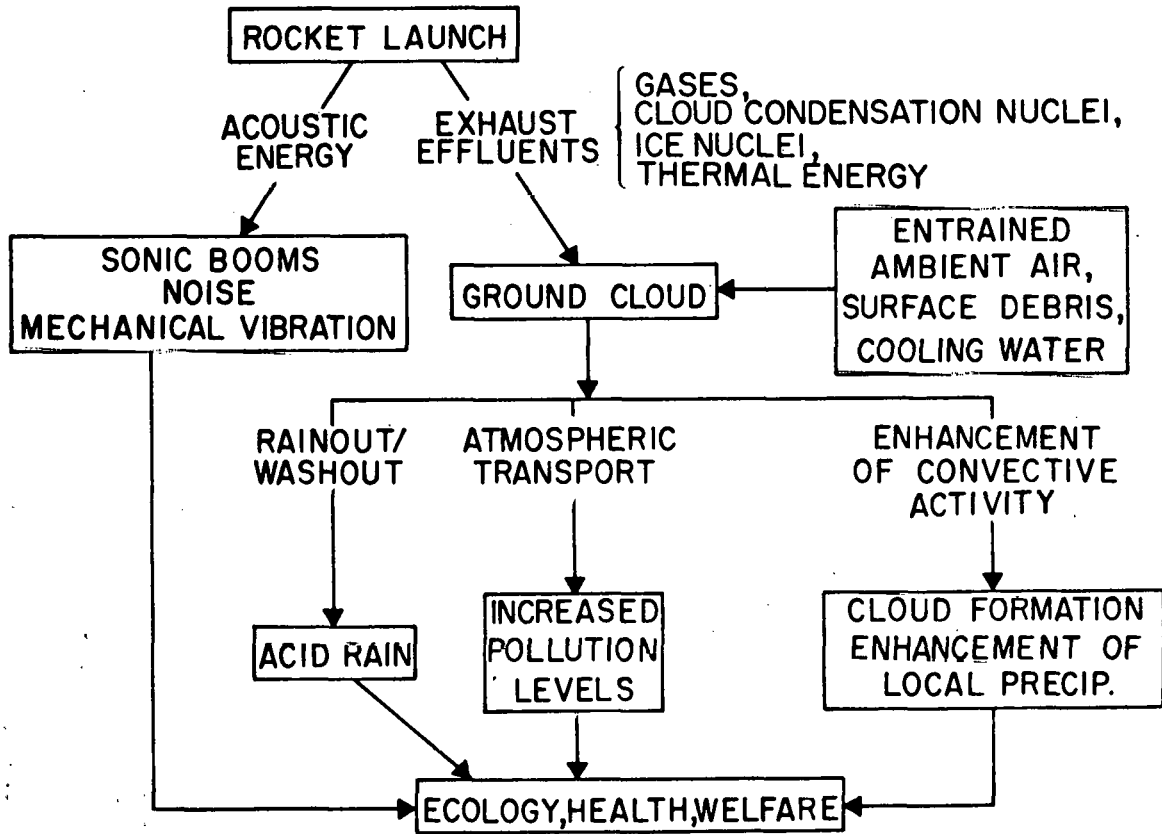


Fig. 5 Potential Tropospheric Effects of Rocket Launches

owing to the alterations of surface albedo. However, by far the most noticeable and well-documented human impact on local weather and climate is that due to the process of urbanization. Temperature and precipitation anomalies have been observed within and downwind, respectively, of large cities.^{14,15,16}

Huff et al.,¹⁷ and Hanna and Gifford¹⁸ presented a comparison of the energy release rates associated with both natural phenomena and man-made features; some of these values are reproduced in Table 4. Based on energy considerations, it appears highly unlikely that large cyclones will be affected in their overall behavior by man's works. However, large cities, having the same spatial scales and energy release rates as thunderstorms, and large islands in the ocean do produce measurable downwind effects on rainfall, storm, and hail frequencies, etc.¹⁵ Weather and climate modifications caused by large cities are largely dominated by anthropogenic features, while modifications by large islands are basically affected by natural causes. A 5-GW conventional steam electric generation plant releases approximately 10 GW of waste heat to the atmosphere, a rate close to that of a city, but the area of release is much smaller (i.e., of much higher power density). Increases in cloudiness and convective activity, precipitation augmentation, fogging, icing, and a long visible plume often result. The waste heat of the proposed SPS rectenna is less than 10% of that of conventional steam electric generation with an area of dissipation comparable to the scale of a "heat island." It is anticipated on this basis that the rectenna waste heat term in the thermal energy balance at the Earth's surface would have negligible meteorological effects.

A preliminary assessment of the possible meteorological effects from SPS rectenna operation was conducted in 1977 at the Lyndon Johnson Space Center.¹⁹ The assessment was based upon the maximum power density in the microwave beam of about 230 W/m² and an average of 7.5 W/m² of waste heat release at a rectenna, covering about 100 km². The findings were that the

Table 4 Comparison of Man-Made and Natural Energy Releases

Energy Source	Area (km ²)	Duration	Approx. Power (CW) input to Atmosphere
St. Louis	250	continuous	16
Chicago	1800	continuous	53
Aruba Island	180	daytime	35
Surtsey volcano	1	month	100
Thunderstorm (2 cm of rain/hr)	100	hour	100
One-megaton nuclear device		dissipated over 1 hr	10 ³
Great Lake snow squall (4 cm of snow/hr)	10 ⁴	hour	10 ⁴
Cyclone (1 cm of rain/day)	10 ⁶	day	10 ⁵
5-GW steam electric Power Plant	1-10	continuous	10
SPS rectenna waste heat	100	continuous	0.75

effects of an SPS rectenna on weather and climate would be small, compared to the direct environmental consequences of construction, and that the rectenna's influence would be similar to that of an average suburban development. The intensity of the atmospheric perturbation resulting from rectenna operation should be very small compared with that of other man-made installations. Microwave heating of the lower atmosphere through gaseous absorption would be negligible. Any actual effects of microwave heating inside a cloud would not be detected in the presence of the natural variance of cloud and storm phenomena. Scattering by particles, even in a heavily-polluted atmosphere, would also be negligible.

The above study was reviewed at the August 1978 workshop⁷ and the conclusions were updated. Three main topics were discussed: the effects of waste-heat release on the atmosphere at the rectenna site, microwave interactions with the atmosphere, and the possible effects of the microwave beam on atmospheric electrification processes. The following brief summaries highlight the most important issues.

Rectenna Waste Heat and Structure

Construction of a rectenna would modify the thermal and radiative properties of the ground on which it is built; operations would introduce a heat source at the surface. Although the magnitude of the perturbation of the average surface heat budget would be on the order of 10%, microwave beam wandering and spreading due to atmospheric refraction may occasionally give rise to larger effects. (However, according to the Reference System design,² beam wandering would be prevented by the beam phase control system.)

It is possible to investigate the effects of the rectenna by studying the effects of land-use changes. Small temperature changes (of the order of 1°C) can be expected under light wind conditions. Changes in cloud populations can also be expected. Somewhat larger man-made dissipation rates over comparable areas have been associated with apparent anomalies in the distribution of rainfall.

In hilly terrain, on scales smaller than the rectenna dimensions, diurnally varying changes occur in the surface energy budget that are larger than the projected rectenna waste heat. It is expected therefore that the meteorological effects of a rectenna would vary from site to site, and that the central maximum heat dissipation (approximately 16 W/m²) might become important in augmenting a naturally occurring topographic effect.

Assessment of possible weather and climate effects over areas larger than the mesoscale should not be confined to the influence of the rectenna alone -- the whole satellite power system must be considered in the context of the energy demand it is designed to meet. The overriding feature of the system is that the major inefficiency, the rejection of waste

heat, occurs in space. Furthermore, there are no significant emissions of material into the troposphere during operation. These aspects are obvious benefits.

Microwave Propagation

The atmospheric absorption of microwave energy at the proposed SPS frequency is negligible in clear air for the projected tropospheric path lengths of about 20 km. However, some absorption by condensed water (clouds and precipitation) would occur when storms entered the beam path.

Atmospheric Electricity

Direct interactions with the atmospheric electrical fields are not thought to be important at the proposed frequency. However, the mere physical presence of the rectenna might have some modifying influence on the occurrence and electrical behavior of thunderstorms over and around the rectenna.

To support these conclusions and to carry out recommendations made during the 1978 workshop, some detailed model calculations were conducted. As mentioned earlier, the rectenna may influence local weather and climate through several mechanisms, namely, the release of waste heat from power conversion, and the alteration of surface roughness and the change in radiative properties from the rectenna structure. More important, the degree of influence depends strongly upon meteorological conditions. Therefore, in the assessment of rectenna effects, at least four basic variables should be considered in principle in this stage of evaluation; they are: type of site, weather situation, system operation conditions, and rectenna structure.

A realistic simulation has been carried out for 24 hr for a potentially unstable boundary layer with light wind over moist, flat terrain, using a limited-area, primitive-equation, meteorological model.²⁰ The calculations were performed for sun positions appropriate for 21 June over an area of 39 x 33 km with and without a rectenna in normal operation. A 3-km square grid was used with eight sub-surface levels down to a depth of 0.8 m and 18 atmospheric levels up to a height of 2000 m. The site was assumed flat with surface roughness equivalent to long grass, and at two-thirds of its saturated moisture capacity. The atmospheric lapse rate was approximately moist adiabatic. The physical constants of the soil and initial meteorological conditions were based on observations at O'Neill, Nebraska, in May 1953 (Great Plains Project). The surface albedo was taken to be 0.18, and longwave emissivity 1.0.

The rectenna structure was taken to be 12 m high with a 46° slope, appropriate to 40° N. latitude. The surface roughness within the rectenna boundary was taken to be equivalent to that of a forest. The microwave beam power level distribution was based on the Reference System report.² The total waste heat was 423 MW, 81 absorbed in the top soil layer and 342 distributed uniformly over the lowest 16 m above grade. At the present time, the detailed

radiative properties of the rectenna structure are unknown. The site albedo was assumed to be unchanged by the structure. An approximate analysis indicated that rectenna albedo change is potentially the source of larger perturbations of surface heat balance than is the waste heat, but that practicable adjustments to the surface finish could greatly modify these perturbations. On the other hand, perturbations due to waste heat and roughness change are relatively uncontrollable. It was decided, therefore, to concentrate attention on the effects of waste heat and roughness change.

The results of the calculations showed that at 1200 hr (6 hr after start of simulation), the temperature at 16 m is slightly lowered at all points in the presence of the rectenna. The maximum change is 0.2°C in the center of the rectenna, and there is a small temperature reduction ($\sim 0.05^{\circ}\text{C}$) outside the rectenna area. The absolute humidity at 16 m is higher by 0.2 to 0.25 g/kg (grams of water per kilogram of dry air) within the rectenna area and 0.05 to 0.1 g/kg outside the rectenna area, and the changes persist directly downwind.

At 0600 hr (24-hr elapsed time), the 16-m temperature with rectenna in operation is slightly higher, by about 0.1°C within the rectenna boundary and 0.03°C outside. The absolute humidity is higher by about 0.1 g/kg within the rectenna area, and unchanged, on the average, outside the boundary.

The most notable changes occur in the cloud amount and cloud base height. At midday the cloud amount is increased within the rectenna area by 0.2-0.3 (cloud amount is measured on a scale of 0 to 1, 1 meaning complete cloud cover) and its base lowered by 100-200 m. Outside the rectenna area there are small decreases in cloud amount.

Table 5 shows the vertical profiles of the temperature, humidity, and vertical wind component changes at the center of rectenna at midday. The surface temperature is reduced by almost 2°C , and the greatest temperature change is about a 3°C reduction at a depth of 1 cm. The temperature change decreases monotonically with height except in the cloud-containing layers around 500-800 m. The absolute humidity is lower at the surface, higher in all layers up to cloud base and marginally lower above.

At midnight, the surface temperature at the center of the rectenna is slightly warmer (by about 0.15°C). The air layers are slightly warmer up to the 32-m level with no significant change above. The absolute humidity is slightly higher at the lower 6.4-m layer. The soil is slightly warmer down to a 2.5-cm depth.

The simulations indicated that, excluding the effects of albedo changes, the major cause of the perturbation is the change in surface roughness rather than the release of waste heat. Air and soil temperature decreased during the daytime and increased only marginally at night. The increased mechanical mixing resulted in increased evaporation and absolute humidity, increased cloud amount, and decreased cloud-base height. The decrease in solar radiation resulting from the increase in cloud amount is greater than the waste heat contribution. Cloud modification would be expected to be quite different if the roughness had not been changed.

Table 5 Changes of Temperature, Humidity, and Vertical Wind Component Profiles at the Center of the Rectenna, 1200 hr

Height (m)	Temp. Change (°C)	Humidity Change (g/kg)	Vertical Velocity Change (cm/s)
1500	-0.002	0	0.142
1000	-0.005	0	0.177
800	-0.010	-0.004	0.229
600	+0.004	-0.300	0.008
400	-0.049	-0.003	0.215
200	-0.084	+0.098	0.126
100	-0.111	+0.167	0.048
16	-0.205	+0.254	0
3.2	-0.420	+0.322	-0.002
0.0	-1.727	-2.340	0
-0.013	-2.914		
-0.050	-1.742		
-0.200	-0.104		

The simulation showed that for the given site characteristics and meteorological conditions, weather/climate effects of the construction and operation of a rectenna are small, particularly outside the boundary of the structure. From these results it seems clear that an albedo change is potentially the source of greater perturbations than appear in this simulation. An approximate analysis of the problem of rectenna albedo and its diurnal variation suggested that it would be difficult to simulate the albedo, even initially, to better than ± 0.02 , which at summer noon corresponds to a greater perturbation than the maximum waste heat in the center of the rectenna. It will be necessary in future work to quantify the albedo of a realistic structure with a realistic surface finish, and investigate the combined effects of albedo and roughness changes.

A first examination of the combined effects caused by the rectenna waste heat, surface roughness, and radiative properties was done by a series of model simulations with various boundary conditions (more specifically, by varying surface boundary temperature). The model used in these simulations was a three-dimensional, second-order turbulence closure equation set. Roughness lengths for the rectenna area and its surrounding were assumed to be 1 m and 3 cm, respectively. Meteorological conditions were selected for a daytime, boundary-layer situation (constant potential temperature up to 650 m in height, then increasing with height at a rate of $3.5^\circ\text{C}/\text{km}$) with logarithmic wind profile up to 4 m/s and remaining constant above that. The surface temperature was gradually increased with time (from $25\text{--}30^\circ\text{C}$ during the 8-hr integration) to simulate an unstable boundary layer. A constant relative humidity of 50% was assumed. No clouds were formed in this (dry convection) situation. The following linear relationships summarize results for some variables of interest for dry convection conditions, provided that the surface temperature difference between the rectenna and its environs remains small ($0 < T < 2^\circ\text{C}$):

$$V_2/V_1 = 0.733 + 0.014 \Delta T \quad (1)$$

$$u_2^*/u_1^* = 1.88 + 0.1 \Delta T \quad (2)$$

$$w = 7.2 + 3.9 \Delta T \text{ and} \quad (3)$$

$$H_2/H_1 = 3.47 + 1.555 \Delta T \quad (4)$$

where:

H = surface heat flux,

u^* = friction velocity,

V = 14 m horizontal wind speed,

$\Delta T(^{\circ}\text{C})$ = excess surface temperature of the
rectenna area over the upwind boundary, and

w (cm/s) = maximum vertical velocity;

subscripts 1 and 2 denote values at the upstream edge and
the center of the rectenna area, respectively.

The results of this numerical simulation indicated that increased roughness over the rectenna area considerably increases the friction velocity (by a factor of 1.9) and decreases the surface wind speed (by a factor of 0.73) at the center of the rectenna area in comparison with values at the upstream boundary. The resulting convergence of horizontal wind causes a maximum vertical wind component of 7.2 cm/s about 10 km downwind from the rectenna center and 700 m above grade. If sufficient energy is available at the surface boundary, an increase in surface heat flux of up to a factor of 3.5 could result from enhancing vertical turbulent mixing due to the increase in surface roughness inside the rectenna area, without perturbing the surface temperature. Sufficient energy could presumably be made available by reducing the surface albedo; under the simulated conditions, however, unrealistically large surface albedo change would be required. The simulated heat fluxes at the upstream boundary and the rectenna center are 91 and 316 W/m^2 , respectively. Inclusion of 8 W/m^2 of waste heat would cause a surface temperature perturbation of only 0.06°C , a negligibly small value. These simulations should, by no means, be considered very realistic. However, the results do show that an increase in roughness results in more efficient transport of vertical turbulent fluxes without significantly increasing the surface temperature. Therefore, it may be appropriate to conclude that surface roughness and albedo changes are the major causes of perturbation and that they may be of equal importance.

In examining the effects of the rectenna and its operation, consideration must also be given to the possible effect of the microwave beam itself, particularly due to absorption in water droplets. Information regarding the amount of microwave absorption per unit path length as a function of rainfall rate is available.²² With the most extreme rainfall rate of 254 mm/hr as an example, the attenuation at the proposed 2.45-GHz frequency is estimated to be about 0.063 db/km. At the proposed maximum power density of 230 W/m^2 , the absorbed microwave power inside the storm would be approximately $3.2 \times 10^{-3} \text{ W}/\text{m}^3$, which is approximately two orders of magnitude smaller than the release rate of the buoyant energy of a typical cumulus cloud. Therefore, it is reasonable to conclude that the absorption of SPS microwave power by a

storm will have no significant influence on cloud dynamics and thermodynamics and the associated precipitation.

2.2 INADVERTENT WEATHER MODIFICATION BECAUSE OF ROCKET LAUNCHES

The principal concerns about inadvertent weather modification by SPS rocket effluents are (1) the possibility that the ground cloud might temporarily modify local weather and (2) the cumulative effects of 375 or more launches per year. We shall discuss these issues of concern through the consideration of (1) the immediate dynamical and thermodynamical responses to the inputs of thermal energy and moisture from the rocket exhaust for given ambient meteorological conditions and (2) the possible alteration of the microphysical processes of clouds in the general area because of rocket effluents and debris and cooling water entrained during the launch. An analysis of the first aspect, namely, the immediate response of the atmosphere to the rocket effluents seems, at least at first glance, to be relatively straight-forward. Presumably one can use existing cloud models validated by previous rocket launch observations to simulate the convective motion of the cloud and the associated precipitation. Such models require as input, the initial thermal energy and total water content of the ground cloud and column cloud. However, measurements of these quantities are not generally available, and they must be estimated. The assessment of potentially significant inadvertent weather modification in terms of changes of cloud microphysical processes is similar to the assessment of planned weather modification. The degree of possible modification is complicated and not easy to assess; it depends upon the time of year, the site location, the origin of the air mass, and the various scales of weather patterns. Unfortunately, the assessment of possible rocket-related, inadvertent modification is even more difficult because there have been few observations made during past rocket launches that are relevant to both aspects.

The central issue with regard to cloud microphysical processes is the possible production of cloud condensation nuclei (CCN) and ice nuclei (IN) in the rocket-exhaust ground cloud. Cloud condensation nuclei serve as particles upon which water vapor condenses to form water droplets that in turn form clouds and fogs. They play an important role in determining the colloidal stability of clouds and the formation of precipitation. In general, the addition of CCN may tend to slow down the warm rain-formation processes if the total CCN exceeds 10^3 cm^{-3} . However, if very large hygroscopic particles (giant nuclei with radii $>25 \mu\text{m}$ like those expected to come from launch pad debris) are present, the rain-formation process may be accelerated. In the Florida area, some rainfalls are associated with condensation-freezing processes in a deep convection cloud system. In an IN-deficient, supercooled cloud, the addition of IN is expected to stimulate ice nucleation processes and lead to precipitation, although the effectiveness of this process (the basis for artificial cloud seeding) remains controversial. The global concentration of IN is about one per liter effective at -20°C . In planned weather modification, approximately 10 effective ice-forming nuclei per liter are added at a supercooled cloud temperature of approximately -10°C to -15°C for precipitation enhancement, and one to several hundred IN per liter are added for thunderstorm modification.

The measurements of the November 1978 Atlas/Centaur (a liquid, hydrocarbon-fueled rocket)^{23,24} ground cloud indicated that the concentrations of ice-forming nuclei generated by the rocket launch had limited potential for weather modification. However, because of uncertainties in measurement techniques and evidence for the creation of new IN as the ground cloud aged, the potential for weather modification due to the production of ice-forming nuclei cannot be assessed with confidence at this time. The concentrations of CCN in the Atlas/Centaur ground cloud were meteorologically significant. The initial emission was about 1.2×10^{17} CCN (active at 0.5% supersaturation), equivalent to a 10-s emission by the City of Denver. In addition, CCN were produced in the ground cloud at a rate of about $1 \text{ CCN cm}^{-3} \text{ s}^{-1}$. It should be noted that field and laboratory measurements^{23,24,25} of Titan III (solid-fueled booster) ground clouds concluded that both the IN and CCN concentrations were of meteorological significance. The maximum IN concentration active at -20°C could be as high as 2×10^5 per liter, which leads to an estimate of 10^{18} IN being produced per launch. The initial emission of CCN from the Titan III rocket was 10^{18} (active at 0.5% supersaturation). In addition, the CCN production rate within the cloud was fairly constant at $0.5\text{-}1 \text{ CCN cm}^{-3} \text{ s}^{-1}$ for a period of four hours after launch. The proposed HLLV would be liquid fueled. According to Table 1, the exhaust products from this liquid propellant are not considered to produce effective ice nuclei; any IN present in the exhaust ground cloud would presumably be derived from debris and cooling water entrained during rocket launch. It is not possible at this time to estimate whether the HLLV would produce a significant number of effective IN in terms of weather modification potential. The long-lasting concentrations of cloud condensation nuclei observed in both solid and liquid-fueled rocket clouds could alter the frequency and persistence of fogs and haze on the surface and the precipitation processes in warm clouds. In view of the high launch frequency proposed for the SPS space vehicles, a potential may exist for cumulative effects. More studies are needed in this regard.

Notable concentrations of dry particles as large as $200 \mu\text{m}$ in diameter have been detected within a few minutes after launch in the 1978 Atlas/Centaur ground cloud.²⁶ A distinct particle mode around $200 \mu\text{m}$ and some much larger particles ($\approx 500 \mu\text{m}$) were also found in the particle spectrum measured in the 1978 Titan III ground cloud.²⁶ Although the $\approx 500 \mu\text{m}$ particles in the Titan cloud were liquid drops, they probably represent entrained debris or cooling water spray droplets, since it is difficult to grow raindrops this large within a few minutes by condensation or collection processes. The entrained debris in the Atlas/Centaur ground cloud and its contribution to the population of giant nuclei were evident from elemental analysis of sample particles. The presence of these giant nuclei in the ground cloud will speed up the formation of precipitation if meteorological conditions are favorable.

In addition to these microphysical effects, the thermal energy and moisture contained in the rocket exhaust ground cloud are directly responsible for inducing a water-saturated, wet convective cloud and associated precipitation under certain meteorological conditions. The thermal energy provides sufficient buoyancy to lift the ground cloud and surrounding air to higher altitudes. During ascent, air cools through adiabatic expansion and, under certain conditions, reaches saturation to form a water-saturated cloud. Subsequently, cloud convection is further enhanced through the release of latent heat, leading in some cases to precipitation.

The phenomenon of a saturated, wet cloud formed by rocket exhaust has been observed on several occasions. Perhaps the most comprehensive and unique data are those obtained during a Titan III launch on December 13, 1978, at Kennedy Space Center. These data together with those observed during the November 1978 Atlas/Centaur launch were used to validate a cloud model* for use in the assessment of HLLV impact. For the case of the Titan III launch, temperature and dew point soundings prior to the time of launch (1940 EST) indicated that air in the surface boundary layer was humid but potentially stable as shown in Fig. 6. Rocket effluents produced a saturated white cloud having the characteristics of a moderately-sized, vigorous cumulus cloud. Aircraft measurements of the column cloud at 4 minutes after launch at a height of 1110 m indicated the cloud had droplet concentrations of 1000 cm^{-3} and a peak liquid water content (measured by the Johnson-Williams hot wire) of 1.2 g/m^3 . The liquid water was observed to decrease steadily later on. Measurements of the ground cloud were not made until 25 min after launch; at that time the ground cloud was still saturated, with a liquid water content of about 0.1 g/m^3 . Thereafter, only portions of the ground cloud were found to be saturated; however, the liquid water content was still detectable until 51 min after launch. The persistence of liquid droplets, long after the cessation of detectable updraft might be due to impurities in the liquid water (perhaps related to the presence of HCl).

The proposed HLLV would emit approximately $1.08 \times 10^{11} \text{ cal/s}$ of thermal energy together with $2.02 \times 10^7 \text{ g/s}$ of water to the atmosphere. Approximately 15 seconds of exhaust would be contained in the ground cloud. Model calculations indicated that, under the same meteorological condition, the HLLV thermal effluent could generate a much more vigorous convective cloud than the Titan effluent, as shown in Fig. 7. The maximum cloud liquid water content in the HLLV cloud was predicted to be about three times that of the Titan cloud, as shown in Fig. 8 (an initial thermal energy of $9.4 \times 10^{10} \text{ cal}$ was assumed in the Titan cloud). Furthermore, light precipitation with a maximum rainwater content of 0.07 g/kg was predicted for the HLLV cloud, but the duration of a saturated cloud was shorter. Virtually all the liquid water and precipitation arises from water initially present in the atmosphere, not from the water content of the HLLV and Titan rocket exhaust. The vigor of the perturbation generated by both rockets lasts for only a short time and the duration of both saturated clouds is less than 30 min. The calculated duration of a wet cloud being shorter than the observed one probably is due to the fact that the model calculation does not consider any impurities in the liquid water in the Titan ground cloud. Such impurity effects could be important and should be considered in future studies.

It should be noted that results given above should not be used to scale predictions of HLLV effects for all weather conditions. The degree of

*The model used was a modified version of one given by Lee (1971)²⁷ in which the eddy diffusivity is parameterized to be proportional to the deformation tensor, similar to that used by Cotton,²⁸ and the time rate of change of cloud radius is parameterized to be proportional to the eddy diffusivity. Documentation and validation (with observed rocket exhaust ground cloud) of the model are in preparation.

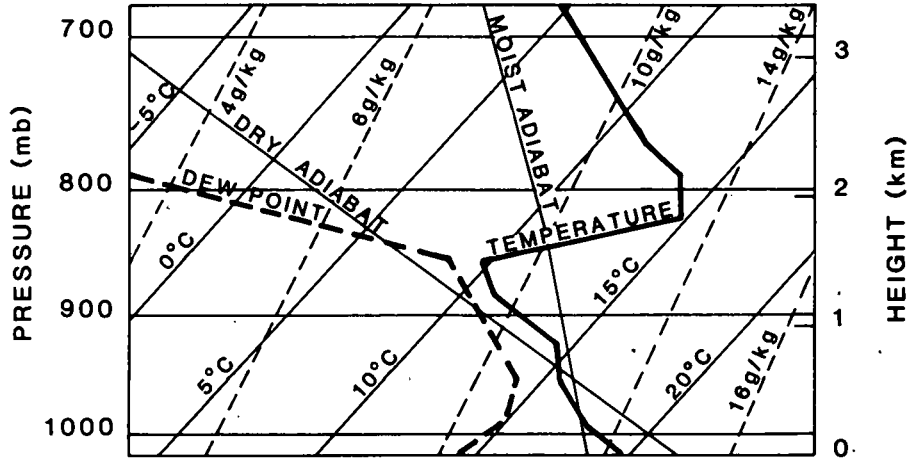


Fig. 6 Basic Sounding for 1936 EST December 13, 1978, Cape Canaveral, Florida

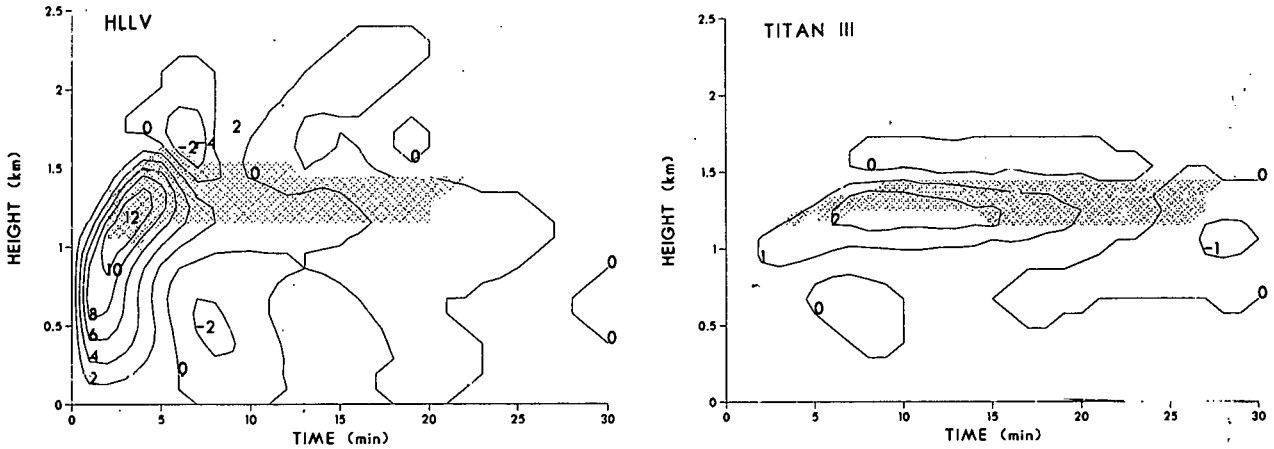


Fig. 7 Comparison of Predicted Maximum Vertical Velocity (m/s): HLLV and Titan III ground clouds (shaded areas are saturation regions)

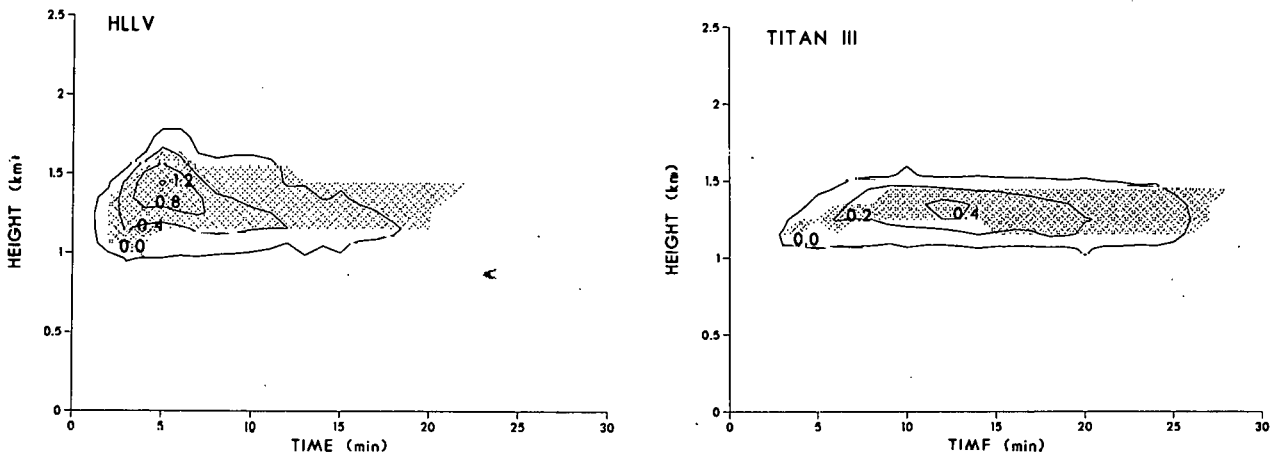


Fig. 8 Comparison of Predicted Mean Cloud Liquid Water Content (g/kg): HLLV and Titan III Ground Clouds (shaded areas are saturation regions)

impact among various sizes of rockets varies from one meteorological condition to another. Some examples, in addition to the above situation, representing typical conditions in the Florida area for various times of day and synoptic situations are discussed briefly in the following cases.

2.2.1 Midday Unstable to Neutral Boundary Layer Capped by Dry Stable Air

This situation is typical at midday when air is influenced by the anticyclonic high pressure system. Soundings used for this case were taken at 1304 EST, on April 16, 1972, at Cape Kennedy during the Apollo 16 launch.²⁹ The air, having an adiabatic lapse rate (on the average), was capped by dry stable air starting at about 2 km in height. Temperature and dew point soundings show that the atmosphere (with 58% relative humidity near the surface and 83% at 2 km) was potentially unstable (but relatively more stable than Case 2.2.4 below) with a lifting condensation level at about 1.2 km. Locally, a few scattered cumulus clouds with bases at about 2 km were observed.²⁹

A real-time motion picture of the evolution of the ground cloud formed after the Saturn V rocket launch is available from NASA. In reviewing this document, it was interesting to observe that a white, rapidly ascending, convective cloud formed at approximately 2 min after launch within the faintly visible brownish column of exhaust gas. This formation was followed by an optically dense convective white bubble penetrating through the top of the brownish ground cloud. Meanwhile, a natural cumulus cloud entered the circulation cell, descended and dissipated within a couple of minutes. Unfortunately, this document only showed the evolution of the ground cloud for the first 5-10 min.

Model simulations under these conditions showed that quasi-steady-state, convective cloud motions with similar intensities were generated by all types of rockets, even those for which the exhaust thermal energies are different by two orders of magnitude (such as the HLLV and the Atlas/Centaur). The results differ only for the first 10-20 min, as indicated by the vertical velocity fields, and approach each other later on. The predicted maximum cloud liquid water for the HLLV cloud was about 1.3 g/kg with light precipitation at about 0.13 g/kg of rainwater. The predicted precipitation is only slightly lower for other smaller rocket clouds.

2.2.2 Mid-Morning Conditionally Unstable Boundary Layer Capped by Dry Stable Air

The synoptic influence is similar to the case above. During mid-morning, the air in the surface boundary layer has not yet absorbed enough solar energy to develop a neutral boundary layer.

A Titan IIIC rocket (AF-777) was launched at the Kennedy Space Center at 1004 EDT on May 20, 1975. Archived sounding data³⁰ taken immediately after launch show that a subsidence inversion existed with a base of about 2 km. Below this dry stable air was a boundary layer with a temperature lapse rate averaging between dry and moist adiabatic and a near-constant relative humidity of about 62%. The Titan ground cloud was observed

to rise to an altitude of 2 km and stabilize at that altitude. Model calculations showed that the HLLV exhaust cloud would penetrate several hundred meters higher through the inversion layer and then subside to the 2-km altitude. Except for a brief period (between 2 and 10 min) of saturation, no significant cloud modification was calculated for the HLLV exhaust ground cloud.

2.2.3 Nighttime Absolutely Stable Condition

The sounding used here is that taken at 0059 EST on November 13, 1978, at Cape Canaveral during the Atlas/Centaur launch. A marine inversion at about 700 m above grade was capped by a dry stable layer above. Relative humidity in the marine inversion layer was about 95%. A strong nocturnal ground-base inversion of approximately 130 m in thickness was also present. Immediately after launch, a cloud was observed to rise vertically from the launch pad to the stable atmospheric layer at about 1100 m; it was then observed to rapidly subside between 2 and 15 min after launch. The ground cloud stabilized within 200 m at 600 m above grade. The cloud was sub-saturated and no liquid water was detected.²⁴

Model prediction showed that the HLLV thermal effluent (about 100 times that of the Atlas/Centaur) would raise the ground cloud to a height of 2 km, penetrating through the stable dry layer, and then stabilize within this stable layer centered at about 1.2 km in height. A couple of brief periods of saturation were predicted, with a maximum liquid water content of only about 0.16 g/kg. No significant cloud modification from the HLLV exhaust was predicted in this case.

2.2.4 Potentially Unstable Air, Thunderstorm Situations

Viking B was launched during the time when a cold front was passing through the Florida Peninsula on September 9, 1975. Thunder to the west was audible, and cumulonimbus clouds were observed in all quadrants at the time of launch.³¹ The sounding taken at 1424 EDT (15 min before launch) shows that the atmosphere was potentially highly unstable with a dry adiabatic temperature lapse rate below 900 m and a moist adiabatic lapse rate above 2300 m. Relative humidities were above 80% throughout the entire layer below 6 km with saturation recorded around the 3.5- and 6-km levels. Wind speeds were light with little shear throughout the layer. The ground cloud formed by the Titan III rocket effluent was observed to rise to 1.7 km at 8 min after launch and then to subside to about 1.5 km at 12 min after launch. During this period the track of a thunderstorm cell located to the west of the launch site intersected the exhaust ground cloud.³¹ No further report about the ground-cloud trajectory was available thereafter. The model calculation using rawinsonde data taken at Cape Canaveral AFS prior to the launch confirmed the observed trajectory. It also showed that the ground cloud only encounters a brief period of saturation (between 4-8 min after launch) between 1.1-1.3 km, with a maximum liquid water content of less than 0.1 g/kg. The characteristic pattern of dry convection is seemingly not influenced by the small amount of latent heat of condensation.

It is interesting to see the calculated results for the HLLV under these conditions. The thermal energy contained in the HLLV ground cloud induces a violent convective storm reaching 6 km in height after 60 min of convection. The first light precipitation reaches the surface at about 15 min, and is followed by a pulse of heavy rainfall at 40 min and a much weaker pulse at 52 min. Fig. 9 shows the calculated mean cloud liquid water content and rainwater as functions of time and height for this case. The storm cloud was predicted to dissipate rapidly and became insignificant after 60 min of calculation.

This result demonstrates an example of a situation in which a convective instability is present, but only the larger rocket has sufficient energy input to "trigger" the instability.

The above discussion highlights only the possible effects in some chosen meteorological conditions. In view of the nonlinearity and the relative insensitivity of the results to the rocket energy output in some situations, a complete future assessment should include a climatology of the HLLV impacts for a given launch site for an updated HLLV design.

In summary, the HLLV with its huge amount of thermal effluent will produce a larger and higher cloud than other existing smaller rockets. Because of its ability to raise the cloud and the surrounding air to higher elevations, the SPS rocket is more likely to generate a saturated wet cloud and light precipitation under high relative humidity but potentially stable conditions, and has a higher potential of releasing an existing convective instability than smaller rockets. The degree of cloud modification attributable to the launching of various rockets is not linearly related and depends sensitively upon meteorological conditions. Generally, the conditions that favor onshore flow without strong westerlies above the planetary boundary layer are conducive to greater inadvertent weather modification by SPS rocket launches in the Florida area. Characteristic synoptic weather regimes that would fall into this category were identified in a theoretical study of space-shuttle exhaust clouds.³² The conditions include hurricanes; easterly waves of summer; stagnating frontal zones; cool-season squall lines; cool-season, low-latitude, mid-tropospheric troughs; warm-season, weak, mid-tropospheric troughs; and coastal sea-breeze convergence regimes.

The upper-bound probabilities for disturbed conditions resulting from the aforementioned events are given in the shuttle ground cloud study.³² Disturbed conditions (cumulonimbus coverage 20-50%), persisting 24-72 hr may encompass the Cape Canaveral area once every three or four years due to the influence of hurricanes or tropical storms. Easterly waves with disturbed conditions persisting 12-24 hr may reach the Florida area every four or five days between mid July and September. Stagnating fronts across central Florida happen five to six days in March and December (less in January and February) and two to three days in early June and late September. Disturbed conditions and accompanying rainfall may persist for 12-24 hr. Occasional squall lines with precipitation duration of one hour or less may sweep across central Florida in winter. Cool season, low-latitude, mid-tropospheric troughs occur only once every three years. Finally, warm-season, weak, mid-tropospheric troughs can interact with the sea breeze convergence regime to produce potentially highly unstable conditions several days each month. This leads to a general rule for mitigation in terms of rocket launches. With

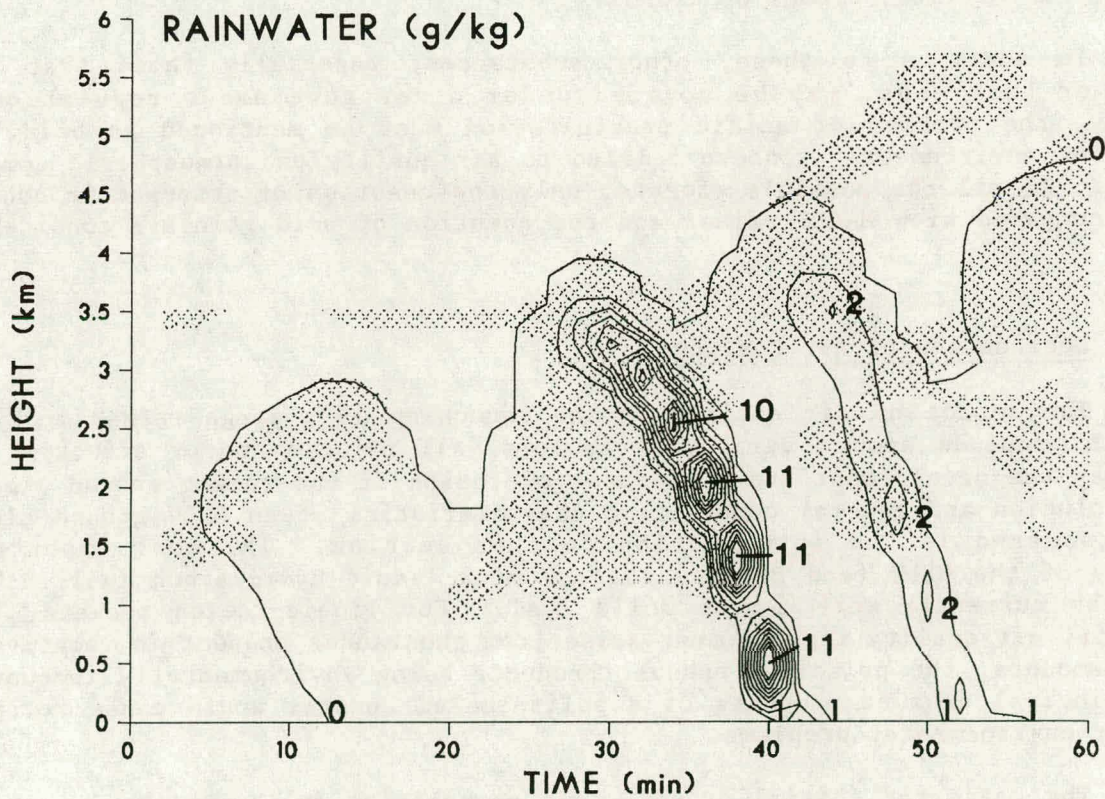
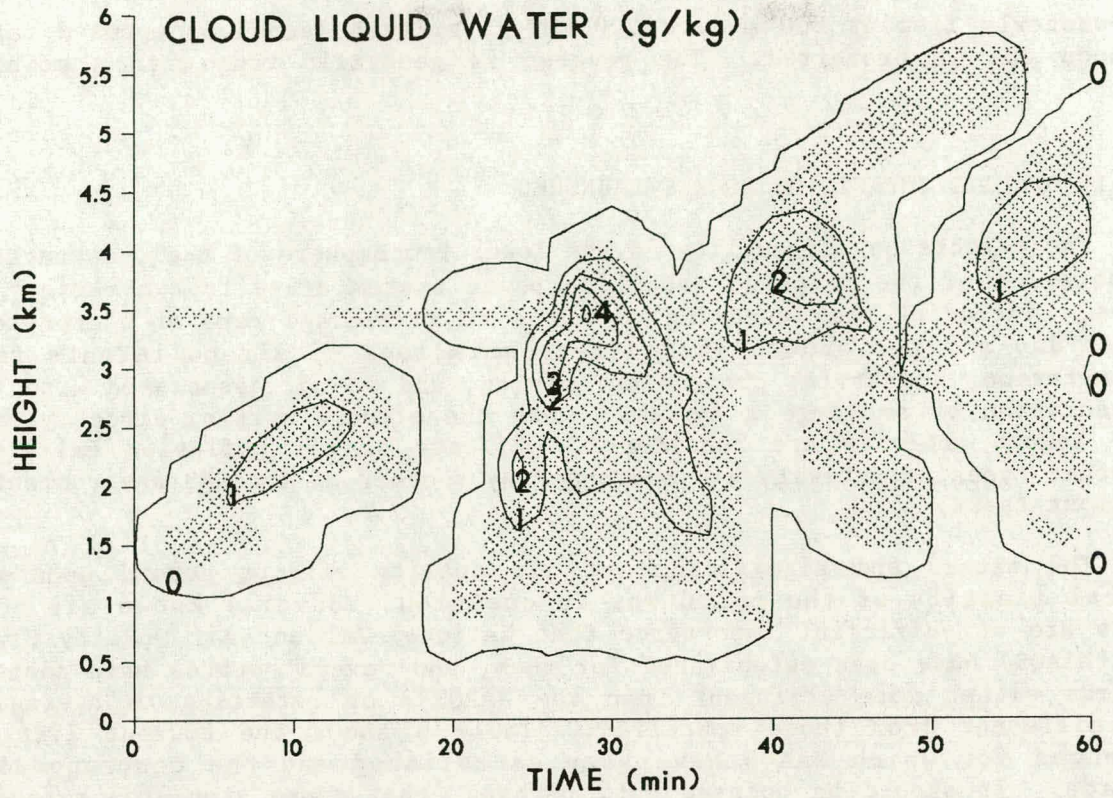


Fig. 9. Predicted Mean Cloud Liquid Water Content and Rainwater Induced by the HLLV Thermal Effluent for Potentially Unstable Air on September 9, 1975 (shaded areas are saturation regions)

south-westerly flow at 850 mb at 1200 GMT, a morning launch as opposed to an afternoon one is preferred. The reverse is generally true with a morning flow.

2.3 AIR QUALITY IMPACTS OF ROCKET LAUNCHES

The effects on air quality in the lower troposphere of the construction and operation of the proposed satellite power system arise from a variety of sources, including the launching of HLLVs at an average rate of better than one per day and the growth in indirect emissions of air pollutants from transportation, industrial production, mining, and so on, associated with the implementation of so large a system. Only the effects arising directly from rocket launch activity are considered. Effects of air pollution emissions from other types of activity are addressed by another of the SPS environmental assessment tasks.

The nature and significance of air quality effects depend upon the chemical identity of the pollutant in question. Several known air pollutants are of sufficient importance that National Ambient Air Quality Standards (NAAQS) have been established for them, and several states have adopted standards either more stringent than the NAAQS³³ or referring to averaging times different from those specified. Table 6 shows the current list of substances for which NAAQS have been established and the corresponding standards. It should be pointed out, perhaps, that these standards refer to ground-level concentrations at fixed locations only, and not to peak concentrations in elevated plumes or clouds.

In addition to these, other substances, especially those that are toxic or hazardous, may be covered under other government regulations. Finally, the problem of acidic precipitation must be mentioned as being an important environmental concern related to air quality and atmospheric composition. Of all the possible effects, only the question of atmospheric conditions compared with NAAQS values and the question of acid rain are considered here.

2.3.1 Degradation of Air Quality

The impact on air quality of the launching of a large rocket such as the HLLV depends upon a variety of factors, all relating to or affecting in some way the development and subsequent evolution of the rocket ground cloud. The evolution and general composition characteristics of an HLLV ground cloud are discussed in the introduction to this section. The most important feature of the HLLV (and PLV) is the use of a liquid hydrocarbon fuel rather than the currently more common solid fuel. For liquid-fueled rockets, the potential air quality impacts must arise from substances present in relatively small amounts, the principal exhaust products being environmentally innocuous. The principal exhaust products of a solid-fueled booster would cause notably greater environmental problems.

The estimated initial ground cloud composition following the launch of an HLLV is given in Table 7. In preparing these estimates, the quantity of exhaust products involved was taken to be that generated during the first

Table 6 National Ambient Air Quality Standards

Pollutant	Concentration (Averaging Time)
Particulate matter	
Primary	260 $\mu\text{g}/\text{m}^3$ (24 hr), 75 $\mu\text{g}/\text{m}^3$ (1 yr) ^a
Secondary	150 $\mu\text{g}/\text{m}^3$ (24 hr), 60 $\mu\text{g}/\text{m}^3$ (1 yr) ^a
Sulfur dioxide	
Primary	365 $\mu\text{g}/\text{m}^3$ (0.14 ppm) ^b (24 hr), 80 $\mu\text{g}/\text{m}^3$ (0.03 ppm) (1 yr)
Secondary	1300 $\mu\text{g}/\text{m}^3$ (0.5 ppm) (3 hr)
Carbon monoxide	40 mg/m^3 (35 ppm) (1 hr) 10 mg/m^3 (8.7 ppm) (8 hr)
Ozone	235 $\mu\text{g}/\text{m}^3$ (0.12 ppm) (1 yr)
Nitrogen dioxide	100 $\mu\text{g}/\text{m}^3$ (0.05 ppm) (1 hr) [470 $\mu\text{g}/\text{m}^3$ (0.25 ppm) (1 hr)] ^c
Lead	1.5 $\mu\text{g}/\text{m}^3$ (3 mo)

^aGeometric mean. All other values in table are arithmetic means.

^bConcentrations given in ppm refer to parts per million by volume, at a total pressure of one atmosphere and a temperature of 25°C.

^cCalifornia standard, possible future national standard.

15-25 s following ignition and liftoff, and all the molecular hydrogen and carbon monoxide present in the exhaust at the rocket engine nozzle was assumed to be converted to water and carbon dioxide by afterburning. In addition, the presence of a sulfur impurity in the fuel at the 0.05% level was assumed; this figure corresponds to about the average sulfur content of hydrocarbon fuels now in use. The actual level of a sulfur impurity in the proposed liquid methane fuel may be considerably lower.

In Table 7, the concentrations given for SO₂, NO, and NO₂ represent the maximum values actually measured in the ground cloud of an Atlas/Centaur rocket, as discussed below. In using the same values for the HLLV ground cloud, the assumptions are that the amounts of NO and NO₂ produced are in the same proportion to the total amount of exhaust in both rockets and that the exhaust is diluted with ambient air by the same factor, which is approximately 4600, based upon the maximum observed SO₂ concentration in the Atlas/Centaur cloud and the corresponding fuel analysis that indicated a sulfur impurity of 0.047%. Also, based on this dilution factor and the data in Table 7, the total initial volume of an HLLV ground cloud will be in excess of 10¹⁰ m³.

These assumptions are supported by airborne measurements of peak NO_x concentrations in several ground clouds associated with launches of the Titan III solid-fueled rocket.³⁶ In those measurements, the greatest peak NO_x concentration observed was 1.4 ppmv; two other peak values

Table 7 Estimated Initial HLLV Ground-Cloud Composition

Substance	Mass (10 ³ kg)	Peak Concentration (ppmv) ^a
H ₂ O ^b	400-650	-
CO ₂	450-700	23.0
SO ₂	0.7-1.1	0.023 (3.25) ^c
NO	9-14	0.64 (0.34) ^d
NO ₂	10-16	0.50 (0.34) ^d
Particulates ^e	0.02-2	

^aParts per million by volume at 1 atm total pressure, 25°C.

^bIncludes only the water actually emitted by rocket engine.

^cAverage value observed in power-plant plumes at a distance of approximately 1 km from the smoke stack.³⁴

^dAverage value of NO_x concentration (NO_x = NO + NO₂) in power-plant plumes at a distance of approximately 1 km from the smoke stack.³⁵

^eEstimated amount of particulate matter after the first 5-10 min, during which the very large particles, including dust and debris, will fall out.

in excess of one ppm were also reported (1.1 and 1.0 ppmv), and the remaining values ranged from 0.65 to 0.25 ppmv. The values shown in Table 7 are quite comparable to these even though the two rockets are significantly different, the Atlas/Centaur rocket being liquid fueled and having only about 22% the gross liftoff weight of the Titan III. The agreement tends to indicate that the peak NO_x concentrations are nearly independent of the type of rocket, in support of the assumptions made in Table 7.

Other factors that affect the extent to which a rocket launch gives rise to air quality degradation include (1) the altitude to which the ground cloud rises because of its initial thermal energy content and (2) the ambient meteorological conditions, which not only affect the ground cloud stabilization altitude but also determine the rates of dispersion of an air pollutant and the rates at which the pollutant is removed from the atmosphere by wet and/or dry deposition or by absorption in liquid water droplets. In addition, the local topography (including presence or absence of a coastline) affects the local wind patterns and thereby the transport of the ground cloud and/or other pollutants. Finally, in the case of chemically reactive substances such as NO and NO₂, the presence of pollutants such as ozone and other photochemical smog-related substances will affect the ground cloud concentrations and the rate at which acidic substances such as nitric acid are produced. From this brief list, it should be clear that making a complete assessment of air quality effects under all possible conditions is a difficult and complex task.

In order to prepare the best possible assessment within time and budgeting constraints, a combination of field monitoring and theoretical modeling approaches was adopted. Prior to the SPS CDEP program, no data obtained by field monitoring of actual launches existed for a liquid-fueled rocket. Field data for solid-fueled rockets were not considered satisfactory in view of the differences in the chemical nature of the exhaust products from solid and liquid fuels. A field monitoring program was undertaken to acquire data considered necessary for the present assessment. The launch of an Atlas/Centaur rocket on November 13, 1978, provided an opportunity to acquire the extensive set of airborne measurements made in connection with that launch.^{24,37}

A total of 52 passes were made through the Atlas/Centaur ground cloud over a 2 1/2-hr period following the launch, using the University of Washington's instrumented B-23 research aircraft. A variety of parameters was measured, including among others the concentrations of several gaseous chemical species, the particle size distribution, the CCN and IN concentrations, and the physical dimensions of the cloud. The maximum concentrations of NO, NO₂, and SO₂ were measured during the second pass, 5 min after liftoff (the aircraft did not travel through the center of the cloud during the first pass), and are given in Table 7 as indicated above. The cloud stabilized within an inversion layer between approximately 450 and 650 m above ground level, and remained at that altitude until it merged with an industrial plume about 2 hr after launch. Dispersion of the cloud in the vertical direction was strongly limited by the inversion layer, and the contribution of the cloud to ground level concentrations was certainly negligible in this case. The cloud grew horizontally during the measurement period, and cloud concentrations decreased accordingly until at 110 min after launch, measured peak concentrations were 7 ppbv (SO₂), 65 ppbv (NO₂), and 45 ppbv (NO). Systematic conversion of NO to NO₂ was observed as the cloud dispersed. This conversion may be attributed to the presence of ozone in the ambient air at a concentration of approximately 55 ppbv; as the ground cloud grew in size and entrained more and more ambient air, the ozone reacted with NO in the cloud, forming NO₂. This explanation is supported by ozone measurements in the cloud, which show that the ozone concentration is strongly depressed below the ambient level until the cloud is just barely detectable by other means. The rate of conversion is also in accord with independent measurements of the rate of growth of the cloud.

The Atlas/Centaur launch took place at 12:24 am local time, i.e., shortly after midnight, and, as indicated, stabilized above the ground cloud within an inversion layer at about 500 m above ground. Meteorological conditions at night are often such that vertical dispersion of an elevated cloud is suppressed; under such conditions, ground-level concentrations of substances in the ground cloud are below detection limits. During the day, however, the level of atmospheric turbulence and the associated rate of dispersion may be very much higher, and the possibility of significant ground-level concentrations of various substances from the ground cloud must be considered. The wind direction frequency in the vicinity of the launch site is an important factor in assessing the importance of ground-level pollutant concentrations. In the vicinity of Kennedy Space Center, Florida, a rocket ground cloud may drift over land about 50% of the time, based upon data presented in the KSC environmental impact statement.³⁶ This frequency should be kept in mind during the following discussion, which assumes that the ground cloud is over land.

The only substances present in the ground cloud of the Atlas/ Centaur, and, by analogy, in the cloud from an HLLV, that are meaningful from an air quality point of view are SO_2 , NO , and NO_2 . The composition of the cloud was quite similar to that of a plume from a coal-fired electric power generating plant with defective control equipment burning low-sulfur coal. In order to put the expected environmental effects of an HLLV launch into perspective, it is useful to pursue this analogy. Table 7 shows that the expected NO_x concentrations in an HLLV cloud are 3-4 times higher than those prevalent in power plant plumes while a sulfur impurity of 0.05% leads to SO_2 concentrations that are only 15-30% of those present in typical power-plant plumes. In addition, the cloud elevation is comparable to plume elevations from large (1000 MW or more) power plants, and therefore the maximum ground-level concentrations of SO_2 and NO_x may be expected to be 15-30% and 300-400%, respectively, of those from a typical large power plant. It must be kept in mind, however, that the ground cloud represents a short, intense emission whereas a plume represents a continuous emission. In order to compare effects, an averaging time must be introduced. In fact, the relevant averaging times are explicitly specified in the corresponding air quality standards. For example, the federal secondary standard for SO_2 specifies that the ambient SO_2 concentration averaged over a 3-hr period should not exceed 500 ppbv more than once per year, as shown in Table 6. Reference to Table 7 shows, however, that even the maximum SO_2 concentration in the center of the ground cloud is much less than this value. In fact, the maximum SO_2 concentration is even less than the primary annual SO_2 standard of 0.03 ppmv. The conclusion is that any reasonable level of sulfur in the fuel will not give rise to significant ambient ground-level SO_2 concentrations, particularly when dispersion and cloud elevation are considered.

No adverse effects of any kind are expected from the level of CO_2 shown in Table 7. To put these emissions into perspective, the total annual emission of CO_2 from the entire first-stage burn of the HLLV (375 flights) amounts to 1×10^9 kg. By way of comparison, the annual emission from a single 1000 MW, coal-fired power plant is approximately 4.5×10^9 kg (using 207 lb $\text{CO}_2/10^6$ Btu heat content, 34% efficiency, 55% annual average load factor). Thus, the entire annual CO_2 emission from SPS-related spaceflight amounts to only 23% of the total from a single typical power plant, a totally negligible amount from the point of view of the global CO_2 content. No local, short-term effects of any kind are expected, since the peak concentration is only 23 ppmv above an ambient value of about 335 ppmv.

No short-term federal air quality standard currently exists for any of the nitrogen oxides. However, it is likely that within a year or two, a one-hr standard for NO_2 of approximately 0.25 ppmv will be promulgated. The problem that one encounters in attempting to predict NO_2 and NO concentrations is that both are reactive substances, and the common techniques used to predict atmospheric concentrations of air pollutants do not apply in this case. The NO_2 concentration is determined by the initial amounts of NO and NO_2 present in the cloud, by the extent of dispersion of the cloud, by the composition of the ambient air, particularly the ambient ozone level, and by the ambient light intensity. Nitrogen dioxide is continually photolyzed by sunlight to form NO and atomic oxygen (which in turn combines with ambient molecular oxygen to form ozone), and at the same time NO and ozone react to re-form NO_2 . The concentrations of all three species are determined by the relative rates of these competing processes. Standard atmospheric dispersion

modeling techniques are not applicable under these circumstances. In order to assess the potential contribution of an HLLV ground cloud and the associated NO_x emissions to ground-level NO_x concentrations, a new modeling approach, applicable specifically to the estimation of short-term NO_2 concentrations and developed for general NO_x sources, was used. The principal chemical reactions^{38,39} involving NO_2 and NO in the atmosphere are (1) the photolysis of NO_2 in the presence of molecular oxygen to produce NO and ozone (O_3):



and (2) the reaction between NO and O_3 to reform NO_2 :



The rate coefficient for reaction A (k_A) varies from zero at night to about 0.6 min^{-1} (0.01 sec^{-1}) for an overhead sun.³⁶ The rate coefficient for reaction B (k_B) is equal to $2.1 \times 10^{-12} \exp(-1450/T) \text{ cm}^3 \text{ sec}^{-1}$ ⁴⁰ (T being the temperature (Kelvin)), or $24.0 \text{ ppm}^{-1} \text{ min}^{-1}$ at 25°C , 1 atmosphere total pressure. The rates of these two reactions are fast compared with the rates of other reactions involving these substances, for typical tropospheric concentrations. Since A and B are the reverse of each other, an approximate photostationary state is established in which the two reactions balance each other.⁴¹ In such a state, the atmospheric concentrations of NO , NO_2 , and O_3 are related by

$$\frac{[\text{NO}][\text{O}_3]}{[\text{NO}_2]} = K \quad (5)$$

in which

$$K = \frac{k_a}{k_b} \quad (6)$$

Given an initial mixture of these substances that does not satisfy Eq. 5, chemical reactions A and B will occur in such a way as to change the concentrations until they do satisfy the photostationary state relation. The final (stationary-state) concentrations may be calculated by the following procedure: denote the change in the NO_2 concentration $[\text{NO}_2]_{\text{ss}} - [\text{NO}_2]_i$ by c (ss denoting stationary state, i denoting initial); then the changes in the NO and O_3 concentrations are equal to $-c$, so that $[\text{NO}]_{\text{ss}} = [\text{NO}]_i - c$, and similarly for the ozone concentration. Substitution into Eq. 5 yields a quadratic equation for c , the physically realizable solution of which is

$$c = 1/2 \left\{ (a + K) - [b^2 + K(4[\text{NO}_2]_i + 2a + K)]^{1/2} \right\} \quad (7)$$

in which

$$a = [\text{NO}]_i + [\text{O}_3]_i \quad (8)$$

and

$$b = [\text{NO}]_i - [\text{O}_3]_i \quad (9)$$

The time required for the photostationary state to be reached depends upon the values of the rate coefficients k_A and k_B and upon the initial concentrations. A convenient measure of this time is the relaxation time (τ) for the stationary state, defined by

$$\tau^{-1} = k_A + k_B ([NO]_{ss} + [O_3]_{ss}) \quad (10)$$

Figure 10 shows the variation of τ as a function of the stationary state concentrations for various values of k_A , with k_B equal to $24.0 \text{ ppm}^{-1} \text{ min}^{-1}$.

A number of investigators have reported deviations of measured atmospheric concentrations from Eq. 5. Calvert⁴² suggested that deviations are caused by spatial inhomogeneities in the atmospheric composition, these inhomogeneities being averaged out in the measurements. Seinfeld⁴¹ presented an analysis in support of this suggestion, in which Eq. 5 was assumed to hold only locally. More recently, Chock and Kumar⁴³ have pointed out that in order to test rigorously the photostationary state equation in the atmosphere, measurements will need to be made with instruments having response times less than the time interval over which significant composition changes are encountered. The validity of the equation has been verified in laboratory studies.⁴⁴

The modeling approach adopted for assessing the potential NO_2 contribution at ground level from HLLVs is based on the assumption that the photostationary state relation, Eq. 5, is valid locally within a diffusing cloud (or plume) initially containing a mixture of NO_x and O_3 different from that in the ambient atmosphere. This assumption is equivalent to the assumption that the rate at which the concentrations within a small parcel of air approach the photostationary state (as measured, for example, by the relaxation time τ) is large compared with the rate at which those concentrations are changing because of diffusional processes. Analyses based upon the assumption of a local photostationary state have been found to explain accurately observations within power-plant plumes at distances sufficiently near the stack.^{34,35,45} In order to make use of Eq. 5 in a dispersion model, a prediction of the atmospheric composition that would obtain in the absence of chemical processes must be available; in particular, one must be able to estimate the volume mixing ratio (χ) of effluent from the source in question on the assumption that the effluent is unreactive. Given an estimate of χ , which will in general depend on the position and time at which a prediction is to be made, one calculates "initial" values of the NO , NO_2 , and O_3 concentrations from

$$[X]_i = (1 - \chi) [X]_{\text{ambient}} + \chi [X]_{\text{source}} \quad (11)$$

where

$[X]_{\text{ambient}}$

and $[X]_{\text{source}}$ = ambient and undiluted source concentrations, respectively

X = any of the NO , NO_2 , or O_3

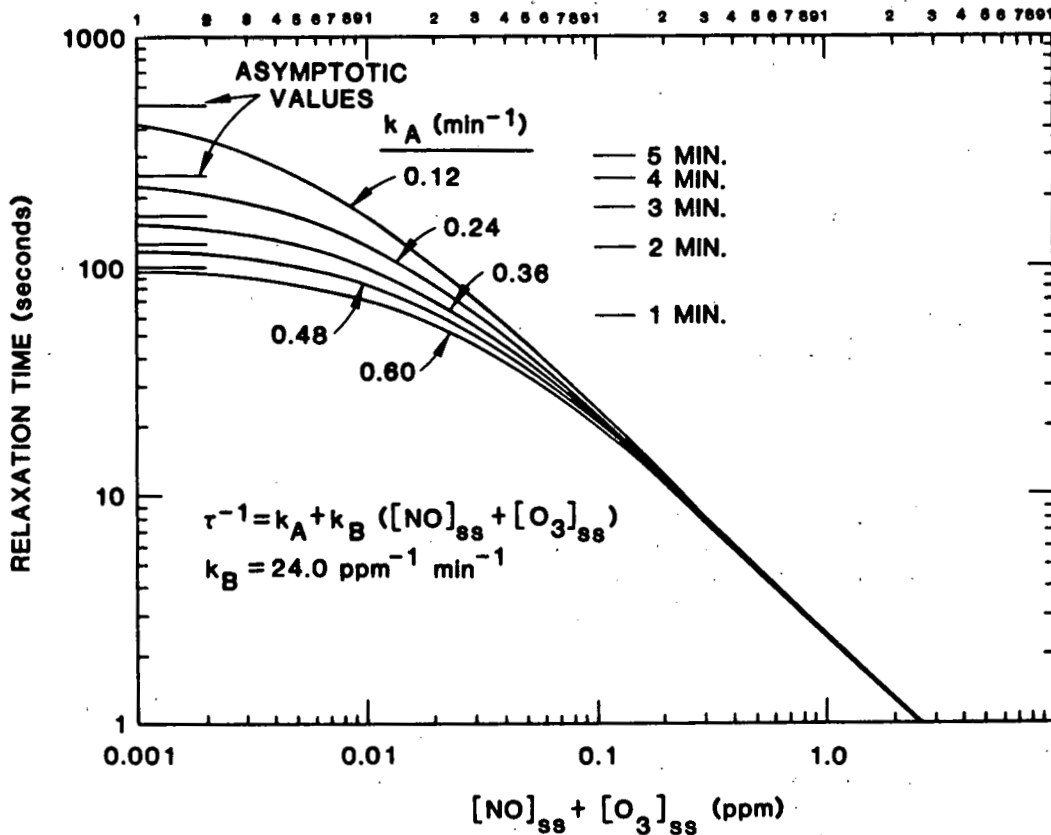


Fig. 10 Photostationary State Relaxation Times

Substitution into Eqs. 7-9 allows the quantity c to be calculated, and thereby the concentrations that satisfy the photostationary state for the location and time in question.

Application of this procedure requires a method for estimating χ . In the case of a rocket ground cloud, the model used for this purpose consisted of an assumed Gaussian distribution $g(r,t)$ in the horizontal plane, the center location and standard deviation $\sigma(t)$ of which are independent of altitude but are time dependent, with the vertical distribution $f(z,t)$ being determined as the numerical solution to a one-dimensional diffusion equation with an (eddy) diffusivity dependent on altitude but constant in time. Thus, the cloud is assumed to be cylindrically symmetric with a vertical axis of symmetry at all times, and the mixing ratio may be written

$$\chi_k(r,t) = A g(r,t) f_k(t) \quad (12)$$

where

A = a normalization coefficient, and k denotes the vertical grid point z_k at which the quantity $f_k(t) = f(z_k,t)$ is being evaluated, and the function g is given by

$$g(r,t) = \frac{1}{2\pi\sigma^2(t)} \exp \left[-1/2 \left(\frac{r}{\sigma(t)} \right)^2 \right]. \quad (13)$$

The coefficient A is determined by the initial condition that was used; specifically, that the ground cloud was uniformly distributed vertically between 1100 and 1500 m altitude, the initial value of σ being determined from the estimated volume of rocket exhaust involved and the estimated initial dilution factor of 4600. It was also assumed that an elevated inversion layer existed between 1300 and 1700, and therefore that the ground cloud had partially penetrated the inversion layer. The assumed initial conditions approximate the stabilization of a ground cloud at an elevated inversion at a typical altitude for daytime conditions.

The eddy diffusivity used in the numerical evaluation of the quantities $f_k(t)$ and the horizontal standard deviation $\sigma(t)$ were essentially those estimated by Lamb et al.⁴⁴ for unstable conditions based upon numerical simulations of dispersion in the planetary boundary layer. The initial value of σ was estimated to be 2.2×10^3 m, and the rate of change of σ with time given by Lamb et al., when dimensionalized by an assumed friction velocity of 0.5 m/sec, resulted in $\sigma(t)$ being given by

$$\sigma^2(t) = \sigma^2(0) + 0.653 t^2 \text{ (sec)}. \quad (14)$$

Table 8 gives the ground-level concentration of NO, NO₂, and O₃ as functions of the scaled distance from the cloud axis (r/σ) at the time of maximum ground-level NO₂ concentration, about 20 min after cloud stabilization. At this time, $\sigma = 2.335 \times 10^3$ m. Ambient concentrations were: [NO_x] = 0.25 ppm and [O₃] = 0.12 ppm. Ambient temperature and pressure were 27°C and 1 atmosphere, k_A was taken to be $4 \times 10^{-3} \text{ sec}^{-1}$ and the photostationary state constant K was therefore equal to 9.82×10^{-3} ppm. The depression of the ozone concentration by 50% at the cloud centerline compared with the ambient value and the increase in NO₂ concentration by 0.07 ppm above the ambient level are clearly seen. Figure 11 shows the time evolution of the ground-level NO₂ concentration as a function of (r/σ) for the same set of conditions. The values designating the different contours are given in ppm. Nitrogen dioxide (NO₂) concentrations in excess of 0.25 ppm persist at the cloud centerline for nearly 2 hr; if the average velocity of the cloud were less than about 4 km/hr, some region at ground level would be exposed to NO₂ concentrations in excess of the (possible) 1-hr standard for a period longer than an hour and an exceedance of the standard would occur. Thus, it seems possible that in some circumstances the launch of an HLLV might cause an exceedance of a possible 0.25 ppm 1-hr NO₂ standard.

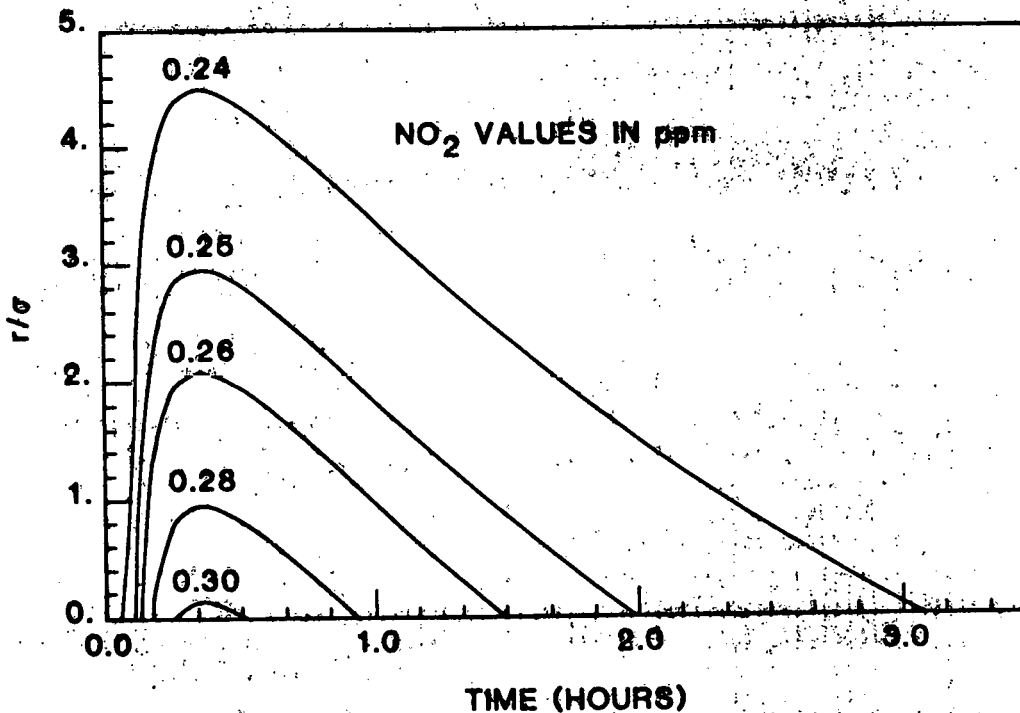
It was not possible to consider a variety of conditions in order to identify the range of those under which a possible exceedance of such a standard might occur. In the example given above, the ambient concentration was already close to the 0.25 ppm value, and the maximum increase due to the ground cloud was only 0.07 ppm. In addition, under neutral rather than unstable conditions, the rate of vertical dispersion is significantly less and so are the calculated ground-level concentrations. On the other hand, the ground-level concentrations are sensitively dependent upon the assumed cloud stabilization height and could be much increased if the cloud stabilized at a lower altitude. Fumigation-type situations brought about by launches just

Table 8 Ground Level NO, NO₂, and O₃ Concentrations^a

Scaled Distance	[NO ₂]	[NO]	[O ₃]
(r/σ)	(ppm)	(ppm)	(ppm)
0.0	0.304	0.050	0.060
0.5	0.297	0.045	0.065
1.0	0.279	0.034	0.080
1.5	0.258	0.026	0.098
2.0	0.243	0.020	0.110
2.5	0.235	0.019	0.117
3.0	0.232	0.019	0.119
	0.231 ^b	0.019 ^b	0.120 ^b

^a20 min after cloud stabilization.

^bConcentrations outside the plume.

Fig. 11 Ground-Level NO₂ Concentration

prior to sunrise might also represent notably adverse conditions and should be considered. The general conclusion is therefore that the possibility exists of an HLLV launch making a significant contribution to ground-level NO₂ concentrations, but that the combinations of meteorological conditions and ambient pollutant levels in which an exceedance of the 0.25 ppm level would result have not been fully identified. Once these situations have been determined, however, the obvious mitigating strategy of not launching a rocket under those conditions could be effectively applied.

A potential problem related to the question of $\text{NO}_x - \text{O}_3$ chemistry in a ground cloud is the possible formation of ozone in excess of ambient levels after several hours of travel. The generation of ozone concentrations in excess of ambient levels has been observed on several occasions in power-plant plumes.⁴⁷⁻⁵¹ This phenomenon is not well understood at present, although there seems to be no reason why it cannot be accounted for by using known chemistry. The possibility exists that a similar phenomenon might occur in the ground cloud from an HLLV in view of its chemical similarity to a power-plant plume, although ground-level ozone concentrations would probably not be greatly affected. It would be desirable to examine the question using a suitable theoretical model in order to determine the conditions under which such a phenomenon is expected to occur and the extent to which an HLLV ground cloud may produce the same effect.

2.3.2 Formation of Acid and Acid Rain from Rocket Exhaust Effluent

The increase in acidity of precipitation in the northeastern United States and eastern Canada in recent years is a subject of serious concern,⁵²⁻⁵⁴ and has prompted the U.S. Environmental Protection Agency to undertake an extensive research program to examine various aspects of the problem.⁵⁵ Although the formation of sulfuric acid from gaseous sulfur dioxide is considered to be the main cause of the observed acid rain, nitric acid formed from oxides of nitrogen has also been implicated.^{53, 55-57} As each piece of additional information becomes available, the scope of the problem seems to increase; thus, evidence for trends of increasing acidity was recently published for Colorado,⁵⁶ and Florida.⁵⁷ In view of the widespread nature of the problem and the current level of concern, an assessment of the possible significance for acid rain of the effluent from nearly 400 HLLV launches per year was undertaken.

Two main aspects must be considered: (1) the question of local, intermittent effects associated with individual launches and (2) regional, long-term contributions representing cumulative effects over the entire period of high-rocket-launch activity. In many respects, the first aspect is the more difficult of the two to access, because one is more concerned with the details of the chemistry and transport; this aspect is treated in the following section. The potential long-term, regional contributions may be reasonably assessed by evaluating the significance of the HLLV launches as sources of acid precursors compared with other sources in the area of the launch site; this approach will be taken in the discussion of long-term effects.

2.3.2.1 Local Short-term Effects

Two general mechanisms may be proposed by which oxides of nitrogen are converted to acids in the atmosphere. The first involves the absorption of NO and NO_2 into existing droplets and subsequent chemical conversion to acids. The acids which must be considered are nitric acid (HNO_3), a strong acid, and nitrous acid (HNO_2), a weak acid. Of the two, nitric acid is much more important as a potential contributor to the acidity of water droplets in the atmosphere. The second general mechanism involves the conversion of NO and NO_2 to nitric acid in the gas phase, followed by absorption of the acid

in ambient droplets. Gas-phase conversion occurs by reaction with free-radical components present in photochemical smog.^{38,39} In addition, in the presence of sodium chloride aerosol and NO₂, hydrochloric acid may be evolved.⁵⁸ Each process is considered in turn.

The chemistry of NO and NO₂ in aqueous solution is rather complex.⁵⁹⁻⁶¹ Two general reactions must be considered: (1) the reaction of NO₂ to form HNO₂ and HNO₃:



and (2) the formation of HNO₂ from an equimolar mixture of NO and NO₂:



Both reactions are reversible, as indicated by the equality signs. A third reaction often considered, the disproportionation of HNO₂,



is not independent of the first two, but may be obtained by subtracting twice reaction (D) from reaction (C). A possible mechanism by which these reactions proceed is shown in Fig. 12.⁶¹⁻⁶⁵ The mechanism shown is compatible with the observations and in particular accounts for the observed rates of the forward and reverse reactions (C) as studied by Abel and Schmid⁶⁶ in the 1920s and 1930s. According to the reactions shown, nitric acid is formed by reaction of N₂O₄ (the dimer of NO₂) with water, and an equimolar mixture of the two acids is formed initially on dissolving only NO₂ in water. As the concentrations of acid increase, HNO₂ is recycled and converted to HNO₃, the entire system eventually approaching equilibrium. Equilibrium constants are known for reactions (C) to (E), and it is possible to calculate the concentrations of NO and NO₂ in the gas phase and the concentrations of HNO₂ and HNO₃ in the liquid phase at equilibrium, given an initial gas-phase composition and the ratio of the volume of solution to the volume of gas at equilibrium. Such a calculation was carried out in order to estimate the equilibrium concentration of HNO₃ in water droplets in the presence of initial concentrations of NO and NO₂ equal to the maximum concentrations expected in the HLLV ground cloud. The atmospheric liquid water content was assumed to be 5.6 x 10⁻⁶ L per L of air at equilibrium, corresponding approximately to a rainfall rate of 10 mm/hr with an average droplet radius of 0.15 mm. Only the equilibria between gaseous NO and NO₂ and aqueous solutions of HNO₂ and HNO₃ were considered:

$$\frac{[\text{HNO}_2]^2}{P_{\text{NO}}P_{\text{NO}_2}} = K_B = 122(65) \text{ at } 25^\circ\text{C}$$

$$\frac{[\text{HNO}_2][\text{H}^+][\text{NO}_3^-]}{P_{\text{NO}_2}} = K_A = 4.3 \times 10^5(65) \text{ at } 25^\circ\text{C}$$

Activity coefficients were assumed to be unity, [H⁺] was assumed equal to [NO₃⁻], and both equal to [HNO₃]. Concentrations are in mol/L, pressures in atmospheres. The initial and equilibrium concentrations and partial pressures are given in Table 9. The acid concentrations shown in the table correspond to a pH of 2.62, which represents the pH after sufficient time has

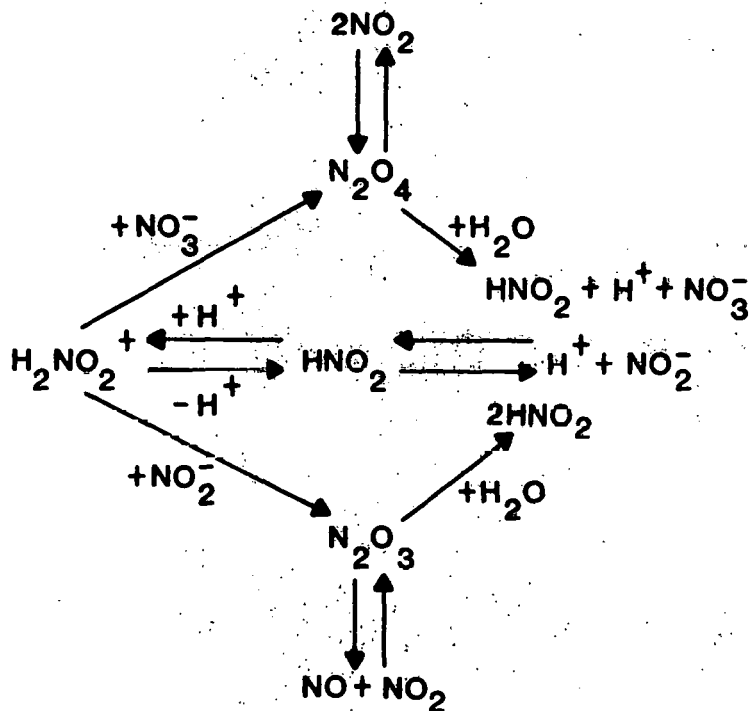


Fig. 12 Possible Mechanism of the Aqueous Chemistry of Nitrogen Oxides

elapsed for the system to come to equilibrium. Estimates of this time are given below. The results depend on the liquid water content; lower values of the water content would yield lower pH values, and the consideration of reaction kinetics becomes more important for obtaining reasonably accurate estimates.

Somewhat similar but more extensive calculations have been carried out in an attempt to explain observed nitrate and sulfate concentrations in atmospheric aerosol.⁶⁷⁻⁶⁹ In those calculations, the equilibrium expressions corresponding to reactions (C) to (E) were assumed to hold at all times, and the aerosol dynamics were assumed to be driven by the rate of oxidation of SO_2 .

Table 9 Equilibrium NO , NO_2 and Acid Concentrations^a

	Partial Pressures (atm)		Concentrations (mol/L)	
	NO	NO_2	HNO_2	HNO_3
Initial	6.40×10^{-7}	5.00×10^{-7}	0	0
Final	8.06×10^{-7}	2.63×10^{-9}	5.08×10^{-7}	2.42×10^{-3}

^aTotal atmospheric pressure = 1 atm, temperature = 25°C, liquid water content = 5.6×10^{-6} L/L air.

Individual measurements of precipitation pH in the northeastern U.S. have given values as low as 2.1,⁵² and average values are as low as 4.1 in a few locations.⁵³ The average in north Florida is about 4.6, with individual measurements as low as 3.8.⁵⁷ As indicated earlier, however, most of this acidity is attributed to sulfuric acid rather than nitric acid, and the estimated pH given above for droplets in an HLLV ground cloud is undoubtedly *too low* (i.e., too acidic), due primarily to the assumption of equilibrium. A better estimate must take into account the finite rates of transfer of gaseous NO and NO₂ into solution and the rates of the chemical reactions by which HNO₃ is produced.

Not all the rate information that would be in principle necessary for this task is available. However, an upper-bound estimate of the rate of HNO₃ production in water droplets in the presence of gaseous NO₂ can be made. The details of the derivation are given in the appendix to this report; the result, for an atmospheric water content of 5.6×10^{-6} L/L and an NO₂ concentration of 500 ppb, is

$$[\text{HNO}_3] = [\text{HNO}_3]_0 + (1.4 \times 10^{-10} \text{ mol/L/s})t \quad (15)$$

where

t = the elapsed time in seconds and

$[\text{HNO}_3]_0$ = an assumed initial nitric acid concentration.

This estimated rate of nitric acid production is extremely slow; if a value of $[\text{HNO}_3]_0$ corresponding to a pH of 4.6 (the annual north Florida average) is assumed and the length of time it would take for the pH to drop by 0.1 unit is calculated, the result is 5×10^4 s, or about 14 hr. Conversely, the amount of acid produced in the time required for a droplet of radius 0.15 mm to fall through the ground cloud and hit the ground would be completely undetectable. It should also be pointed out that the approximations made in the course of the calculation, specifically the neglect of the chemical effect of NO, the neglect of the removal of NO₂ from the air (and in fact of any process such as atmospheric dispersion that would tend to decrease the ambient NO₂ partial pressure), and the neglect of evaporation of HNO₃ from the droplet, are all conservative in the sense that they all lead to an overestimate of the aqueous HNO₃ concentration. It seems safe to conclude, therefore, that acid production by this mechanism can be neglected in an assessment of short-term effects of an HLLV ground cloud. This qualitative result would still hold, although less comfortably, if the acid production rate were an order of magnitude higher.

Production of acid in the gas phase by involvement in photochemical smog takes place primarily by reaction of NO₂ with hydroxyl radicals:^{38,39}



The hydroxyl radicals themselves are generated by a variety of processes, including photolysis of ozone, to give molecular oxygen plus an excited oxygen atom and subsequent reaction of the excited oxygen atom with ambient water vapor. The significance of this process for short-term effects is difficult to establish without detailed modeling of the gas-phase chemistry

occurring within the cloud, a task outside the scope of this assessment. However, it is possible to estimate the amount of nitric acid produced over 1-2 hr by this reaction by using a representative tropospheric daytime hydroxyl radical concentration of $1 \times 10^6 \text{ cm}^{-3}$ ⁷⁰ in combination with the rate constant for reaction (F) of $1.1 \times 10^{-11} \text{ cm}^3 \text{ molecule}^{-1} \text{ sec}^{-1}$.⁴⁰ The product of these two quantities gives an approximate conversion rate of $4\% \text{ hr}^{-1}$ for NO_2 conversion to HNO_3 . While nitric acid is being produced at this rate, the ground cloud disperses in both vertical and horizontal directions. If the horizontal distribution of the cloud is assumed to be Gaussian, the vertically integrated NO_2 concentration will also be Gaussian (neglecting chemical interaction of cloud constituents with ambient ozone), and the maximum (central) value of the integrated NO_2 concentration will decrease with time by the factor $h(t)$ given by (with t in hr):

$$h(t) = \frac{1}{1 + 1.14 t^2} \quad (16)$$

This expression represents the ratio $\sigma^2(0)/\sigma^2(t)$ and may be easily obtained from Eq. 14. If nitric acid is assumed to be uniformly produced at a constant rate of 0.04 hr^{-1} , the central vertically integrated HNO_3 gas-phase concentration will be given by

$$\int [\text{HNO}_3]_{(\text{gas})} dz = (8.2 \times 10^{-3} \frac{\text{mol}}{\text{m}^2}) (1 - e^{-0.04t}) h(t) \quad (17)$$

in which $8.2 \times 10^{-3} \text{ mol/m}^2$ is the estimated central, vertically integrated HNO_3 concentration in the initial ground cloud. The highest value attained by the integrated HNO_3 concentration is reached about 1 hr after launch, and is equal to $1.5 \times 10^{-4} \text{ mol/m}^2$. If the vertical distribution after 1 hr is taken to be uniform up to an assumed inversion height of 1500 m, the corresponding gas-phase nitric acid concentration is $1 \times 10^{-7} \text{ mol/m}^3$ corresponding to $1 \times 10^{-10} \text{ mol/L}$ or 24.5 ppb. Assuming that all this nitric acid is absorbed into rainfall ($V_{\text{liq}}/V_{\text{air}} = 5.6 \times 10^{-6} \text{ L/L}$), the corresponding aqueous acid concentration is $1.8 \times 10^{-5} \text{ mol/L}$. For purposes of comparison, an aqueous solution with a pH of 4.6 contains 2.5×10^{-5} moles of hydrogen ions per liter, and if the estimated contribution from the ground cloud NO_2 is added to this, the pH would decrease by 0.23 units to 4.37.

In many respects, the approximations involved in the above calculation lead to an overestimate of the amount of acid produced. Due to the limitations of the study, no attempt was made to identify adverse or worst-case meteorological or other conditions. It seems safe to conclude that, on a local, short-term basis, the nitric acid produced by an HLLV launch will not have any significant adverse effects, although an assessment of ecological effects is also outside the scope of this analysis. It would, however, be desirable to take a more detailed look at the question, taking into account both the gas-phase chemistry of the ground cloud as well as the dynamics of the cloud itself, in order to establish with greater certainty the conclusion just reached.

The third process by which atmospheric NO_2 may give rise to acid is that reported by Robbins et al.⁵⁸ In that work, it was found that in mixtures of sodium chloride (NaCl) aerosol and gaseous NO_2 in air of varying

degrees of humidity, the sodium chloride was replaced to an extent (dependent on humidity) by sodium nitrate, and an equivalent amount of hydrochloric acid was produced. In the experiments, up to 80% of the initial NaCl aerosol was converted to NaNO_3 within less than ten minutes at 97% relative humidity. The experimental conditions generally involved NO_2 and NaCl levels several orders of magnitude higher than natural levels, although one experiment was carried out with 0.1 ppm NO_2 and 12 g NaCl/m^3 at 88% relative humidity, with the result that 44% of the NaCl was converted to NaNO_3 , and presumably also resulted in the simultaneous production of 9×10^{-11} mol HCl/L air. Using a value of $7 \text{ g}/\text{m}^3$ ⁵⁸ for the concentration of NaCl aerosol on the Florida coast and assuming complete conversion into HCl, the corresponding HCl concentration is estimated to be 1.2×10^{-10} mol HCl/L.

If this acid is completely absorbed into droplets ($V_{\text{liq}}/V_{\text{air}} = 5.6 \times 10^{-6}$ L/L), the aqueous HCl concentration would be 2.1×10^{-5} mol/L, which would reduce the pH from an assumed 4.60 initially to 4.34. The amount of NO_2 used up in the process is also 1.2×10^{-10} mol/L so that the initial ground cloud level of 500 ppb (2.0×10^{-8} mol/L) would not be noticeably affected and the other acid-forming processes discussed above would operate as before. If the possible 1.2×10^{-10} mol/L of strong acid from the aerosol reaction is combined with 1.0×10^{-10} mol/L from photochemical conversion, the total hydrogen ion concentration in liquid droplets with $V_{\text{liq}}/V_{\text{air}} = 5.6 \times 10^{-6}$ L/L would be 3.9×10^{-5} mol/L, corresponding to a total pH change from 4.6 to 4.2. A change in pH from 4.6 to 4.2 corresponds to increasing the hydrogen ion concentration by a factor of about 2.5. It does not seem likely that the pH level in a single precipitation event in north Florida could be lowered by an amount significantly greater than this due to a single HLLV launch.

2.3.2.2 Regional, Long-term Effects

For the purpose of assessing long-term effects, it is most convenient to consider the projected 375 HLLV launches/yr in comparison with other NO_x sources and in combination with available data regarding the effect of oxides of nitrogen on acidic precipitation in Florida. Recently, data has been published by Brezonik et al.⁵⁷ that indicates that over the past 20-25 years, the average pH of precipitation in northern Florida has fallen by about one unit to a present average value of 4.6-4.7. This decrease has been accompanied by a 60% increase in the concentration of "excess" sulfate, that amount in excess of the quantity due to sea spray, and an increase by a *factor* of 4.5 in the nitrate concentration in precipitation. Ionic balance in precipitation samples indicate that sulfuric acid is 2.0-2.5 times more important than nitric acid, that the two together account for nearly all the acidity of Florida rainfall, and that the geographic distribution of NO_3^- is relatively isotropic, indicating that nitric acid contributes a "diffuse background of H^+ to precipitation. . ."57

Multiple regression analyses indicated that 75% of the variation in H^+ could be explained by a linear two-variable equation, including the concentrations of sulfate and calcium ions; inclusion of nitrate did not improve the correlation. The interpretation of this result was that, in general, the acidity of Florida rainfall may be accounted for in terms of an interaction between sulfuric acid and terrestrial basic calcium salts (e.g., limestone-calcium carbonate), although an isotropic contribution from nitric acid would not have been detected in the regression analysis.

In order to estimate the effect of a single new source of NO_x on regional or statewide precipitation acidity, it has been assumed that the average hydrogen ion concentration can be written approximately as

$$[\text{H}^+] = a + b ([\text{NO}_3^-] + [\text{SO}_4^{2-}]_{\text{xs}}) - c [\text{Ca}^{++}]_{\text{xs}} \quad (18)$$

in which the subscript xs indicates excess over the sea spray contribution. This equation is similar to that of Brezonik et al.⁵⁷ but with a term linear in nitrate added with the same coefficient as sulfate. If now the average nitrate concentration is increased by a factor $(1 + \mu)$, due to an assumed new source of nitric acid, the new hydrogen ion concentration becomes

$$[\text{H}^+]_{\text{new}} = [\text{H}^+]_{\text{old}} \left(1 + b\mu \frac{[\text{NO}_3^-]_{\text{old}}}{[\text{H}^+]_{\text{old}}} \right). \quad (19)$$

The quantity μ may be identified with the fractional increase in NO_x emissions due to the new source, assuming that contributions to acidity are proportional to NO_x source strengths. Ambiguity in this procedure arises when one tries to decide exactly what other NO_x sources to use in the comparison. The apparent isotropy of the nitric acid contribution may imply that the other relevant sources are all those within the entire statewide area. To be on the conservative side, we consider only NO_x emissions from within Brevard Co., the county in which Kennedy Space Center is located; the result will be a substantial overestimate of the impact. Total estimated annual NO_x emissions for Brevard Co. for the year 1985 are 3.1×10^7 kg,³⁴ while the estimated total NO_x production from 375 HLLV launches per year amounts to approximately 1.13×10^7 kg/yr, or 36% of the 1985 Brevard Co. total. Taking $\mu = 0.36$ and using values of $b = 0.54$, $[\text{NO}_3^-]_{\text{old}} = 1.41 \times 10^{-5}$ mol/L and $[\text{H}^+]_{\text{old}} = 1.73 \times 10^{-5}$ mol/L,⁵⁷ the fractional increase in the average hydrogen-ion concentration is estimated from Eq. 19 to be 0.16, and the corresponding decrease in the average pH value is 0.07 units. A change of this magnitude would probably not be detectable using present methods.

3 ASSESSMENT OF UPPER ATMOSPHERIC EFFECTS

3.1 THE NATURAL UPPER ATMOSPHERE

The stratosphere and mesosphere are the lowest of those regions that taken all together constitute the "upper atmosphere" as shown in Fig. 4. The stratosphere lies immediately above the lowest atmospheric region, the troposphere, in which mankind conducts its daily activities and which contains approximately 85% of the total mass of the atmosphere. The classification of atmospheric regions is most commonly made with regard to the observed variation of temperature with altitude, although if one is mainly concerned with ionic processes, a different classification scheme is often used for altitudes above about 60 km. Thus, the mean temperature of the atmosphere is observed to decrease up to an altitude of about 11-12 km, depending, for example, upon latitude and seasons of the year.

The tropopause may be defined as the altitude at which the atmospheric temperature stops decreasing, and marks the boundary between the troposphere and the stratosphere. The lower stratosphere (up to about 20 km) is nearly isothermal, but from 20 km up to about 50 km the temperature increases. The altitude at which this increase stops and the temperature begins to fall again is termed the stratopause and marks the boundary between the stratosphere and the mesosphere. The mesosphere is that region in which the temperature is observed to decrease with altitude again, and spans the range from 50 to about 80 km. Above 80 km (the mesopause), the temperature is observed to increase with altitude again, and this higher region is termed the thermosphere. The thermosphere, which extends up to about 500 km where it blends into the exosphere, is a transition region in which the temperature becomes uniform with altitude. The exosphere is so named because of neutral particles that are moving at escape velocity (about 10 km/s at 1000 km) or greater can escape from Earth's gravitational field once they reach this altitude. It is well known that Earth continuously loses hydrogen atoms via this route. As high-velocity hydrogen atoms escape from the exosphere, other H atoms continuously diffuse upward from the denser regions of the atmosphere toward maintaining thermodynamic equilibrium in Earth's gravitational field. This upward transport and removal of hydrogen is called the "natural hydrogen cycle."

At an altitude of about 60 km, the ambient concentration of ionic species and electrons begins to be high enough to affect the propagation of radio waves, and the ionosphere is generally considered to begin at about this altitude. Historically, the ionosphere has been divided into several overlapping regions: the D-region, which essentially coincides with the mesosphere, extends from about 60-90 km; the E-region, which occupies the lowest portion of the thermosphere and extends up to about 140-150 km; and the F-region, which extends out to about 800-1000 km where it blends smoothly into the plasmasphere.

Because of the competing production (photoionization) and loss (diffusion and recombination) mechanism, the density of free electrons in the ionosphere increases to a peak value, called the F2 peak, around 250-400 km. Regions above and below this level are referred to as the topside and bottomside ionosphere, respectively. Radio waves transmitted from the earth's surface having a frequency below the critical frequency characteristic of

the F2 peak are reflected from the bottomside ionosphere. Frequencies above the critical frequency are transmitted through the ionosphere. Depending upon the frequency, the amplitude and/or phases of the transmitted waves may become altered.

The D- and E-region ion population consists primarily of molecular ions, but the F-region ion population is dominated by atomic ions, especially atomic oxygen ions (O^+). Above about 1000 km protons (atomic hydrogen ions, H^+) become the dominant ionic species.

Near the upper boundary of the E-region, the mean free path becomes long enough so that charged-particle motion becomes strongly influenced by the geomagnetic field. This altitude marks the beginning of the magnetosphere. In this plasmasphere, extending from about 1000 km to $4 R_E$ (Earth radii), thermal ions collide with other particles about as often as they bounce back and forth along the dipole-like geomagnetic field lines, whereas energetic particles are likely to gyrate and bounce back and forth for much longer periods of time before colliding with other particles.

The overall temperature structure of the atmosphere is such that it tends to inhibit the transport of material from one region to the next. Consequently, the extent to which one region influences another is often quite unclear and involves coupling mechanisms of a rather subtle nature. Furthermore, while the atmosphere becomes increasingly rarified with altitude, its susceptibility to change, due both to natural and man-made perturbations, increases. Unfortunately, since our knowledge of detailed processes in the atmosphere tends to diminish with increasing altitude, the problem of assessing the impacts of large-scale, man-made perturbations is difficult. In the next section an attempt is made to place the possible extent of SPS-related perturbations in perspective.

3.2 SPS VS AMBIENT LOADINGS

The Satellite Power System involves an extremely large space construction project, each of 60 proposed satellites having a solar collector array of dimensions 5 x 10 km and a mass of 35,000-50,000 tons. Construction of two such satellites at geosynchronous orbit (GEO) at 35,800-km altitude would involve the launch of a vehicle five times the size of the Space Shuttle at a rate of once or twice per day for 30 years. Both the vehicle size and launch frequency are unprecedented. The purpose of this section is to put the emissions from the various space transportation vehicles required for SPS into perspective by comparing them with ambient concentrations of the various exhaust constituents in different atmospheric regions. Such comparisons are intended to give a very rough idea of the potential importance of the rocket effluents. More detailed analyses are given in the following sections.

In order to make this comparison in a meaningful way, it is convenient to introduce a dimensionless quantity that we term the perturbation factor (PF). The perturbation factor is defined as the ratio of the change in concentration of a given chemical substance divided by the ambient unperturbed concentration, and is therefore simply the fractional change in concentrations:

$$PF = \frac{\text{Expected Concentration Change}}{\text{Ambient Concentration}} \quad (20)$$

As can be seen from this definition, the numerical value of the PF in a particular situation is a measure of the significance of the change in concentration of a given substance being injected into the atmosphere relative to its ambient concentration. Thus, for example, a value of unity for the PF would indicate that the concentration of the particular substance is expected to increase by an amount equal to its existing concentration, and a PF value of very much less than one, would indicate that the expected fractional change in concentration will be small.

In order to estimate numerical values for the PF, an estimate of the expected concentration changes must be made. Although in general the atmospheric concentration of a given rocket exhaust effluent will change with time following a launch, the simplifying assumption will be made here that the SPS emissions are continuous at the average yearly emission rate corresponding to the proposed launch frequency. A simple one-dimensional model was used to estimate the corresponding change in the steady-state concentration of water vapor at altitudes up to 1000 km. For other effluents, the following approximate linear relation was used:

$$\begin{aligned} \text{Expected Concentration Change} = \\ (\text{Injection Rate}) \times (\text{Residence Time}), \end{aligned} \quad (21)$$

which applies in a steady-state situation, provided that the Residence Time is taken to be the inverse of the effective rate of disappearance of the given substance from the domain in question. It may be shown that Eq. 21 represents a lower bound to the result obtained when using a more sophisticated model for a steady-state situation. The Residence Time represents the combined effects of transport and chemical (or photochemical) removal, and in general is given by an expression of the form

$$\begin{aligned} (\text{Residence Time})^{-1} = \text{Transport Removal Rate} + \\ \text{Chemical Removal Rate}. \end{aligned} \quad (22)$$

The concept of a PF is very convenient for indicating the potential magnitude of the expected effects.

This section is concerned primarily with rocket emissions into the stratosphere and above. Emissions into the troposphere are expected to have only localized effects, which are discussed in Sec. 2. The principal exhaust products, as discussed in the introduction, are CO₂ and H₂O from the HLLV main burn up to staging at 56 km, H₂O and H₂ between 56 and 125 km from the HLLV second stage, H₂O and H₂ at LEO due to HLLV second stage circularization and deorbit burns, NO between about 60 and 90 km due to HLLV second stage reentry, H₂O and H₂ due to POTV transit between LEO and GEO, and an Ar⁺ ion plasma between LEO and GEO due to the COTV. Each substance is considered in turn.

3.2.1 Carbon Dioxide

In Sec. 2.3, it was pointed out that the total annual emissions from 375 HLLV launches would amount to only 23% of the quantity produced by a typical 1000-MW, coal-fired power plant in a year. If emissions into the stratosphere are considered and a stratospheric residence time for CO₂ of one year is assumed, Eqs. 20 and 21 yield values of the perturbation factor that range about an average of 4×10^{-6} , depending on the specific value chosen for the ambient CO₂ number density.⁹ The previous conclusion that CO₂ emissions would not be significant is strongly supported by this result.

3.2.2 Water and Molecular Hydrogen

Total emissions of H₂O and H₂ were given in the introduction. The distribution in altitude of those emissions is such that nearly as much water is deposited in the 120-125 km altitude range as in that of 0-5 km, but the ambient density at 120 km is only 2×10^{-8} times that at sea level; the overall effect at 120 km is very large.

Figure 13 gives the corresponding change in the water vapor concentration as a function of altitude from 0-1000 km, based on a simplified steady-state, one-dimensional model with diffusive transport and an altitude-dependent, effective chemical removal rate coefficient. The concentration profile for water in the absence of chemical reaction with oxygen ions (O⁺) is also shown, and the importance of this process for removing water is clearly seen. The O⁺ profile used in the calculation is also given. The peak in the H₂O profile at 120 km is a result of the large emission rate near that altitude; the peak in the lower stratosphere results from the relatively low value of the eddy diffusivity at those altitudes.

In order to compute the perturbation factor as a function of altitude using Eq. 20, values of the ambient water concentration at various altitudes are needed. The problem of measuring and explaining water vapor concentrations in the stratosphere and mesosphere has occupied atmospheric scientists for many years, and several recent reviews and discussions are available.⁷¹⁻⁷³ Up to about 80-90 km, the water mixing ratio is a reasonably constant 3-4 ppmv; no measurements are available above this altitude range. Model calculations^{74,75} indicate that the water-mixing ratio drops dramatically above 80 km because of the increasing rates of photolysis and reaction with ambient species such as O⁺. A calculation was carried out using the same one-dimensional model used to estimate the change in the water concentration because of SPS-related emissions, but in which the flux at 80 km was adjusted so as to produce a mixing ratio of 3 ppmv at 80 km. No rocket emissions were included. Table 10 lists some of the results and also gives, for comparison, the expected contributions from rocket emissions and the corresponding values of the perturbation factor. Again, the effect of emissions in the vicinity of 120 km may be seen, and the expected effect on water concentrations above 120 km is an increase of approximately a factor of 130-150.

As indicated in the introduction, HLLV emissions of molecular hydrogen are about one-third of the water emissions on a number basis. Model calculations^{74,75} indicate, however, that the natural abundance of H₂ is two or more orders of magnitude higher than that of H₂O in the lower thermosphere. As a

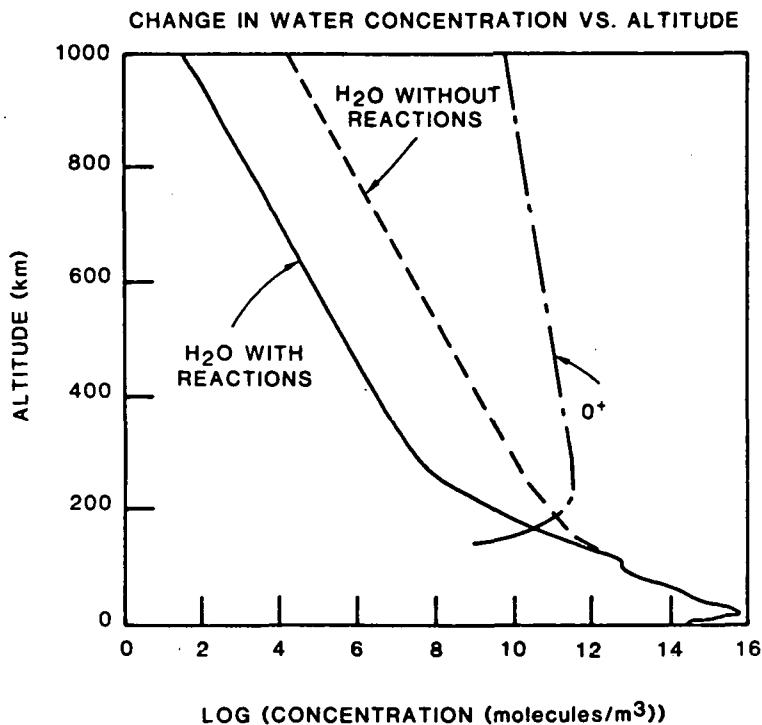


Fig. 13 Estimated Change in Water Vapor Concentration

result, the molecular hydrogen emissions are far less significant with respect to natural ambient concentrations than are the water emissions. Figure 14 gives the estimated change in the H_2 concentration because of SPS-related rocket emissions.

Taking both water and molecular hydrogen into account, the exhaust emissions from SPS activities will be about double the total hydrogen content of the thermosphere above about 125 km. This increase is projected based on the calculations, shown in Figs. 13 and 14, of the expected changes in H_2O and H_2 concentrations and the observation that the natural total hydrogen content in the lower thermosphere is of the same order of magnitude as these expected changes. A doubling of the hydrogen content could affect the natural hydrogen cycle and perhaps increase the exosphere density.

Enhancements in the water concentration by factors of 100 or more may notably affect the free electron density above 150-200 km. The electron density is determined by the balance between the rate of electron production (mainly by photoionization) and the rate of electron loss via recombination with ambient positive molecular ions to form neutral molecules. In the normal ionosphere, the principal positive ion above 160-180 km is O^+ . Charge exchange between the O^+ ions and the dominant neutral species N_2 and O_2 leads to the relatively slow formation of molecular ions that recombine with electrons to form neutral species. If water molecules are deposited in the ionosphere, however, charge transfer between H_2O and O^+ occurs much more rapidly than with N_2 or O_2 to produce H_2O^+ and O , and the H_2O^+ ion recombines very rapidly

Table 10. Estimated Natural Water Concentrations above 80 km

Altitude (km)	H ₂ O Concentration (m ⁻³)		Perturbation Factor
	Unperturbed	Change due to SPS	
500	2.67 (3) ^a	3.98 (5)	148
450	6.98 (3)	1.04 (6)	148
400	1.82 (4)	2.70 (6)	148
350	5.33 (4)	7.90 (6)	148
300	1.83 (5)	2.67 (7)	146
250	1.36 (6)	1.91 (8)	140
200	2.56 (7)	3.45 (9)	135
150	1.56 (9)	2.03 (11)	130
125	2.51 (10)	3.20 (12)	127
120	3.90 (10)	4.97 (12)	127
115	1.75 (11)	6.40 (12)	36.6
110	3.11 (11)	7.83 (12)	25.2
105	2.77 (12)	6.83 (12)	2.47
100	5.24 (12)	5.83 (12)	1.11
80	1.15 (15)	1.42 (13)	

^a2.67 (3) denotes $2.67 \times 10^{+3}$.

with electrons (dissociatively). The net result, therefore, of depositing significant amounts of water in the ionosphere is to increase the rate at which O^+ ions and electrons are removed and to shift the overall balance in the direction of *lower* free electron density. The same effect is expected from injections of H_2 , which can function in much the same manner as H_2O . Significant localized regions of electron depletion (or "holes") have been observed following single rocket launches.⁷⁶ The degree to which electron depletion is expected to occur on a global scale will be discussed in later sections. Suffice it to say here that the results of the one-dimensional, steady-state calculation indicate a possibility that detectable, global-scale electron depletions may result from the projected SPS transportation systems.

The rocket exhaust products cool to very low temperatures in expanding to the low ambient pressures above 70-80 km, and a certain fraction (5-35%) of the water vapor will condense to form ice crystals.⁷⁷ Effects of condensation have not been considered in the calculations for Fig. 13.

Some water vapor (and hydrogen) is also emitted at higher altitudes by the COTV* and POTV in their passage from LEO to GEO. The perturbation calculation is outlined in Table 11.

*A Chemical Propulsion System may be used in lieu of battery storage for attitude control during occultations.

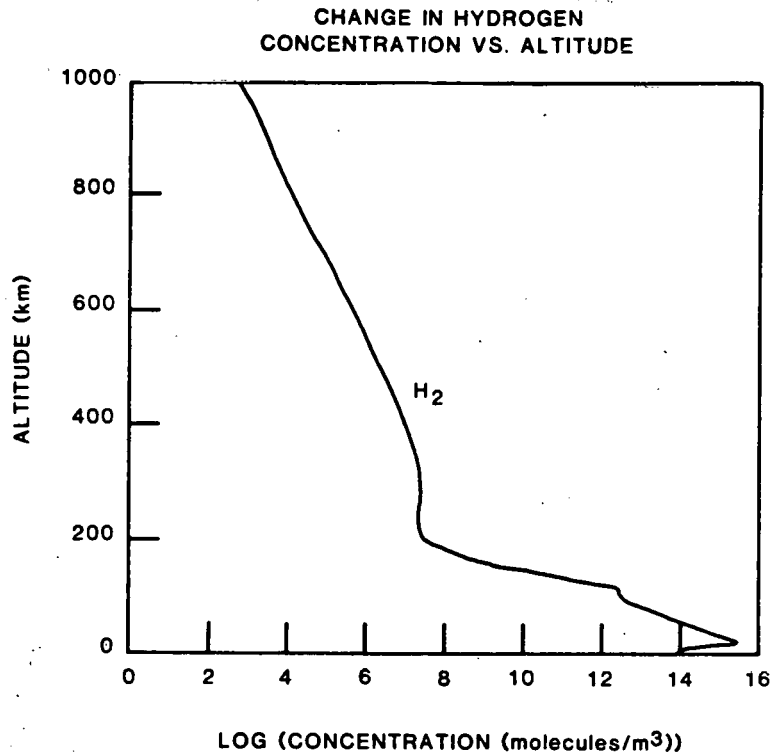


Fig. 14 Estimated Change in Molecular Hydrogen Concentration

3.2.3 Nitric Oxide

During reentry of the HLLV second stage a rather large quantity of air is heated above 2000 K, and a significant amount of nitric oxide is formed. This nitric oxide then "freezes-in" as the air cools, the net result being equivalent to a direct injection. The possible mesospheric significance in terms of a perturbation factor is listed in Table 12. The question of effects due to nitric oxide is discussed later. The principal importance of NO in the 80-100 km altitude region is that it can be ionized by the rather intense solar hydrogen Lyman α line (120.7 nm), and thus leads to an increased level of ionization in this lowest portion of the ionosphere. At lower altitudes, NO may affect the ozone concentration.

3.2.4 Argon Ion Plasma

The COTV uses electrical propulsion, in particular a 1-5 keV argon ion plasma, in traveling between LEO and GEO. Thus, a very large number of argon ions (and the accompanying lower energy neutralizing electrons) are injected into the extremely tenuous magnetosphere. An indication of the attributable perturbation is given in Table 13; note that the "residence time" applies to the processes of de-energization and de-ionization and is rather uncertain.

The magnetosphere is a region in which the overall physics is quite different from that of lower altitudes because of the very low density and

Table 11 Significance of H-atom Emissions
Above LEO

Vehicles responsible -	
POTV main propulsion, COTV attitude control	
Injection rate	= 1.3×10^{32} H atoms/yr
Ambient content	= 10^{17} H atoms/m ² above 500 km (integrated column density)
Residence time	= 8.2×10^6 sec = 0.26 yr (time for photoionization) ⁷⁸
Perturbation factor	= 0.57

Table 12 Significance of NO Injections

Altitude:	55-100 km (peak effect at 70 km)
Vehicles involved:	HLLV Second Stage (reentry)
Injection rate:	6×10^{32} molecules/yr
Ambient loadings:	4.7×10^{32} molecules
Residence time:	4 days \approx 0.011 yr
Perturbation factor:	0.014

Table 13 Significance of Ar Injections

Altitude:	LEO to GEO
Vehicles involved:	COTV main propulsion
Injection rate:	4×10^{32} Ar ⁺ ions/year
Ambient loading:	4×10^{25} ions
Residence time:	1-10 days (time for de-ionization and de-energization)
Perturbation factor:	>>1

relatively high degree of ionization, so that the long-range coupling between charged particles and the geomagnetic field dominates the phenomenology. The disturbances arising from SPS injections are large, but the overall effects of the argon plasma injection are not as clearly defined as are those of lower altitude injections. A recent discussion of the possible effects was given by Chiu et al.⁷⁹

3.3 EFFECTS OF ROCKET EFFLUENTS BELOW 500 km

A perspective on the importance of SPS rocket effluents relative to ambient concentrations is given in Sec. 3.2. In the present section specific

issues regarding these effluents are addressed in detail. Subsections 3.3.1 through 3.3.6 deal with neutral atmospheric impacts. The remaining subsections address effects on the ionized components of the atmosphere.

3.3.1 Composition Changes in Stratosphere, Mesosphere, and Lower Thermosphere

As indicated in the introduction and in Sec. 3.2, injections of water, molecular hydrogen, and nitric oxide would take place in the stratosphere, mesosphere, and lower thermosphere. One- and two-dimensional model calculations were carried out⁸⁰ to estimate the magnitudes of the resulting perturbations to the trace composition of these regions of the atmosphere. A description of the models used may be found in Refs. 80-82; the 1-D model extends to 120 km, and the 2-D model to 90 km. Figure 15 shows the exhaust emission profile used in the calculations, and Fig. 16 shows the NO production due to reentry of an HLLV second stage.⁸⁰ Figure 15 represents the reference design exhaust emission figures² and an estimate of the amount of NO produced by rocket exhaust afterburning.⁹ Figure 16 represents the results of calculations⁸⁰ carried out to estimate the amount of NO produced on reentry. These calculations were carried out for steep, nominal, and shallow reentry trajectories, all of which corresponded to a landing site in southern California; only the results for the "nominal" trajectory are shown. The reentry trajectory is such that the NO production rate depends on both altitude and latitude, and the production rate is shown on Fig. 16 as a function of altitude for latitudes from 10°-35°N.

Stratospheric perturbations were all found to be substantially less than 1% of the natural abundance, in qualitative agreement with preliminary estimates reported earlier.⁹ The effect on ozone, and particularly the integrated-ozone-column density is summarized in the next section.

Figure 17⁸⁰ shows the calculated change in the water vapor concentration for three different latitudes and for altitudes between 30 and 90 km. Also shown is the percent increase in the water concentration as a function of altitude at the assumed launch latitude of 30°N. No water vapor "corridor effect," i.e., enhanced concentration or buildup in a latitude band encircling the globe, exists below about 70 km. Above 70 km, a definite maximum is observed in the water concentration considered as a function of latitude at a given altitude; at 90 km, the change in the water concentration is approximately $6-7 \times 10^8 \text{ cm}^{-3}$ at 30°N, but only about $2 \times 10^8 \text{ cm}^{-3}$ at 10°N and 50°N. At the launch latitude of 30°N, the estimated percent increase in the water concentration increases dramatically above 80 km, is in qualitative agreement with the simple 1-D results discussed in Sec. 3.2, and is already 15% at 80 km. The corresponding global-average percent increase at 80 km is expected to be about 8%.

Figure 18⁸⁰ shows the expected increase in NO concentration in the mesosphere as a function of altitude for various latitudes. The spatial distribution of the NO production rate is somewhat complicated, as shown on Fig. 16 and one cannot think in terms of a single specific reentry latitude; nevertheless, reentry occurs roughly between 15°S and 30°N latitude, with the maximum NO production taking place at between 20-30°N. The resulting corridor effect is clearly seen in Fig. 18, and is much more pronounced than the

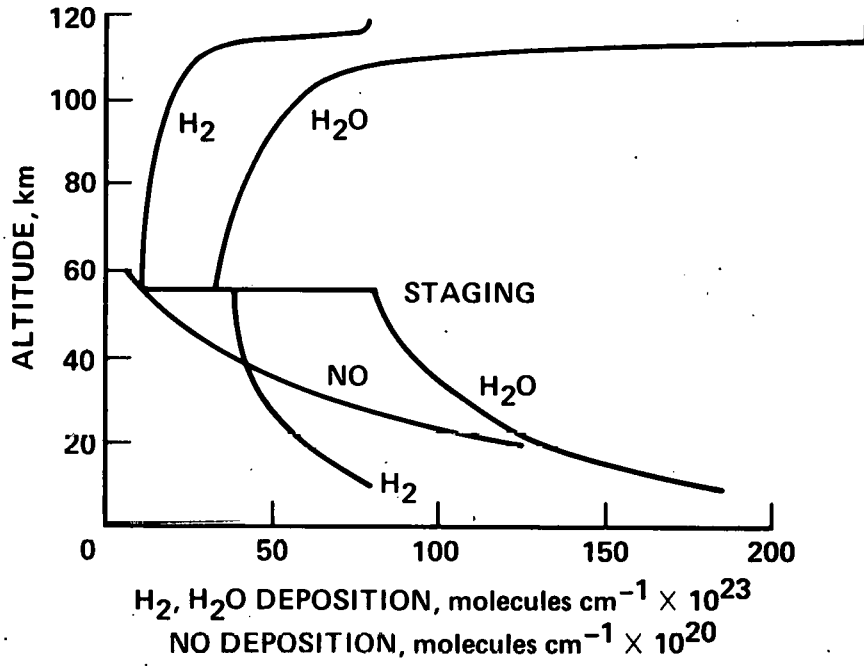


Fig. 15 Rocket Emission Profile

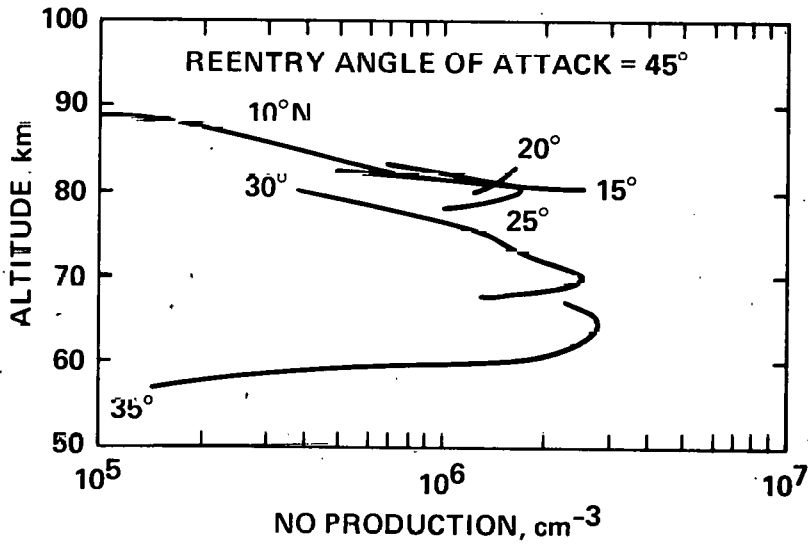
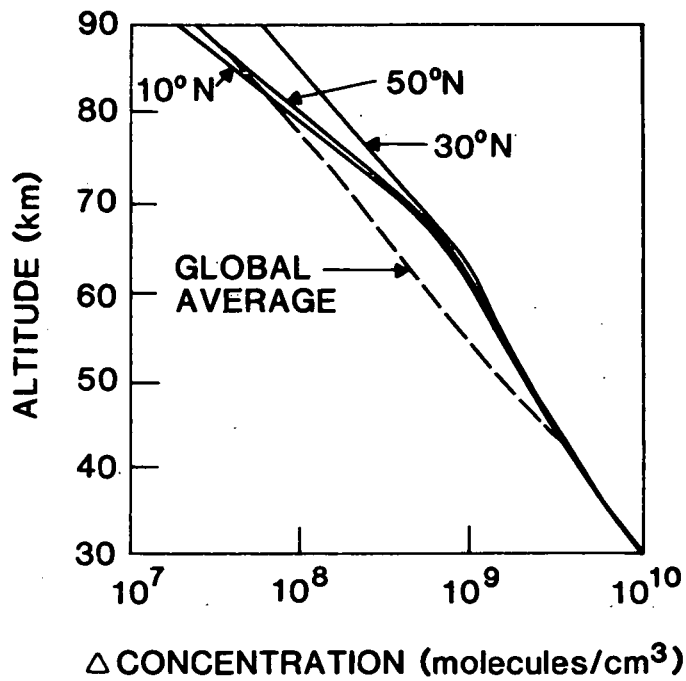
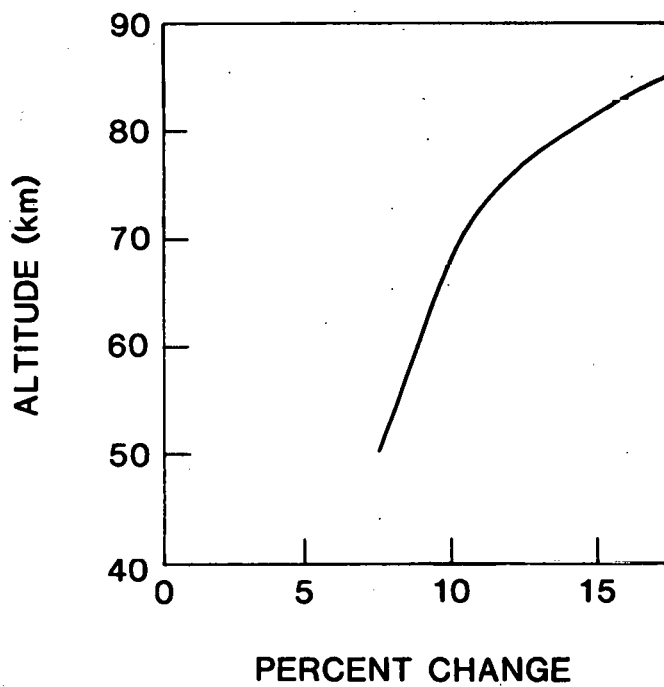


Fig. 16 Estimated Reentry Production of NO



(a)



(b)

Fig. 17 Calculated Change in Water Vapor Concentration at Several Latitudes and Altitudes

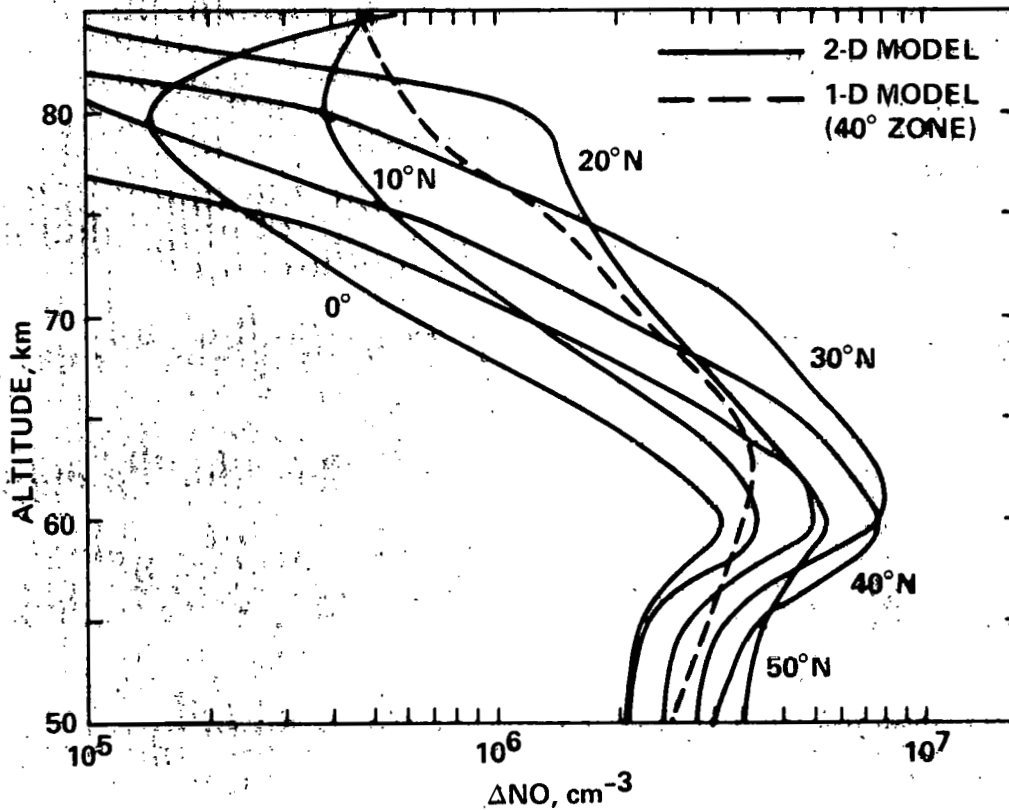


Fig. 18 Estimated Change in the NO Concentration and the NO Corridor Effect

water corridor effect. This difference is attributed to the shorter chemical lifetime (about 4 days) of NO at mesospheric altitudes, compared with about 35-40 days for water.⁸³ As a result, the mesospheric corridor is "narrow" for NO.

The consequences of the mesospheric composition perturbations just described are not expected to be environmentally important. The effect on ozone is discussed in the next section, and the question of noctilucent cloud (or contrail) formation, in the one following.

3.3.2 Ozone Depletion

Since the early 1970s, there has been much concern about the possible alteration of the chemical composition of the stratosphere by a wide variety of natural and man-made phenomena, including volcanic emissions, solar flares, supersonic aircraft, chlorofluoromethane emissions, nuclear weapons testing, and increased biological production of nitrous oxide (N₂O).^{73,84} Possible changes in the concentration of ozone have been the greatest concern because of the role that upper-atmospheric ozone plays in shielding the Earth's surface from solar ultraviolet radiation.

Preliminary calculations based upon 1-D atmospheric models indicated that on a global-average basis, injections of water into the stratosphere and

mesosphere would reduce the total integrated-ozone column by approximately 0.02%, with the greatest change occurring in the mesosphere.⁹ More recent calculations,⁸⁰ involving both 1D and 2-D models, including both water injections and NO deposition in the stratosphere due to exhaust afterburning indicate that the expected change in the ozone column is approximately -0.015%, globally averaged, in agreement with the earlier results. The effect of NO deposition owing to reentry of the HLLV second stage on the ozone column is an *increase* by about 0.02%, globally averaged. Very little corridor effect is present in the results of the water vapor calculations, except at altitudes above 70 km, but a slight corridor effect may be seen in the results of the reentry-produced NO calculations as shown in Fig. 19. A change in ozone-column density of the order of magnitude indicated by these results would be undetectable with available measurement techniques.

It should also be pointed out that the predicted effects of reentry-produced NO on the ozone-column density are subject to considerable uncertainty, and that if realistic error bounds are attached to the predicted values as shown in Fig. 19, they would include the possibility of a net zero or a small negative effect. In addition, new values of the rate coefficients for $\text{OH} + \text{HNO}_3$, $\text{OH} + \text{H}_2\text{O}_2$, $\text{HO}_2 + \text{HO}_2$, and $\text{OH} + \text{HO}_2$ will probably alter the values of the change in ozone computed by Whitten, et al.⁸⁰ The smaller ozone gain is expected to become a small ozone loss (Whitten, private communication, Aug. 1980).

3.3.3 High-Altitude Cloud Formation

Formation of artificial noctilucent clouds following rocket launches has been observed on several occasions.⁸⁵⁻⁸⁷ Noctilucent clouds are very

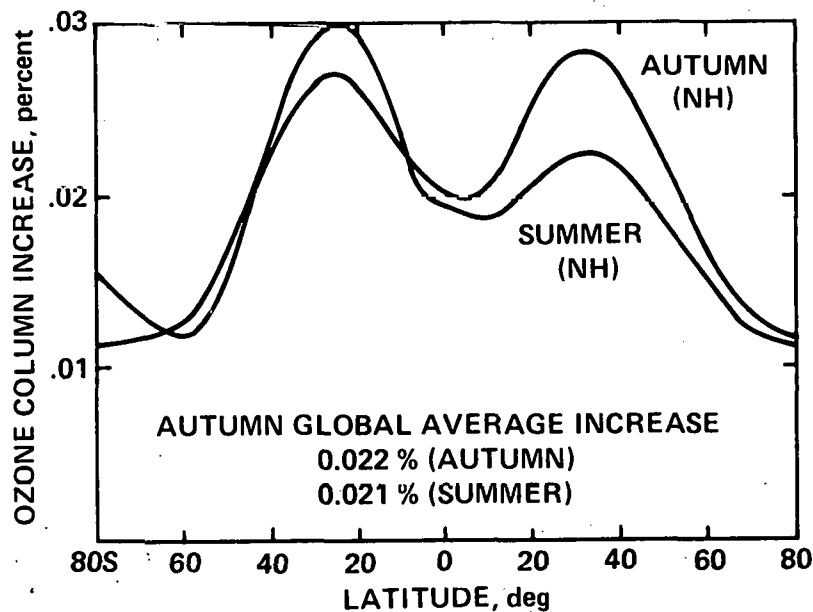


Fig. 19 Ozone Column Density Corridor Effect Due to Reentry NO

tenuous, optically thin clouds occurring at or near the mesopause, the location of the minimum in atmospheric temperature marking the upper boundary of the mesosphere, at an altitude of about 80-85 km. Natural noctilucent clouds occur only during the summer months at latitudes above approximately 65°N, and can be observed only with a glancing solar angle due to their optical thinness. The significance of the observations of such clouds following rocket launches is in part that they were made at latitudes of 45°N in winter on two occasions,⁸⁵ 35°N in late spring,⁸⁶ and 35°N in autumn,⁸⁷ and indicate, therefore, a potential for enhanced noctilucent cloud formation at mid-latitudes from SPS-related rocket launches. That such a potential exists may be emphasized perhaps by noting that on two occasions,⁸⁵ the artificial cloud was apparently produced or triggered by a relatively small research rocket that reportedly deposited only 260 gm of water in the 80-90-km-altitude range, whereas an HLLV second stage will deposit approximately 1.3×10^7 gm of water in the same altitude range. It may be significant that each of the four reported clouds was associated with the launch of a solid-fueled, rather than a liquid-fueled, rocket, although this observation may be due simply to the prevalence of solid-fueled rockets and the very small number of sightings.

It seems highly probable on this basis alone that transient noctilucent clouds or, perhaps more correctly, persistent contrails will be formed in association with individual HLLV or PLV launches. Theoretical calculations⁸⁰ support this conclusion and indicate that the increase in optical depth following an HLLV launch may approach 25 along the trail axis 1 hr after launch. (Optical depth is defined as the negative of the natural logarithm of the ratio of the observed light intensity at a point to the light intensity incident on the absorbing medium in which the observation is made. An optical depth of unity implies that 1/e of the incident light has been absorbed before reaching the observer, or equivalently, that 63% of the incident light is transmitted.) An optical depth of 25 implies that the fraction of light transmitted along the cloud axis is only 1.4×10^{-11} . However the cloud comprises ice particles that quickly disperse and evaporate, and the expected cloud lifetime is less than one day. The optical depth perpendicular to the cloud axis would be much smaller.

Figure 20 shows the results of a calculation of the dynamics of the contrail of an HLLV in the altitude range near the mesopause.⁸⁰ The contrail diffuses from a cross-sectional area of about 1 km² immediately after passage of the rocket to 900 km² after a period of a day. The corresponding evolution of the associated optical depth is shown. Immediately after injection into the atmosphere, a period exists during which homogeneous nucleation occurs and, because of the high super-saturation, the particles grow rapidly to radii on the order of 1 μm. Nucleation and growth cease after the water vapor concentration has fallen to near the saturation point, and turbulent dispersion of the contrail becomes the dominant process, with coagulation and sedimentation being important secondary processes. As the expansion of the contrail proceeds, the water vapor concentration remains essentially constant at saturation because of evaporation of the ice particles. Eventually, a point is reached at which evaporation proceeds very rapidly, and the contrail consists mainly of a region containing no ice particles and in which the water vapor concentration is just at saturation. This point is determined by the ambient temperature; hence after some time, the contrail is confined to the coldest region of the atmosphere, and disappears entirely when that region becomes unsaturated.

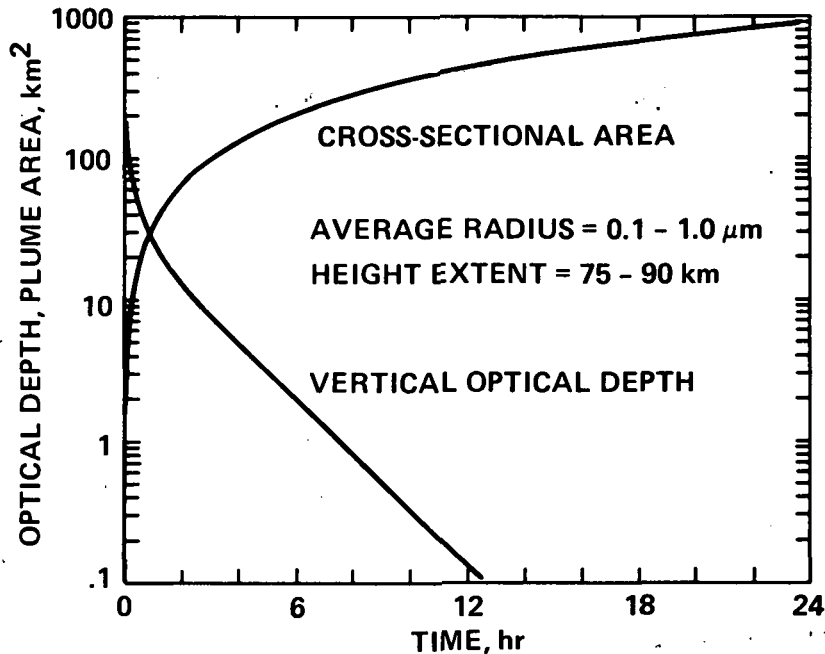


Fig. 20 Optical Depth in an HLLV Contrail

The calculations and the results shown in Fig. 20 indicate that a sub-visible cloud could persist for about a day before water saturation is reached at all altitudes, and a visual noctilucent cloud display is therefore possible at the first dusk or dawn following an HLLV launch.

Based upon the theoretically expected increase in the globally averaged water vapor concentration at 80-90 km,⁸⁰ an increase of only 1×10^{-4} in the optical depth at visible wavelengths is expected in the natural noctilucent cloud zone. If the optical depth were increased by the same amount over the entire earth, the estimated temperature change would be only -1×10^{-3} K, compared with the accepted threshold for perceptible climate change of 0.1 K.⁸⁰ Thus, although transient artificial noctilucent clouds are expected to form in association with individual rocket launches and may, on occasion, give rise to colorful displays under the right viewing conditions, no global-scale increase in optical depth is expected and no detectable climatic effects are expected.

3.3.4 Global Hydrogen-Cycle Changes

Hydrogen, chemically bound, in the form of water and molecular hydrogen will be deposited in significant quantities by the HLLV-second-stage main burns, as already mentioned. These molecules will eventually undergo photodissociation as well as chemical, charge exchange, and dissociative-recombination reactions that will liberate hydrogen atoms. Preliminary calculations by Zinn et al.⁸⁸ and Whitten et al.⁸⁰ of the source terms, suggest that this source of H-atoms could lead to a doubling of the natural upward flux of

hydrogen into the thermosphere and exosphere. This problem has not been addressed in detail by either group of investigators. While the mass of these H-atoms would probably not substantially increase the neutral atmosphere density in the thermosphere, it could result in a significant increase in the total mass density above about 800 km where hydrogen is the dominant constituent. Two major sources of uncertainty in the H-atom balance problem are the magnitude of the natural upward flux of H-atoms and how an enhancement of this flux would affect the global escape rate. While this natural upward flux is diffusion limited below 100 km, it is governed by other mechanisms, including charge exchange and Jeans escape, at higher altitudes.

Several current reviews are available that discuss the natural hydrogen cycle.^{74,75,89-91} If these additional H-atoms from the SPS rocket exhaust were to accumulate and significantly increase the density above 800 km, important consequences could result. Since satellite drag measurements have been used to detect variations in atmospheric density at altitudes as high as 1100 km or more, it is plausible that large chronic changes in density in that same range could slowly alter satellite orbits, especially sun-synchronous orbits near 900 km. Although rather speculative, it has also been suggested that substantially increased upper thermospheric densities may alter wind patterns at those altitudes and also may affect ionospheric-magnetospheric coupling processes, especially those involving the precipitation of high energy particles. Such processes are believed to be a principal source of energy to the high latitude thermosphere. The problem of altering the natural H-atom cycle requires further analysis.

3.3.5 Density and Temperature Changes

As mentioned above, injections of H-atoms in the form of H₂O and H₂ molecules could lead to enhancement of exospheric densities above 800 km if the natural H-atom escape rate did not increase. Density changes below this altitude due to injection of exhaust mass are not likely to be very large because of the relatively massive neutral atmosphere. However, mechanisms have been suggested for the alteration of the upper atmospheric temperature that could lead to a number of consequences, including thermospheric density changes. Unfortunately, these mechanisms involve compositional changes, estimates of which have only just begun to become available. Hence, temperature and density changes will have to await future analysis. Of particular interest are possible temperature effects of the addition of the IR active H₂O molecules and OH radicals produced from the H₂O and H₂ exhaust molecules. It has been suggested that the presence of these IR-active species could lead to a net radiative cooling near the mesopause. This could lead in turn to the formation of more persistent noctilucent clouds (see Subsec. 3.3.3).

Some experimental evidence for a cooling effect has been reported by Chenurnoy and Charina.⁹² They studied variations in hydroxyl band IR emission before, during, and after a noctilucent cloud display and concluded that higher H₂O concentrations before formation lead to enhanced OH⁻ emission, causing local cooling. After formation, the water vapor is condensed into cloud particles that, according to these authors, reduce the IR emission.

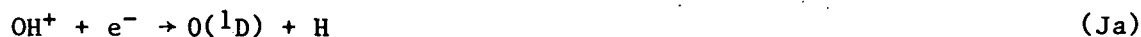
Another mechanism for temperature and density change arises in the upper ionosphere and plasmasphere. Chiu et al.⁹³ have estimated that a substantial fraction of the energy carried by the ion beam from a COTV would be converted into thermal energy in the mid-latitude ionosphere. This conversion would result in a marked temperature increase and a transformation of the plasmasphere from a region dominated by H-atoms to one dominated by Ar and O atoms. Consequences of this change are discussed in a subsequent subsection.

3.3.6 Airglow Emission Changes

Natural airglow emissions occur over a broad range of the electromagnetic spectrum from the far IR to the extreme ultraviolet (EUV). Some regions of this spectrum are used for satellite-borne, remote sensing, and surveillance systems, and, therefore, account must be made of the natural background radiation levels and their natural fluctuations. The release of H₂O and H₂ molecules in the mesosphere and thermosphere will lead to interactions with ambient neutrals, ions, and electrons, producing reaction products in excited states that upon decay will emit photons in the airglow spectrum. The following reactions have been proposed:

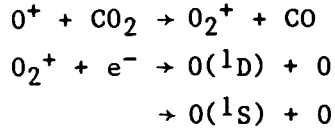


Decay of the excited states of oxygen atoms produced in reactions (Ha) and (Hb) result in the emission of red lines (630 and 636.4 nm) and a green line (557.7 nm), respectively. The OH radicals produced in reaction (Hc) could be vibrationally excited, in which case decay would yield OH band emissions. Another source of the red and green lines could be the reaction sequence:



Again the reactions (Ja) and (Jb) result in excited states of the oxygen atom but the reaction (Hc), which is supposed to be more probable, leads to the ground state of the oxygen atom and, therefore, no decay photon is emitted.

The artificial airglow emissions produced by chemicals released in the upper atmosphere have been observed in the Lagopedo experiments of 1977^{94,95} and in the monitoring of the HEAO-C Satellite launch^{96,97} in September, 1979. In both cases, strong emissions were observed at 630 and 557.7 nm as expected. However, in the case of the Lagopedo experiments the chemical release was produced by an explosive discharge whose products include H₂O, H₂, and CO₂. The reaction sequence



was believed to be the dominant source of the red and green lines. In the case of the HEAO-C satellite launch, the burning of the second-stage rocket engine produced only H₂O and H₂. The only other emission detected in the Lagopedo experiments was a weak signal at 666 nm. However, rather than being produced from the decay of excited OH radicals produced by reaction (Hc) as first expected on theoretical grounds, it was produced by the scattering of sunlight by ice crystals. The 666-nm line was not observed in the HEAO-C monitoring experiment. However, emission in the 2.9 nm OH band was observed in the latter experiment and is not fully understood at present.

Airglow emission intensities (red and green lines) of the order of 10 kiloRayleighs were observed over about a 30-minute period, covering an area of 10⁵ to 10⁶ km², following the launch of the HEAO-C satellite. Since the number of exhaust molecules injected into the 200-500-km region during this launch is essentially equal to that of an HLLV circularization burn, similar airglow enhancements can be expected from such burns. The impact of such airglow enhancements on the performance of remote sensing equipment and astronomical observations is not known for certain; and although probably not very important, the impact should be determined. Emissions in near-UV or IR regions might be more significant for remote sensing (see "Artificial Airglow Emissions", Section 3.4.5). For example, the observed emission at 2.9 nm may be important to remote sensing systems. The nature of this emission requires further investigation. Unfortunately, no measurements have been made that would indicate the extent of artificial airglow emissions produced by an HLLV-second-stage burn. It is important to note that the branching ratios for the abovementioned reaction sequences have not been measured in the laboratory.

3.3.7 D- and E-Region Effects

The participants in the Ionospheric and Magnetospheric Workshop¹⁰ held in August 1978 suggested that water injection will tend to convert the region from 80-100 km from one dominated by light molecular ions to one dominated by heavier water-cluster ions that can recombine more rapidly with the free electrons. This action will lead to a reduction in the daytime free-electron density. The idea has since received further support both by participants at the La Jolla Workshop¹¹ and by calculations recently performed by Forbes.⁹⁸ It has most recently been investigated by Whitten et al.⁸⁰ Forbes has estimated that water vapor concentrations of approximately 100 ppmv (parts per million by volume) would lead to a near complete conversion to water cluster ions within a cylindrical volume between 70 and 100 km altitude and having an area of 20,000 km². Combined with the loss of electron production caused by screening of UV radiation by water molecules, Forbes estimates a 75% reduction in the daytime free-electron density over this same volume. If such depletions were indeed limited to such small volumes, they would not be expected to be of much concern. The size of the affected region and the possibility for cumulative effects is critically dependent on a number of parameter values used by Forbes, including neutral wind speeds, diffusion rates, and the photolytic lifetime of water. Unfortunately, Forbes used

a photolytic lifetime for water of only two days. The accepted value of this parameter is substantially greater and is strongly dependent on altitude. For example, Zinn et al.⁸⁸ have examined the photolytic lifetime of H₂O in the upper atmosphere and have determined that it varies rapidly with altitude between 50 and 100 km (in agreement with other investigators) and approaches an asymptotic value of four days near 125-150 km. The four-day lifetime assumes that the 24-hr average photodissociation rate coefficient is one-half the peak value at noon at 150 km.

Zinn et al.⁸⁸ obtain a lifetime for water of 36 days at 85 km and about 15 days at 100 km. Assuming a longitudinal transport rate quoted by Forbes,⁹⁸ this data would suggest the possibility that a launch rate of once or twice per day could lead to cumulative effects. That is, that effluents deposited by one rocket could be transported at the launch latitude around the globe (in about 1-2 weeks) in time to be augmented by another rocket launch before being completely removed by chemical or transport processes. Of course, the concentration of effluents would be greatly diminished by horizontal and vertical diffusion during the time required for transport around the globe. Whitten et al.⁸⁰ have examined the extent of free electron depletion in the region from 70-100 km caused by ion-water cluster formation. On the *long-term, global scale* they found that for altitudes below 80 km the water vapor concentration was increased by only about 7-8%, or less, above ambient, and since the clustering is already rapid in that region, the change in ionic composition and free electron density was quite small. Above 80 km, even though they predicted increases in water vapor concentration $\geq 20\%$ near the launch latitude, they still did not find marked reductions in free electron content, because, according to the authors, the water clustering to ions was found to be too inefficient. They did not report on the effect of enhanced-ion-water clusters on the short-term, local scale. However, one would expect significant short-term, local-scale electron depletions in accordance with Forbes' result cited above.

Another potential effect of the large injections of water is a significant enhancement of the populations of OH radicals and H atoms. The hydroxyl radicals are important sources of airglow emissions in the near infrared. The H atoms will participate in mesospheric chemistry to some extent, but most will tend to diffuse into the thermosphere, as mentioned earlier. A qualitative estimate by Forbes⁹⁸ suggests that this upward flux of H atoms will result in about a 10% or more attenuation of Lyman α radiation on a global scale. This attenuation would probably produce a small chronic reduction in D- and E-region *daytime* ionization. On the other hand, increased concentrations of H atoms in the thermosphere and exosphere will scatter Lyman α radiation at nighttime, which will cause an increase in *nighttime* D- and E-region ionization. It is not known what effects such long-term changes in the diurnal pattern of ionization would have. However, it has been noted that such patterns are related to electromagnetic coupling between the E- and F-regions.

A possible compensating effect on the D- and E-region plasma density could result from the production of NO by reentering space vehicles and debris. Since ionization of naturally occurring NO by solar Lyman α radiation is the main source of free electrons in the D- and lower E-region, enhanced NO concentration is expected to result in increased free-electron content. Whitten et al.⁸⁰ performed two preliminary case studies of the long-term

global increases in the NO and free-electron concentrations. The Case 1 study consisted of calculations using their one-dimensional photochemical model and provided a lower bound on the expected results. Case 2 utilized their two-dimensional photochemical model and provided an upper bound on the expected results. The results are shown in Table 14. Note that the maximum increase in e^- is about 18% and that there is a slight decrease in e^- above 85 km and at 60 km. The decreases are caused by the ion-water clustering, which slightly enhances the recombination rate over the increased production rate because of increased NO concentration. A longterm increase of up to 18% in the e^- concentration in a band surrounding Earth may not be insignificant and should be examined further in the future.

Since the D-region causes reflection of very low frequency (VLF) radio waves (3-30 kHz) and attenuation of high frequency (HF) radio waves (3-30 MHz), changes in e^- will result in corresponding changes in radio wave propagation properties. Whitten et al.⁸⁰ have estimated that the corresponding increase in HF attenuation in the frequency range of 3-30 MHz is about 10% during the daytime. Since much larger increases in attenuation that create serious communication problems are encountered during large-scale geophysical disturbances (e.g., aurora, solar proton events, and sudden ionospheric disturbances), it is doubtful whether a 10% increase in D-region absorption would be very serious. However, the question does deserve further attention. The effect of an 18% increase in the electron density on extremely low frequency (ELF) radio wave propagation was also examined by Whitten et al. They estimated the effects to be negligible.

Short-term, local effects of NO production by reentry are expected to be quite large, but confined to the immediate neighborhood of the reentry trajectory. Note that since such trajectories will not in general overlap launch trajectories, one would not expect short-term local compensating

Table 14 Predicted Ionospheric Perturbations from Nitric Oxide Deposition (from Ref. 80)

Alt. (km)	Unperturbed		Perturbed				Collision Frequency
	[NO]	[e]	Case 1		Case 2		
	cm ⁻³	cm ⁻³	[NO]	[e]	[NO]	[e]	
100	1.0 (8) ^a	4.5 (4)	1.0 (8)	4.5 (4)	1.0 (8)	4.5 (4)	3.2 (4)
90	4.8 (7)	7.5 (3)	4.8 (7)	7.4 (4)	4.8 (7)	7.4 (3)	1.9 (5)
85	1.5 (7)	4.1 (3)	1.6 (7)	3.9 (3)	1.6 (7)	3.9 (3)	4.5 (5)
80	1.0 (7)	8.0 (2)	1.1 (7)	8.3 (2)	1.3 (7)	8.7 (2)	1.1 (6)
75	1.0 (7)	6.1 (2)	1.3 (7)	6.5 (2)	1.5 (7)	6.8 (2)	2.4 (6)
70	1.0 (7)	1.7 (2)	1.4 (7)	1.9 (2)	1.9 (7)	2.3 (2)	5.3 (6)
65	9.6 (6)	2.9 (1)	1.7 (7)	3.4 (1)	2.4 (7)	3.8 (1)	1.1 (7)
60	6.5 (7)	7.6 (0)	7.3 (7)	7.2 (0)	8.5 (7)	7.2 (0)	2.2 (7)

^a1.0 (8) = 1.0 × 10⁸

effects such as arise in the long-term global-scale situation discussed above. Table 15, from Whitten et al.,⁸⁰ shows estimates of the NO and corresponding e^- number densities as a function of altitude and time after reentry. Comparing the perturbed e^- densities with the unperturbed daytime values (last column, Table 15), it appears that the density increases are up to almost two orders of magnitude even after 24 hr. Comparing the perturbed NO densities in Table 15 with the unperturbed values given in Table 14, it is seen that after 24 hr the perturbed values range from two to three orders of magnitude greater. However, the authors caution that their estimates are based on a very simplistic modeling approach and, in light of the magnitude of the increases, require a more sophisticated examination.

In summary, preliminary calculations suggest that the long-term global changes in e^- in the D- and E-regions will probably be relatively small ($\leq 10\%$) owing to a combination of enhanced recombination by increased positive ion-water clustering and enhanced production resulting from increased NO concentrations caused by reentry. Changes of this magnitude probably could not seriously affect radio wave propagation. Changes in Lyman α intensity in both the daytime and nighttime D- and lower E-region conceivably could have a somewhat greater effect, but they have not been investigated in detail. On the other hand, short-term local changes due to water injection (promotion of ion-water cluster formation) and NO production by reentry appear to be very large and have lifetimes of the order of days. It remains to be determined through more sophisticated theoretical studies and field experiments focused on actual rocket launches whether these large magnitude effects are real, what their actual temporal and spatial scales are, and what consequences they would have on radio wave propagation.

The only experimental evidence pertaining to VLF propagation effects caused by a rocket launch is that obtained during the monitoring of the HEAO-C Satellite launch by an Atlas-Centaur rocket in September 1979. One of the groups of participants in that monitoring campaign, Meltz and DarRold,⁹⁹ observed VLF propagation (10-13 kHz), using an operational radionavigation

Table 15 Predicted Nitric Oxide and Electron Concentrations near Reentry after 6 Hours and 1 Day

Alt. (km)	Densities (cm^{-3})				Unperturbed n_e from Table 14
	NO		e^-		
	6 Hours	1 Day	6 Hours	1 Day	
100	2.0 (+9)	6.0 (+8)	6.0 (+4)	4.8 (+4)	4.5 (+4)
90	9.0 (+10)	9.0 (+9)	2.7 (+5)	8.6 (+4)	7.5 (+3)
80	6.2 (+10)	1.2 (+10)	1.8 (+5)	7.3 (+4)	8.0 (+2)
70	1.5 (+11)	2.5 (+10)	2.6 (+4)	8.0 (+3)	1.7 (+2)
60	8.0 (+10)	1.5 (+10)	1.6 (+1)	8.5 (+0)	7.6 (+0)

system "OMEGA" before, during, and after the launch. Such a system utilizes the earth-ionosphere waveguide mode of propagation. For such a mode the observed VLF phase delay between the transmitter and receiver is sensitive to changes in the D- and lower E-regions. For example, a sudden ionospheric disturbance (SID) such as caused by an X-ray burst from a solar flare will change the phase delay. A similar change of phase could be caused by the deposition of rocket-exhaust vapor if the e^- density or collision frequency of electrons with neutrals were changed. On the basis of their observations, Meltz and DarRold suggested that the exhaust products from the Centaur stage of the rocket (injected between 211 and 501 km) descended into the upper D-region and altered its composition. The observed-phase advance suggests that the electron density and/or collision frequency was increased and in turn either increased the phase velocity or lowered the effective reflection height. A satisfactory theoretical explanation of these phenomena has not yet been put forward. The investigators observed VLF-phase anomalies up to 24 hr after the rocket launch and claimed that they were caused by the launch. There were two solar flares that produced characteristic phase-change signatures during this same period. The principal evidence in support of the claim that the rocket launch produced anomalies over and above those caused by the solar flares is the fact that of the two transmitter-receiver propagation paths used in the observations both showed the influence of the solar flares, but the one that passed near the rocket trajectory showed a distinctly different phase-delay pattern that continued to appear anomalous during the evening following the launch when no solar flares occurred. If such phase anomalies are indeed a characteristic feature of rocket launches, then the extent to which they affect navigation should be examined further.

A somewhat more subtle issue of concern regarding potential D- and E-region effects relates to the fact that the high-latitude conductivity distribution is especially important because the high-latitude ionosphere completes the electrical circuit that couples the auroral zone to the outer magnetosphere. The electrical currents that flow through this circuit undergo large fluctuations during magnetic substorms and have been known to induce damaging current surges in long telephone and power transmission lines. Alteration of the auroral zone conductivity (in the E-region) could modify the morphology of this current system and influence the occurrence or intensity of terrestrial current surges, perhaps moving them to more populated areas.

Finally, Markson^{100,101} has suggested that since the lower ionosphere is part of the global atmospheric electric circuit through which currents are driven by thunderstorms, large-scale perturbations in the conductivity, especially if they reach down to the middle stratosphere, may influence thunderstorm processes and therefore weather and climate.

The present state of knowledge regarding the alteration of the morphology of the auroral current system and Markson's suggestion does not permit an assessment of these effects at this time.

3.3.8 F-Region Ionospheric Holes

Rocket effluents (such as H_2O and H_2) react rapidly with the dominant F₂-region ions (O^+) to form molecular ions (e.g., H_2O^+), which recombine very rapidly with free electrons. The net results are the removal of elec-

tron-ion pairs at a rate 1000 times faster than normal, the production of prompt, intense chemical airflow, and the release of large numbers of hydrogen atoms. The rapid loss of electron-ion pairs results in an ionospheric hole that extends far beyond the local source of injected molecules and may be extended to the conjugate ionosphere (i.e., cause a hole at the opposite end of the geomagnetic line that passes through the initially depleted region).

The 1973 launch of Skylab with a Saturn V rocket injected a substantial amount of rocket exhaust into the F₂-region of the ionosphere. The trajectory of that rocket accidentally crossed radio-signal ray paths connecting the ATS-3 communication satellite with the ground-based observatory at Sagamore Hill in Massachusetts. The deep depression in the total electron content (TEC) observed at Sagamore Hill after the rocket passed the intersection point is shown in Fig. 21. Based on observations made at four other ground-based observatories whose lines-of-sight with ATS-3 also were intercepted by the resulting depleted region, it was reported that the ionospheric hole had a radius of about 1000 km and lasted for about 4 hr.⁷⁶ Because the observations were not planned, however, they were somewhat incomplete.

The first successful attempt to produce a mid-latitude, ionospheric depletion under controlled experimental conditions occurred in September 1977. Project "Lagopedo"⁹⁴ involved two rocket-borne experiments. Each rocket carried 88 kg of high explosives and an instrument package that was separated from the rocket prior to detonation of the explosive. The detonation products included about 30 kg of H₂O, 16 kg of CO₂, and 20 kg of N₂. These experiments generally confirmed theoretical predictions of a signature effect, i.e., an initial period of several seconds during which the ambient plasma was swept away by the rapidly expanding detonation cloud over a distance of less than

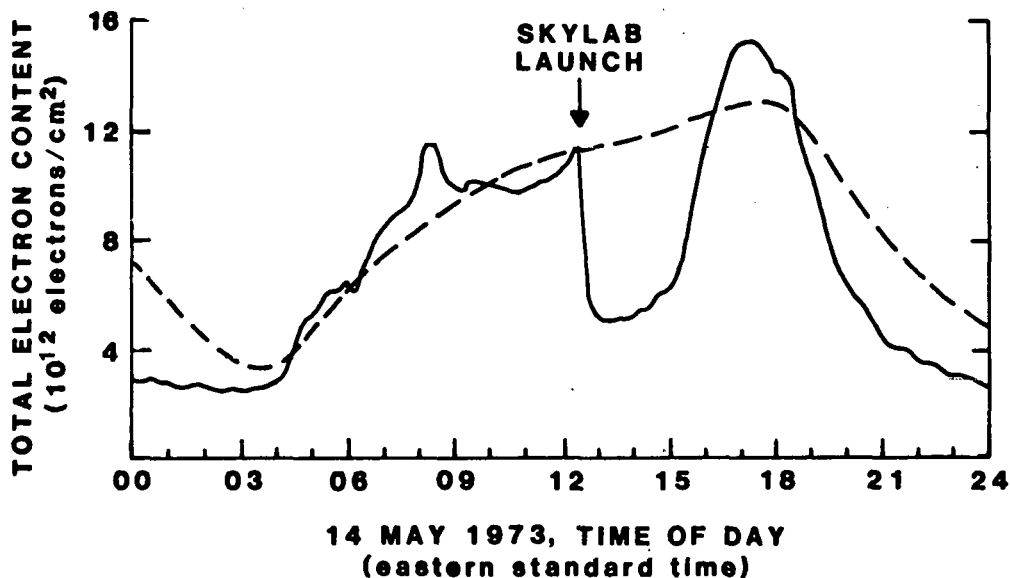


Fig. 21 Sudden Reduction in the Total Electron Content

1 km, followed by a much larger-scale, plasma-depletion process lasting for at least 30 min and extending to a radius of 30 km or more. In addition to the rocket-borne instrument package, which, unfortunately, only sampled the disturbed area for 2-3 min, ground-based instruments monitored the hole for about 30 min until the satellite beacon being used as a diagnostic device was turned off.

A major question that arose from these two experiments was the importance of the suppression of the participation of the water molecules in the electron-ion removal process through initial formation of ice crystals, followed by gravitational settling to lower altitudes where the presence of the water was unimportant. This was an especially important question for prediction of SPS effects since H_2O and H_2 are the major exhaust products of the HLLV, PLV, and POTV, all of which inject such products above the F_2 -region. If most of the water is frozen into ice crystals as originally suggested by preliminary analysis of the Lagopedo data, then the crystals would fall through the F_2 -region more rapidly than water vapor molecules and would not contribute to the formation of ionospheric holes, according to Zinn et al.⁸⁸ If on the other hand, most of the water molecules remained in the vapor state, then they could substantially increase the extent of the ionospheric holes. Furthermore, preliminary analysis of the ionospheric hole caused by the launch of Skylab in 1973 suggested that the hole should have lasted much longer than it appeared to. Consequently, in an attempt to make the model predictions correspond to the apparent lifetime of the hole created by Skylab, it was assumed that the water molecules did not participate in the hole formation process because of ice crystal formation.

Subsequently, however, it was learned that even complete suppression of the water molecules would not bring the theoretical model predictions into agreement with the Skylab observations. Something else was happening that had not been previously considered. In addition, Bernhardt and his collaborators⁷⁷ made an investigation of the extent to which water vapor molecules form ice crystals in the upper atmosphere and found in particular that the degree of condensation depends on the initial conditions of release. For explosive, spherically symmetrical releases such as the Lagopedo releases the initial high pressure and temperature of the exhaust cloud does not permit reliable estimation of the degree of condensation. For water molecules released from a rocket nozzle the initial exit pressure is about 10^5 times lower and the degree of condensation has been calculated to be 7, 24, and 31% for the Space Shuttle OMS engines, Saturn J-2 engines, and Space Shuttle Main Engines (SSME), respectively. Note that the second stage of the proposed HLLV would use 14 SSMEs. Hence, the range of condensed mass fractions agreed upon by workers in the field is about 5-35% for rocket nozzle releases. This fraction would not significantly influence the extent of the F_2 -region ionospheric hole.

The previously unaccounted for factor in determining the lifetime of the ionospheric hole created by the Skylab launch vehicle effluent that was alluded to above, was believed to be the thermospheric wind pattern that, according to Zinn et al.,⁸⁸ probably moved the depleted region out of the line of sight of the observatories, yielding the apparent lifetime of ± 4 hr. Based on current estimates, including the probable effects of such transport winds, Zinn et al. conclude that the hole probably did not recover until the following day, some 16 hr after launch. Unfortunately, no actual thermospheric wind observations were made at the time of the Skylab launch.

Hence, through 1978, the sum total of data relevant to ionospheric holes consisted of an incomplete set of accidental observations made during the Skylab launch of 1973 and the planned but rather short-term and small-scale observations of the two explosive releases of the Lagopedo experiments. The planned launching of the HEAO-C Satellite in September 1979 did provide an excellent opportunity¹⁰² to observe a large-scale ionospheric modification caused by an LO_2/LH_2 -fueled-rocket-engine burn. Through careful planning and coordination, a monitoring campaign involving some 17 separate research groups plus the cooperation of approximately 150 Ham Radio Operators¹⁰³ and several commercial station operators and listeners provided the most complete case study of a large-scale ionospheric modification to date.¹⁰⁴ Figure 22 shows the ground track of the rocket launch trajectory through the ionosphere and several of the ray paths to the SIRIO, ATS-3, and ATS-5 communications satellites used to monitor the ionospheric disturbance. The ionospheric hole is shown as a dashed, kidney-bean-shaped area. The Centaur stage of the Atlas/Centaur launch vehicle burned from 211-501 km altitude and injected about 7×10^{29} H_2O and H_2 molecules. For comparison, the HLLV circularization burn at 477-km altitude will inject about 9×10^{29} molecules. Except for the relatively minor difference in the altitude distribution of the injection, the HEAO-C launch is a good simulation of the HLLV circularization burn. Consequently, the results of the monitoring campaign,¹⁰⁴ which are summarized below, are quite useful for assessing the short-term effects of a circularization burn.

3.3.8.1 Hole Morphology

The hole formed within a few minutes of the passage of the rocket. Nearly a 100% depletion of electrons was observed within 75 km of the trajectory. At its maximum extent, the hole was about 2500 km long, 800 wide, and 300-400 thick. The hole remained until shortly after sunrise some 4 hr after launch. During that time, it remained at roughly the same latitude for the first 2 hr or so and then drifted slowly northward. The hole did not appear to be symmetrical in its north-south extent about the trajectory.

3.3.8.2 Airglow Enhancement

Airglow emissions offer a useful diagnostic source for determining hole morphology, rates of formation of excited states of reaction products, and potential interference with remote sensing and surveillance systems. Emissions from excited oxygen atoms and hydroxyl radicals were of greatest interest because the corresponding reactions play major roles in the depletion process but have never been measured in the laboratory. The oxygen red line and to a lesser extent the green line were of much greater intensity but of shorter duration than anticipated. This activity was consistent with the greater than expected initial growth rate of the hole. The natural red airglow intensity at 6300Å was enhanced from a pre-launch value of 100 to about 8300 Rayleighs (viewed vertically upward). The intensity of the green line at 5577Å increased from 300-900 Rayleighs. The red line enhancement peaked within a few minutes of the rocket's passage and then dropped off by a factor of 10 within about 20 min, as reported by Stone and Zinn⁹⁷ and by Kofsky et al.⁹⁶ In general, the model calculations of Zinn et al.,⁸⁸ both for airglow and hole morphology, are in relatively good agreement with the

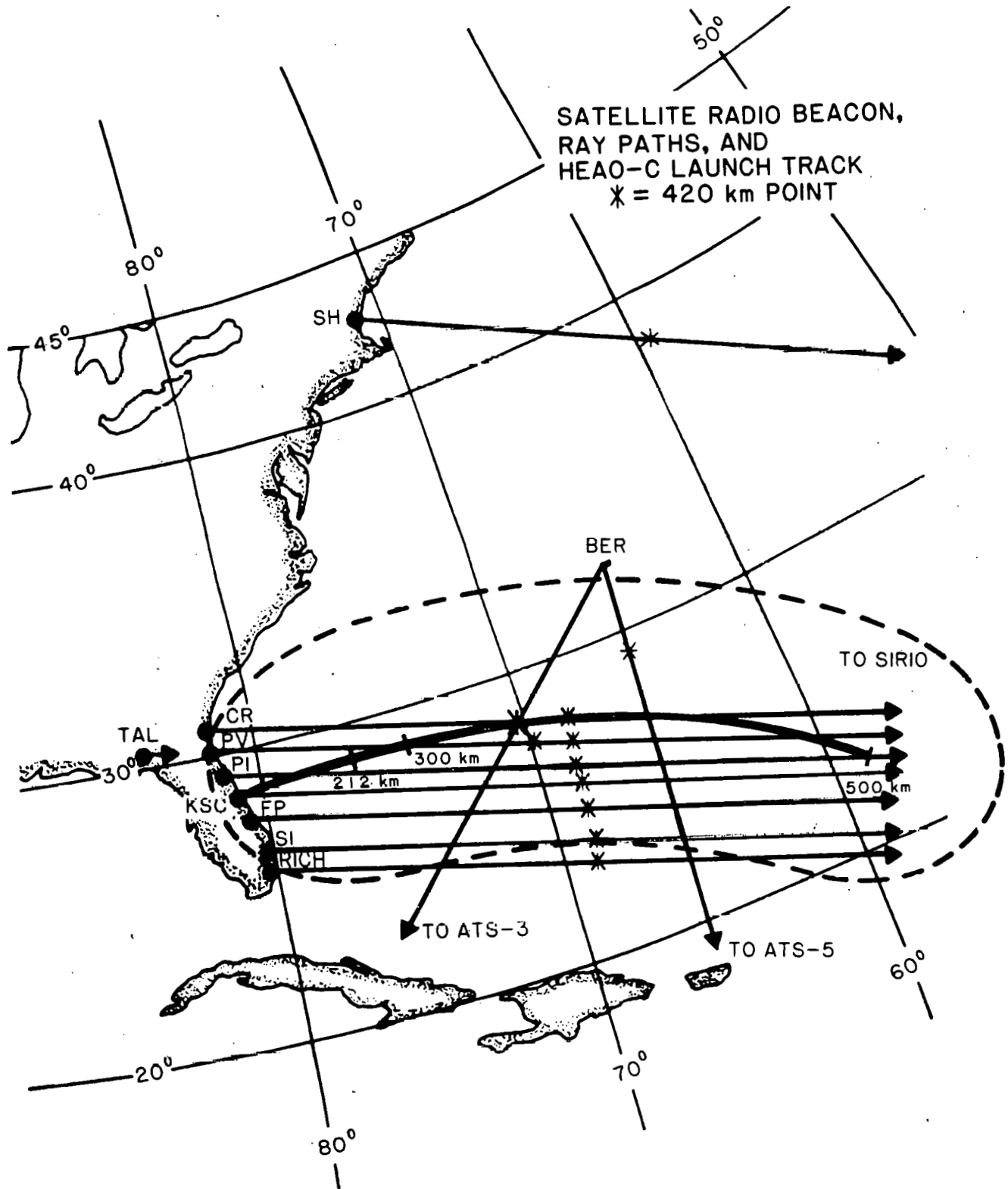


Fig. 22 Observation of the Ionospheric Hole Caused by the Launch of the HEAO-C Satellite

observations. Emissions from excited OH radicals were also sought in the HEAO-C experiment. Current theory is consistent with the failure to detect the 6660A emission line. However, the 30% enhancement of the background 2.9 mm OH band that was observed requires further theoretical analysis. Such an enhancement may have implications for remote sensing systems.

3.3.8.3 Radio Wave Propagation Effects

Plasma depletion in the bottomside ionosphere was examined using VLF transmissions at 10-13 kHz and HF transmissions from 3.6-21.0 MHz. The maximum usable frequency was in excess of 21 MHz. These transmissions would be reflected from the unperturbed bottomside ionosphere. Frequencies from 50-400 MHz that would penetrate the ionosphere were used to look for plasma structural irregularities (striations).

The VLF data⁹⁹ showed significant phase changes that occurred several hours after the launch and could not be associated with known natural disturbances, e.g., solar flares (see discussion in Sec. 3.3.7).

The HF diagnostics included a Coherent Backscatter Radar.^{105,106} Observers noted an alteration in the character of Earth backscatter that began 6 min after the launch and lasted for 5 min. A strong interfering signal was also detected at the same time, which may have been the result of anomalous refraction, but further analysis is required. HF communication experiments performed in the frequency range of 8-13 MHz did not show any severe long-term disruptions or even any effects unequivocally tied to the rocket launch, but signals recorded after the launch appeared to have somewhat lower amplitudes and the general tendency was for fading to be more intense than before.¹⁰⁵ Analysis of reports received from over 150 Amateur Radio Operators¹⁰⁷ that participated in HF communication experiments (3.6-50.0 MHz) indicated that no unusual effects or major band dropouts were seen at any of the frequency bands observed. While some signal fading at 14 and 21 MHz was noted after the rocket passed through the F₂ peak, such fading was not distinguishable from that caused by natural ionospheric fluctuations that were seen before and after the rocket launch. Most attention, based on the prelaunch prediction of the maximum usable frequency, was focused on the 14 MHz band. It turned out, however, that the maximum usable frequency was in excess of 21 MHz. The map in Fig. 23 shows the locations of the reporting Ham Radio Operators. From the transmitting station in Puerto Rico, the stations cover an azimuthal range labelled ϕ in the figure, from about 310-360°. The azimuthal range labelled θ covers all the stations, amongst others, that reported signal fading at 14 and 21 MHz. More is said about this below.

Observations of commercial radio links¹⁰⁸ (15-21 MHz) indicated some minor fading believed to be connected to the rocket launch. In particular, the Radio Nederland Signal at 15.3 MHz faded out for 30 s 8 min after the launch and the Deutsche Welle Signal at 17.8 MHz faded out for 5-6 min after the launch.

The search for plasma irregularities produced ambiguous results. One observer¹⁰⁵ noted what appeared to be signal scintillations in the ATS-5 communications satellite beacon operating at 136.47 MHz. Another observer¹⁰⁹ noted signal dropouts from a polar-orbiting Navy Navigation satellite beacon

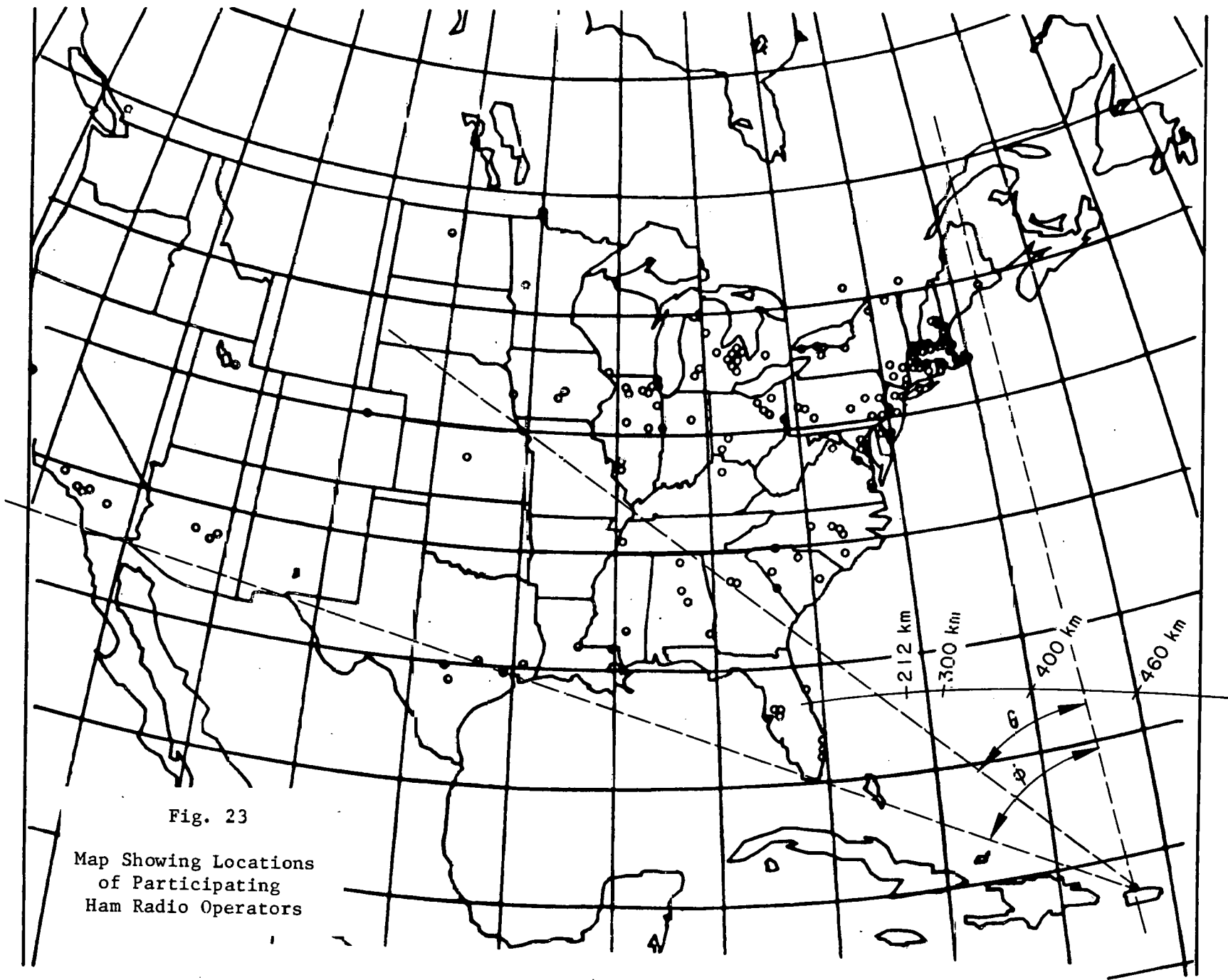


Fig. 23

Map Showing Locations
of Participating
Ham Radio Operators

at 140 MHz during substantial portions of the time when the ray path traversed the center region of the hole on two passes. On the other hand, observers^{110,111,112} monitoring the SIRIO satellite beacon¹¹³ (136.14 MHz) and the LES 8 satellite beacon (250 MHz)¹¹⁴ saw no evidence of scintillations. Results of optical measurements also showed no irregular structures or striations down to a size range ≥ 1 km (limit of resolving power).

3.3.8.4 Theoretical Predictions of Radio Wave Propagation Effects

It is worthwhile comparing the effects observed during the HEAO-C launch with theoretical expectations.

Through the use of model simulations of ionospheric depletions combined with ray tracing techniques, it is possible to estimate theoretically the effects of such depletions on electromagnetic waves in various frequency ranges.⁷⁷ The VLF waves propagate either via the earth-ionosphere waveguide or along field-aligned ducts in Earth's magnetosphere. Since the first propagation mode uses the lower boundary of the ionosphere, (D- and E-regions), any changes in that boundary are likely to influence such propagation. On the other hand, F-region modifications as observed in the HEAO-C launch can influence coupling between the earth-ionosphere waveguide and the ducts in the magnetosphere. Hence, VLF waves that might otherwise propagate via the duct mode may be prevented from either entering or leaving such a duct by F-region modifications.¹ It is not known at present if the VLF observations during the HEAO-C experiment are consistent with modifications to either of these propagation modes.* One final way in which VLF propagation might be influenced by F-region depletion is via the formation of artificial ducts caused by the depletion of plasma from an entire magnetic flux tube. According to theoretical estimates⁷⁷ such a depletion duct requires about 3 hr to form and could have been formed in the HEAO-C launch releases.

HF waves (3-30 MHz) can be refracted by plasma density gradients in F-region holes. Hence, such holes can produce focusing, defocusing, and/or fading, depending on the relative locations of the transmitter, receiver, and hole. In the case of the HEAO-C launch, estimates were made of the most probable locations for observing refraction effects of the F-region hole. Weak fading effects were indeed seen at 14 and 21 MHz, as expected, but their magnitude and duration were similar to those caused by natural effects unrelated to the rocket launch. In fact, the question that bothered most investigators with respect to the HF observations during the HEAO-C launch was: In view of the fact that a region of the ionosphere of the order of 600-1000 km in north-south extent and 2000 km or more in the downrange direction was affected, and the middle of that region suffered a 75% of total vertical-column, electron-content loss, why were no significant effects seen in the HF propagation for paths that went through that region?

This question has not been satisfactorily answered as yet although some suggestions have been made. First, as suggested by Klobuchar et al.¹⁰⁷ and illustrated in Bernhardt's paper,⁷⁷ HF propagation does not depend upon a

*Such a determination would require additional data analysis and model simulations beyond the scope of the present program.

point reflection from a small area in the ionosphere, but rather involves an ensemble of all possible ray paths that connect the transmitter and receiver. Such an ensemble of ray paths can cover a fairly large geographic region having varying degrees of refractivity. Hence, if the disturbed region is not sufficiently large, many ray paths could still connect receiver to transmitter or, possibly, new ray paths could be formed. This explanation seems plausible and suggests that for similar-sized-rocket-engine burns one would expect similar insignificant effects. It also suggests, however, that if the disturbed region were larger, it would not be possible to scale up the expected effects in any simple way. A threshold may exist beyond which single-reflection-type communication links might be severely disturbed, although no evidence for such a threshold now exists. In fact, the whole question of how ray paths can be influenced by holes of varying sizes requires a substantially more detailed analysis. Second, the geometry of the Ham Radio Operator network relative to the launch trajectory could have been more favorable. The entire launch trajectory was east to southeast across the ocean, making placement of diagnostic equipment difficult. As shown in Fig. 23 from the transmitter location in Puerto Rico, the Ham network was contained within the azimuthal range ϕ of 310-360°. Referring back to Fig. 22, it is seen that this azimuthal range covers the Centaur stage burn from ignition at 211 to about 400 km altitude. The Ham network monitored frequencies up to 21 MHz that were below the maximum usable frequency (≈ 22 MHz) and therefore would be reflected from the bottomside ionosphere (below 300-400 km altitude). Only those exhaust molecules injected into or settling downward from above the region of 200-400 km would affect the bottomside reflection paths.

Because the exhaust molecules are injected with a net absolute velocity (rocket-velocity-exhaust) in the forward direction, not all of the molecules injected below 400 and within ϕ end up within ϕ . Hence, only a fraction of the molecules would be expected to influence ray paths in ϕ . To put it differently, of the total region affected by the exhaust molecules (800 x 2500 km), only about 1/2 or less was being monitored by the Ham network. Furthermore, only a relatively small fraction of the Ham operators were tuned into the 21 MHz frequency band that was most likely to be affected. Those operators that reported fading at 14 and 21 MHz were located within the azimuthal range labelled θ in Fig. 23, as noted earlier. (Note that not all the stations pictured within θ were tuned into these frequencies and of those that were, not all reported signal fading, possibly connected with the rocket launch.)

VHF waves (30-300 MHz) are expected to be much less affected than HF waves. However, free-electron density changes cause changes in the phase and polarization of VHF waves, which is, in fact, the principle upon which changes in total electron content measurements are made. So long as the hole is fairly large, with smoothly varying densities, the refraction of VHF waves is expected to be very small. However, if the hole were unstable and tended to break up into small irregular regions as has been predicted⁷⁷ for depletions in magnetic equatorial F-regions, (equatorial spread-F), then the possibility arises that amplitude and phase scintillations could be generated in trans-ionospheric waves (satellite-borne communication links).

Although the HEAO-C F-region depletion occurred in mid-magnetic latitudes, evidence for plasma irregularities or striations was still sought. The results of this search, as mentioned, were ambiguous.

The major uncertainty in VHF effects is whether exhaust releases at the magnetic equator will trigger plasma irregularities as mentioned above. Such releases would result from HLLV launches to equatorial LEOs. Evidence for effects of equatorial launches may be obtainable from future Ariane rocket launches. Also, a special burn of the Space Shuttle OMS is planned¹¹⁵ for the future to search for such effects.

3.3.8.5 SPS Predictions for a Single HLLV and PLV Circularization Burn

Zinn et al.,⁸⁸ predicted that a noontime circularization burn would give rise to a 1000 x 2000 km region within which the electron concentration would be reduced to 1/3 its normal *daytime* value. The ionosphere would return to normal in about 5 hr. A nighttime hole would be deeper and remain until shortly after sunrise, as did the HEAO-C hole. Based upon experience with the HEAO-C launch, a hole of at least the size predicted by Zinn et al. and perhaps one somewhat larger in the downrange direction can be expected. In addition, one would expect airglow intensities and radio wave propagation effects similar to those seen in the HEAO-C case. The circularization maneuver would likely occur over the south Indian Ocean or Australia for a launch from Cape Kennedy. The radio wave effects would probably be detectable if looked for, but would not be expected to seriously degrade amateur or commercial shortwave radio operations. The impacts could conceivably be somewhat more important if the entirety of the hole were located over Australia or any other populated large land mass. Effects on VLF and VHF beacon propagation seem rather uncertain at this time and should be looked into further. If LEO is to be located near the magnetic equator, then observations should be made to verify or refute the prediction of creation of artificial Spread-F.

3.3.8.6 SPS Predictions for a Single POTV Burn Near LEO

A single POTV injection burn produces 10 times as many exhaust molecules (10^{31}) as the HLLV circularization burn. That is, about as many exhaust molecules as the Skylab burn of 1973. If other things remain unchanged, then the geographic area covered by the hole would be roughly 10 times that of the HLLV circularization burn hole or about $2 \times 10^7 \text{ km}^2$ (2.6 times the size of the continental U.S.). It is known, however, that the size of the hole does not scale linearly with the number of exhaust molecules.¹¹⁶ The Saturn V hole morphology is not completely known because of incompleteness of the data. Mendillo⁷⁶ reported that it covered an area of about $4 \times 10^6 \text{ km}^2$ and lasted for about 4 hr after a noon-time launch. Zinn et al.,⁸⁸ however, has suggested that if characteristic thermospheric winds prevailed at launch time (no actual wind data is available), the hole actually lasted until sunrise the next day. Zinn et al. reconcile their estimate with the 4-hr lifetime reported by Mendillo by pointing out that the hole was probably blown out of the observatory's line of sight with the ATS-3 satellite after 3 or 4 hr.

Hence, it is expected that a noontime POTV burn would produce a hole lasting for 4-16 hr. Since the hole will cover an area as big or bigger than the continental U.S.A., the radio wave propagation effects would be more wide-spread but not necessarily more severe or longer lasting than those observed during the HEAO-C launch. The severity and duration of the fading of HF signals during that launch were judged to be unimportant. However, as

already noted, the relative magnitude of the observed effects is not completely understood at present and therefore it is not possible to transfer those observations to other situations, especially when 10 times as many molecules are involved. The geographic location of the POTV burn has not been specified.

3.3.8.7 SPS Predictions for a Single HLLV Second-Stage Burn (56-124 km)

Unfortunately, no data is available to assist in the prediction of the F-region depletion caused by the HLLV second-stage burn, and it is necessary to rely entirely on the theoretical model calculations. Each such burn releases about 2×10^6 kg of exhaust, most of it in an altitude range of 110-124 km. Zinn et al.⁸⁸ have performed a calculation taking into account only those molecules released between 118 and 124 km (about 40% of the total second-stage emissions). Following a launch at noon from Cape Kennedy, some of the exhaust molecules slowly diffuse up to the F₂ layer where they react with O⁺ ions. During the daytime O⁺ ions are produced about as fast as they are destroyed and the net effect of an HLLV launch is only a 10% reduction in plasma density. However, during the following nighttime, the O⁺ and free-electron concentrations were gradually reduced to about 70% of their normal values in the model simulation. These 10-30% reductions in daytime and nighttime plasma densities are likely to continue for a few days until the remaining H₂O and H₂ are photolyzed by sunlight. This low-level, long-lived depletion from a single second-stage engine burn is not expected to have any significant impacts. In fact, it would probably be difficult to detect except possibly during magnetically quiet times. However, since second-stage HLLV burns occur once every 12-24 hr, these low-level depletions will be long-lasting and may spread over a large portion of the globe.

3.3.8.8 Multiple Launch Effects in the F-region

No detailed calculations have been possible of the effects of multiple launches on the F-region. Existing models are not adequate to handle the global-scale processes. The development of appropriate modeling techniques would require a minimum of one to three years effort and additional basic data. Hence, for the purposes of this assessment, only relatively simple, crude estimates can be made.

The total number of electrons and ions in the ionosphere is of the order of 10^{32} . Taking into account 2 HLLV launches/24-hr period and the fact that each exhaust molecule can potentially recombine 2 electron-ion pairs, the theoretical maximum number of pairs that could be removed in a 24-hr period, assuming that the exhaust molecules were ideally distributed (100% efficiency), is 4 times as great as the number naturally present. This is, of course, a gross overestimate. If the HLLV-exhaust-distribution system were 2.5% efficient (still an overestimate), the *total global* ionosphere would suffer a chronic 10% depletion (varying with time of day and location). For the sake of comparison, about 2.5% of the exhaust molecules from the Saturn V burn of 1973 participated in the recombination of ions and electrons. For the HEAO-C launch the efficiency was about 15%. However, in both cases the molecules were released in the sensitive F-region.

It is therefore at least plausible that a *belt-like* region of the ionosphere surrounding the globe at the launch latitude would have a roughly 10% or more chronically reduced free-electron density in addition to the twice daily depleted regions of 1000 x 2500 km size. To this would be added one POTV burn per month producing a depleted region of the order of 3000 x 8000 km. Such a belt could have a width of the order of 2,000-10,000 km, which would require an overall electron/ion recombination efficiency of between 0.2% and 1.0% for all of the HLLV exhaust molecules. The consequences of such a belt of partial depletion on HF communications have not been assessed as yet. However, since lesser developed countries are most likely to utilize HF skywave (F-region) in lieu of satellite VHF communication links, even minor reductions in the utility of this communication mode may be important. Again, it must be stressed that this discussion of potential global-scale, F-region effects from multiple to rocket launches is very preliminary.

3.4 EFFECTS OF ROCKET EFFLUENTS ABOVE 500 km ALTITUDE

3.4.1 Plasmasphere and Magnetosphere

As shown in Fig. 4, the plasmasphere and magnetosphere make up the outermost regions of the Earth's atmosphere. Figure 24 shows a very simplified cross-sectional view of these regions in a plane perpendicular to the Earth's orbital plane. The curved lines represent some of the geomagnetic field lines. The structure of Earth's magnetic field resembles that of a dipole field within the toroidally shaped region referred to as the plasmasphere but departs dramatically from such a simple structure beyond the plasmopause at roughly $4 R_E$ due to interactions with the flux of charged particles or plasma that constitutes the solar wind. Aside from distorting the field lines, this interaction plays an important but not completely understood role in various auroral and other processes that influence upper atmospheric dynamics and terrestrial radio communications. Recently, serious consideration has been given to the role that such interactions may play in tropospheric weather. In fact, the so-called "solar terrestrial weather effect" was the subject of a symposium/workshop entitled, *Solar-Terrestrial Influences on Weather and Climate*, that was held at Ohio State University of July 24-28, 1978.¹¹⁷

The plasmasphere is populated with charged particles (i.e., a plasma) whose density falls off rather abruptly at the plasmopause. The innermost region of the plasmasphere is bounded by the ionosphere (not shown in the figure because it is too small on the scale of the figure to be distinguished from Earth's surface.) An important feature of the plasmasphere is the region of maximum density of high-energy-charged particles that is called the inner radiation or Van Allen Belt. The outer Van Allen Belt is located roughly between the plasmopause and the region of geosynchronous orbit, i.e., between about 3 and $7 R_E$. Hence, the SPS satellites will corotate with Earth in the equatorial plane at the outer edge ($6.6 R_E$) of the outer Van Allen Belt (VAB). For simplicity, the VAB is not shown in the figure.

The two SPS-related sources of disturbance of the natural plasmasphere and magnetosphere are (1) the rocket effluents used to propel the cargo carrying vehicles (COTVs) and personnel carrying vehicles (POTVs)

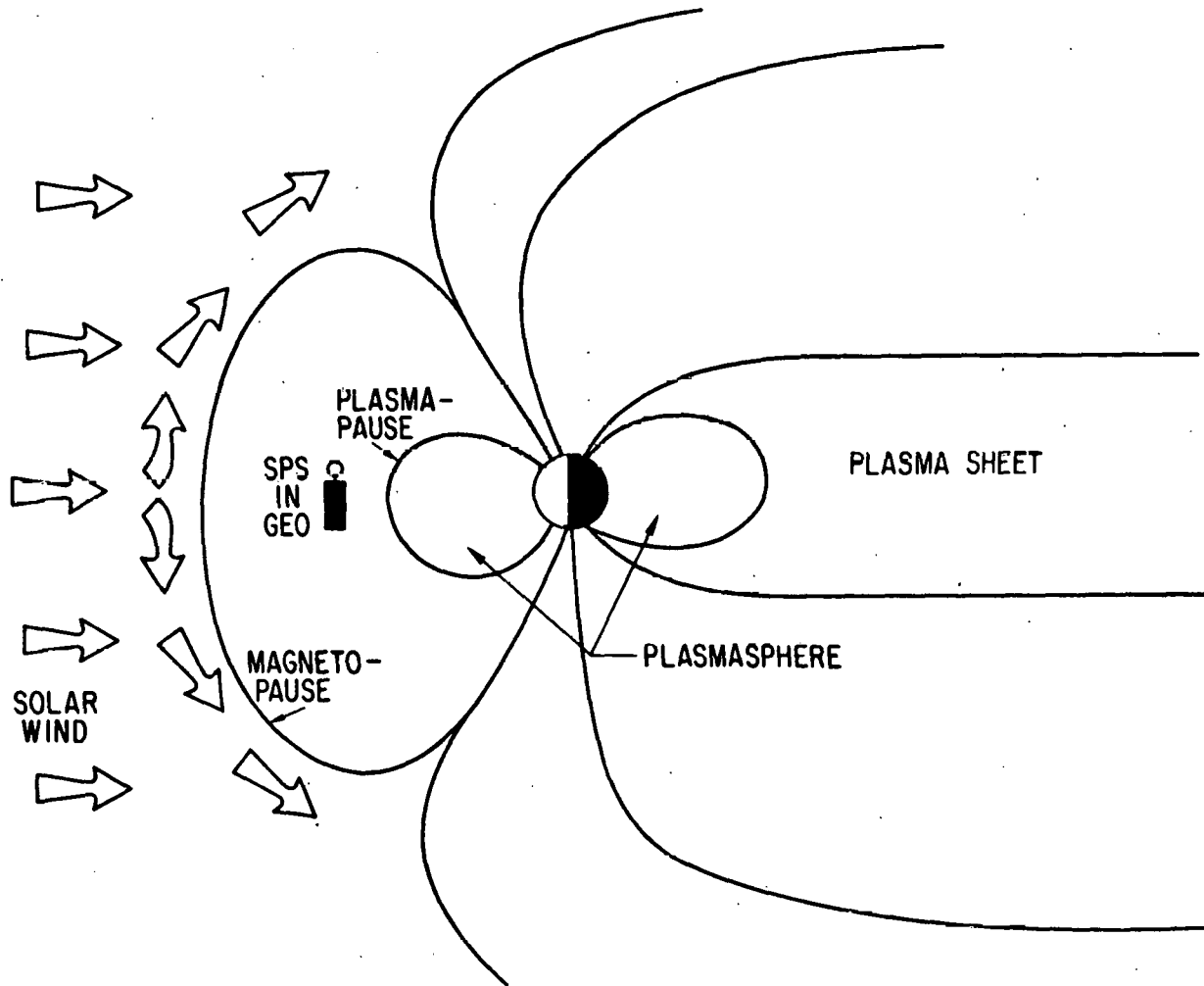


Fig. 24 Cross Section of Magnetosphere and Plasmasphere

between LEO and GEO and (2) the presence of the satellite structures themselves (including debris). The consensus reached at both the Westmont¹⁰ and La Jolla¹¹ Workshops held in August 1978 and June 1979, respectively, was that the presence of the satellite structures themselves would probably not lead to notable environmental effects. On the other hand, exhaust emissions from propulsion and stationkeeping activities of SPS spacecraft would be expected to induce substantial modifications of magnetospheric processes on both the local and the global scale. This expectation is due primarily to the relatively large mass and energy contents of these emissions when compared with the total mass and energy contents of the inner magnetosphere.

As a consequence of these opinions, the major emphasis in the Atmospheric Effects Assessment Program was devoted to the exhaust effluent effects.

3.4.2 COTV

Before discussing specific effects, it is important to set forth a reasonable emission scenario for the relevant Reference System² space transportation operations. Unfortunately, neither the POTV nor the COTV have been defined in great detail at this time. However, in view of the present state of knowledge regarding the potential impacts of propellant exhaust in the magnetosphere and plasmasphere, it is completely consistent to specify the characteristics of the POTV and COTV in an approximate manner. That is, at this stage of the assessment estimates to within a factor of 2 are regarded as acceptable.

According to the Reference System design,² two options for the COTV are being considered. Tables 16 and 17 give the relevant propulsion system characteristics and operational description of these two options. Table 16 also lists ion thruster parameters for two other options that have been discussed in the literature.

Details regarding the degree of divergence of the Ar^+ (argon ion) beam are not available. Smaller ion bombardment-type thrusters have been built and tested in high altitude test chambers. According to Reynolds,¹¹⁹ who has summarized the results of such tests, the degree of divergence of these beams varies considerably with their design. Since the ratio of the actual thrust obtained to the one ideally available (called the thrust factor) varies inversely with the beam divergence, it is obviously desirable to minimize the divergence. According to a January 1980 private communication from D.C. Byers of Lewis Research Center, beams having low divergence have a representative angular density profile $f_l(\theta)$ defined by

$$f_l(\theta) \approx \exp[-[36(1-\cos\theta)]^{1.83}],$$

where

= the polar angle measured from the beam axis,

while those having high divergence have a profile $f_h(\theta)$ defined by

$$f_h(\theta) \approx \exp[-[8.1(1-\cos\theta)]^{2.24}].$$

Using data presented in Reynolds,¹¹⁷ the low divergence beams $f_l(\theta)$ above, would have about 75% of the beam contained within about 10° of the beam axis, corresponding to a thrust factor of 0.98. High divergence beams have 75% of the beam enclosed within about $\leq 30^\circ$ of the beam axis, with a corresponding thrust efficiency of 90%. These details regarding beam parameters are presented here because they do influence the potential atmospheric impacts of the COTVs, even though not critically. The details also serve to connect the discussion of impacts given below with laboratory beam experiments and various discussions of beam impacts given elsewhere.

3.4.3 POTV

According to the Reference System design,² the POTV is a 2-stage, chemically-propelled (LO_2/LH_2) vehicle having a payload from LEO to GEO

Table 16 COTV Ion-Thruster Parameters

OPTION	Isp(s)	V(km/s)	E(keV)	D(cm)	N	n	T(N)
-	5000	49	0.49	-	-	-	-
Si	7000	69	0.97	120	4	269	-
GaAlAs	13000	127	3.34	100	1	259	13 ^a
-	16000	157	5.06	-	-	-	-

NOTE: Isp = specific impulse;
 V = argon ion speed = gIsp, where g is the acceleration due to gravity;
 E = beam particle kinetic energy = $\frac{1}{2}mV^2$, where m is the argon ion mass;
 D = thruster diameter,
 N = number of thruster arrays per COTV;
 n = number of thrusters per array;
 T = thrust per thruster in newtons.

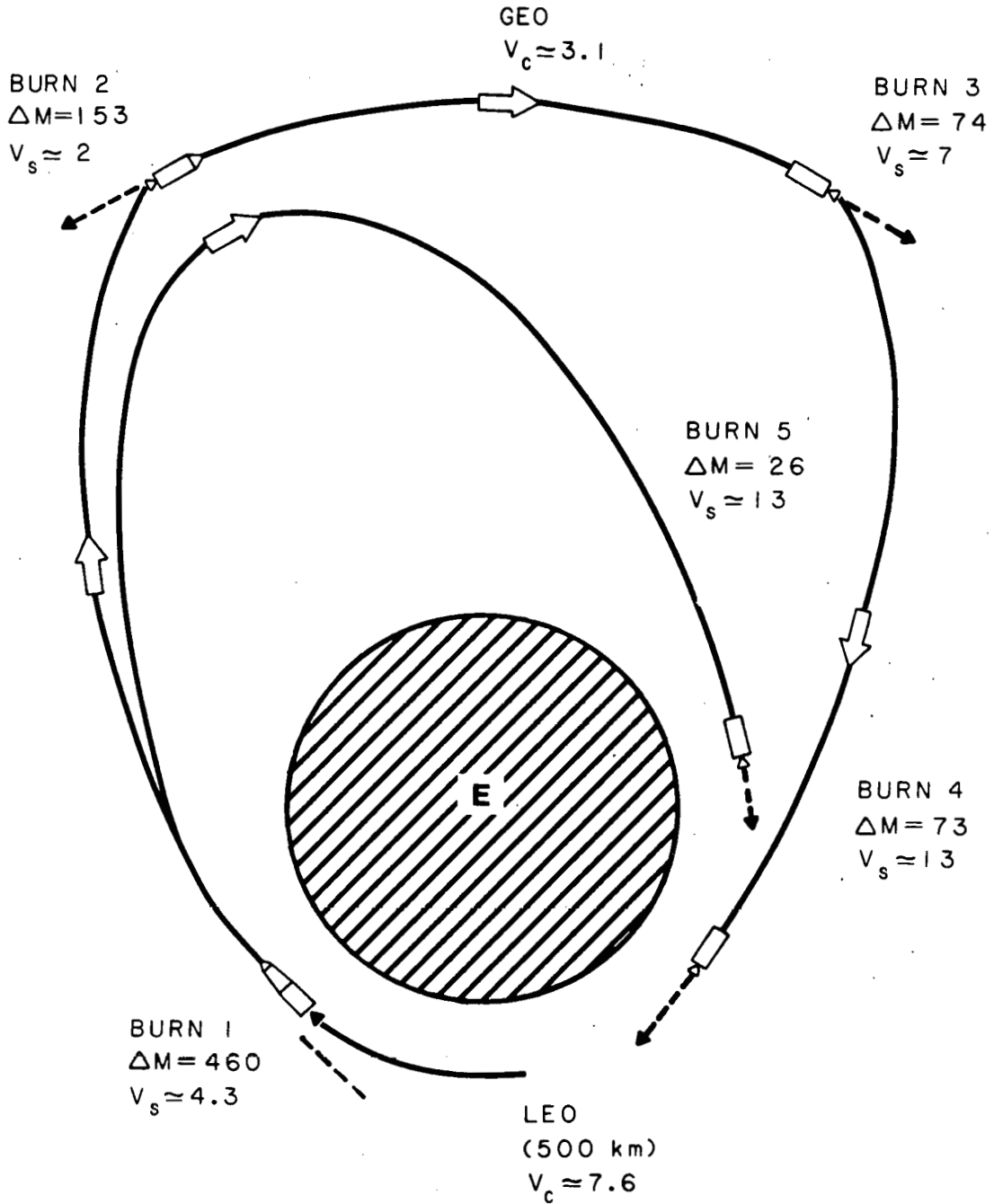
^aAssumes 90% thrust factor including effects of double ionization and beam divergence (Ref. 118).

Table 17 COTV Description

OPTION	$m_U(t)$	$m_D(t)$	N	M (t)	ΔT (days)	A(km ²)	PAYLOAD UP (t)
Si	835	150	30	30000	160	1.21	4000
GaAlAs	185	27	22	4700	180	3.0	3500

NOTE: m_U = propellant mass per flight from LEO to GEO,
 m_D = propellant mass per flight from GEO to LEO,
 N = no. of trips per year,
 M = total annual propellant mass,
 ΔT = round trip time,
 A = cross-sectional area of COTV. (From Ref. 2.)

of 151 t and a return payload of 55 t. Using these payloads and the fact that each stage has an inert mass of 30 t, an approximate flight scenario can be estimated. Such a scenario, as illustrated in Fig. 25 consists of 5 burns. Burns 1 and 5 are stage-1 burns. Burn 1 injects the POTV into an elliptical orbit having an apogee at GEO. After burn 1 is completed, stage 1 is separated and returns to LEO where burn 5 circularizes the orbit of the stage 1 vehicle at LEO. Burns 2, 3, and 4 are stage-2 burns that circularize the orbit at GEO, deorbit from GEO for the return flight to LEO, and circularize again at LEO, respectively. Assuming the above payloads and burn sequence, and a typical specific impulse for LO₂/LH₂ engines of 390 s, the required propellant masses obtained from a self-consistent but approximate calculation are as given in Table 18.



ΔM = MASS OF FUEL USED, TONNES

V_s = ABSOLUTE SPEED OF EFFLUENT, km/sec

V_c = ORBITAL SPEED, km/sec

ESCAPE SPEED: LEO = 10.76 km/sec

GEO = 4.34 km/sec

Fig. 25 Hypothetical Burn Sequence for a 2-Stage POTV Flight

Table 18 Hypothetical Burn Sequence for a
Typical POTV Flight

BURN NO.	VELOCITY CHANGE (km/sec) ^a	PROPELLANT MASS (t)	FATE OF EXHAUST GAS
1	2.37	460	Fall into atmosphere
2	1.466	153	Trapped in orbit
3	1.466	74	Escape to outer space
4	2.37	73	Escape to outer space
5	2.37	26	Escape to outer space
TOTAL			
PROPELLANT			
MASS (t)		786	

^aVelocity changes are from Park.¹²⁰ (His propellant masses correspond to different payload and vehicle assumptions.)

The total mass of the rocket at the start of burn 1 would be the sum of the total propellant required, the payload up (151 t) and the inert mass of both stages (60 t). Allowing for the fact that some additional propellant would be required for on-orbit maneuvering, attitude and reaction control, and reserve, the total propellant mass would have to be somewhat in excess of 800 t. The corresponding total mass of the rocket would have to be somewhat in excess of 1000 t. The Reference System document² (page 53) quotes a value of 830 t for propellant and a total mass of 1041 t for the rocket, which is quite consistent with the above calculation.*

3.4.4 Effects of POTV Emissions

Table 18 lists the fate of the POTV exhaust products for each of the 5 burns. These fates are determined by comparing the absolute velocity of the exhaust molecules with the gravitational escape velocity at the altitude of the burns. Only for burns 1 and 2 will the exhaust molecules remain within the atmosphere. The exhaust molecules from burns 3, 4, and 5 will exceed the gravitational escape velocity and will therefore, on the average, escape into outer space.

Estimates of the mass of H₂O and H₂ injected at LEO and GEO by POTV burns 1 and 2, respectively, are given in Table 19. On an annual basis, of the order of 5×10^{32} and 2×10^{32} H atoms would be injected at LEO and GEO, respectively, and about one-half as many oxygen atoms. The normal H-atom content between 500 km and the plasmopause ($4 R_E$) is about 3×10^{32} atoms. Atomic hydrogen is the dominant component above 800 km.

*The propellant requirement of 460 t indicated in Fig. 31 on page 55 of the Reference System document² is obviously inconsistent with the above numbers and has therefore been ignored.

Table 19 Emissions per POTV Flight^a

	Mass of Exhaust Molecules (t)		No. of Exhaust Molecules (10^{30})	
	H ₂ O	H ₂	H ₂ O	H ₂
Burn 1	443	17	14.8	5.1
Burn 2	147	6	4.9	1.8

^aAssumes an oxidizer to fuel ratio of 6.0.

Burn 1 will result in the deposition of a large amount of exhaust molecules above the F-region of the ionosphere. After injection, the exhaust molecules will have a net component of velocity in the forward direction, and, hence, will follow ballistic trajectories as they gradually descend into the F-region of the ionosphere. As discussed earlier, these molecules will cause a rather substantial ionospheric hole to form. (See Sec. 3.3.8 for further details.) The exhaust molecules injected at GEO during burn 2 will have sufficient velocity in the forward direction to remain gravitationally trapped in elliptical orbits centered around 15,000 km above Earth's surface. Since the velocity of the molecules is distributed over a range of magnitudes and directions, one would expect the orbiting cloud to gradually grow in size and cover a large portion of the magnetosphere from about 3,000-40,000 km altitude.⁹³ Although the density of the artificial cloud would be comparable to that of the neutrals naturally present in that region, it would not be sufficient to alter the relatively collision-free character of that region. Hence, the orbiting cloud would be expected to live for a fairly long time (of the order of weeks to months). As the POTV launch rate is once or twice per month, the possibility of an accumulating artificial cloud exists. This possibility has not been assessed at this time because of large uncertainties regarding loss processes and ambient densities.

The neutral exhaust cloud from a single POTV burn 2 would overlap a region of space normally occupied by the Van Allen Radiation Belts and the ring current. The ring current is formed during magnetic storms and consists of magnetically trapped particles in the energy range of 1-100 keV. This feature of the magnetosphere is associated with a part of the time-varying component of the geomagnetic field observed during magnetic storms and also is supposed to play a role in the precipitation of radiation belt particles into the atmosphere.

The Van Allen Belt particles have energies in excess of 100 keV and consequently pose a significant radiation hazard to equipment and personnel operating in or traversing that region of space. Magnetically trapped particles characteristically follow spiral orbits along magnetic field lines and reverse direction or "mirror" at sufficiently high altitudes to remain in the collision-free environment. A sudden change in the magnetic field or in the particle trajectory may scatter the particle into the so-called "loss cone." Trajectories with pitch angles within the loss cone have mirroring altitudes so low that the particles collide with neutral constituents of the atmosphere and escape magnetic confinement. High-energy particles naturally precipitated into the auroral zones (not necessarily from stable trapped

orbits) are a major source of thermal energy, ionizing radiation, and auroral displays at high latitudes. A number of mechanisms have been proposed to explain the natural precipitation of high-energy particles either from the inner magnetosphere that is normally occupied by magnetically stably trapped particles or from the outer magnetosphere (including the tail region). One mechanism, relevant to the SPS case, for removing relativistic electrons from magnetic confinement via precipitation into the atmosphere involves the interaction of ring-current ions with Van Allen Belt electrons. This mechanism is discussed later.

In addition to precipitation, magnetically trapped, high-energy, ions can also escape confinement by charge exchanging with ambient, low-energy neutral particles. The lifetime for charge exchange is quite long, varying from 1 hr near LEO to weeks or months at GEO.⁹³ The result of a typical charge-exchange process would be the production of a fast neutral that, depending upon its direction of travel, would either enter the atmosphere or escape into outer space, and a slow charged particle that would be magnetically trapped.

The main effect of the neutral POTV exhaust cloud is that it would serve as an artificial source for charge-exchange interactions with ring-current ions.⁹³ The natural lifetime of ring-current ions has been estimated to be of the order of 10^5 s (about a day), based on the duration of the recovery phase of magnetic storms. Chiu et al.⁹³ have estimated that the enhanced charge-exchange caused by the neutral POTV exhaust cloud could reduce the lifetime to between 10^3 and 2×10^4 s, i.e., an 80-99% reduction. This reduction might be regarded as a benefit if it corresponded to a reduction in the recovery phase of a magnetic storm, since there would be a shorter time during which natural electromagnetic interference effects would be experienced.

The Van Allen Belt radiation ions have much higher energies (average energy ≈ 100 keV) than the ring-current ions and are relatively immune to charge-exchange loss induced by the neutral POTV exhaust cloud, according to Chiu et al.⁹³ However, these investigators note that a plasma effect called the "critical-velocity phenomenon" may play an important role in determining the effects of the neutral cloud. Characteristically, this phenomenon results in the conversion of a fraction of neutrals into thermal ions when the neutrals are moving with velocity $V_c = (|e|V_i/m)^{1/2}$ relative to an ambient plasma, where $|e|V_i$ is the ionization potential energy of the neutral particles of mass m . Although this phenomenon has not been extensively verified in the laboratory or in space, Chiu et al.⁹³ suggest that it may result in the conversion of at least part of the POTV neutral exhaust molecules into low-energy atomic oxygen ions. The effects of these ions would be similar to those of the Ar^+ ions emitted by the COTV in the plasmasphere that are discussed below. In particular, the presence of cold heavy ions suppresses one of the mechanisms supposedly responsible for the removal of relativistic electrons from the Van Allen Belts, leading therefore to the enhancement of the relativistic electron dosage.

3.4.5 Effects of COTV Emissions

As already indicated, the number of Ar^+ ions injected into the plasmasphere between 500 km and the plasmopause ($4 R_E$) is roughly comparable to the

number of H atoms naturally present and exceeds the natural Ar^+ ion content by several orders of magnitude. Hence, not only is the mass of effluent greater than that naturally present, but the kinetic energy of the Ar^+ ion (≈ 3.5 keV) is about three orders of magnitude greater than the thermal energy normally present in that region and is comparable on an annual average basis to the energy injected by magnetic storms during relatively quiet periods. Hence, the magnitude of the perturbations to the natural state of the near-Earth space environment (500-36,000 km) is expected to be large.

The overall spatial and temporal distribution of the COTV effluents is governed both by the relatively low thrust of the ion engine array and by the inclination of the LEO plane. If the HLLVs are launched from the U.S., the LEO plane will be inclined relative to the equatorial plane (GEO is in the latter plane). Therefore, it will be necessary to make a plane transition during the course of the trip from LEO to GEO, meaning that during at least part of the trip, the Ar^+ ion beam will not be pointed orthogonal to the geomagnetic field lines. In fact, it has been suggested¹²¹ that the most fuel-efficient transfer orbit would involve oblique angles between the COTV velocity vector and the ion engine thrust vector, i.e., oblique angles between the geomagnetic field \vec{B} and mean beam velocity \vec{V} . These angles would be varied as a function of altitude. However, the consequences of varying the angle between \vec{B} and \vec{V} with regard to magnetospheric impacts have not been examined in detail and have been examined in only a preliminary manner with respect to the fate of the Ar^+ by Holze and Liemohn.¹²¹ Consequently, it will be assumed that, for the most part, $\vec{V} \perp \vec{B}$. The low thrust characteristic of ion propulsion systems requires that the transfer from LEO to GEO occur over a gradually unraveling spiral trajectory about the Earth requiring about 130 days. During that time, the orbit period with respect to the rotating Earth will increase from about 1 1/2 hours to infinity (geosynchronous). The beam from the ion thruster array consists of a neutral plasma of Ar^+ ions and electrons. These charged particles do not recombine within the beam. For purposes of this discussion, we are assuming that the Ar^+ ions are ejected with a kinetic energy of 3.5 keV and that the beam is of the low-divergence type, i.e., it is confined to within about 10° of the beam axis.

In order for the beam to propagate across the geomagnetic field \vec{B} , the $\frac{1}{c} \vec{V}_b \times \vec{B}$ force, when \vec{V}_b is the beam propagation velocity, and c is the velocity of light, must be balanced by an electric field \vec{E} called the polarization electric field. This field is established within the beam by the separation of the Ar^+ ions and the electrons. So long as the equality $\vec{E} = \frac{1}{c} \vec{V}_b \times \vec{B}$ can be satisfied, the beam will propagate as an entity across the geomagnetic field lines. However, as the beam expands and becomes more dilute, the polarization electric field becomes weaker and may even be partially shorted out by the ambient plasma if current paths can be established, either across the field or through a conductor at the ends of the field lines. Hence, the beam would slow down in accordance with the above expression and eventually come to rest (unless it escaped the magnetic field altogether).

The total energy content of the beam is partitioned between a thermal component associated with random particle motion, a net kinetic energy of beam propagation ($\frac{1}{2} M V_b^2$, where M = total beam mass propagating at velocity V_b), and the kinetic energy associated with the ordered, individual particle motion relative to \vec{V}_b . Depending upon the specific mechanism involved, stopping of the beam as a whole would not necessarily dissipate a large

fraction of the total beam energy. The precise nature of the interaction between the dense, energetic Ar^+ plasma beam from the COTV and the ambient plasma embedded in the geomagnetic field is not presently known for certain and no actual experiments have been performed that come even close to the parameter values envisioned here. (The archived results of beam experiments and ion-release experiments have been reviewed by Chiu.)¹²² Consequently, some basic questions such as the following must be addressed:

1. What are the dominant interactions between the beam, the ambient plasma, and \vec{B} ?
2. Will those interactions cause the beam to propagate as an entity or will the beam rapidly break down into particles interacting individually with the environment?
3. Will the beam be brought to rest within the geomagnetic field or will the bulk of it escape into outer space?
4. Will any Ar^+ ions be stripped away from the beam and become individually confined within the geomagnetic field?
5. What will become of the ion kinetic energy and beam streaming energy?

The answers to the above questions, to a large extent, determine the impacts that the COTV effluents will have on both the near-Earth Space environment and the terrestrial environment. Unfortunately, it is currently not possible to provide definitive answers to any of the above basic questions. Rather, it is necessary, for the purposes of this assessment, to try to obtain some idea of the range of possible effects based upon extrapolation of archived data and current theories and models. It is not surprising, therefore, that no complete consensus exists regarding the potential impacts of the COTV effluents. Several models* have been proposed to describe the evolution of the Ar^+ ions after injection from the COTV ion thruster array. Of those models, that of Chiu et al.⁹³ has been developed and studied in the greatest detail, and while it relies on experimental data and theories of uncertain applicability to the SPS case, it appears at least plausible within the scientific community. (That is not to say that other models are implausible). It also admits the greatest range and magnitude of environmental impacts, and, therefore, if it turns out to be in error, it will be to the benefit of the environment. Conclusions based on the Chiu et al. work will be compared in the following discussion, where appropriate, with those of other investigators.

According to the Chiu et al. model,⁹³ the beam creates a wake in the geomagnetic field line structure as it emerges from the thruster array, much as a boat creates a wake in water as pictured in Fig. 26. This distortion of the field lines is described as the propagation of a wave along the field lines through the ambient plasma and is called an Alfvén wave. The Alfvén wave propagates the polarization electric field poleward along the \vec{B} field

*The term "model" is used here not to denote a formal mathematical computerized model but rather to denote a more descriptive construction, including statements of the applicable physics and governing mathematical equations.

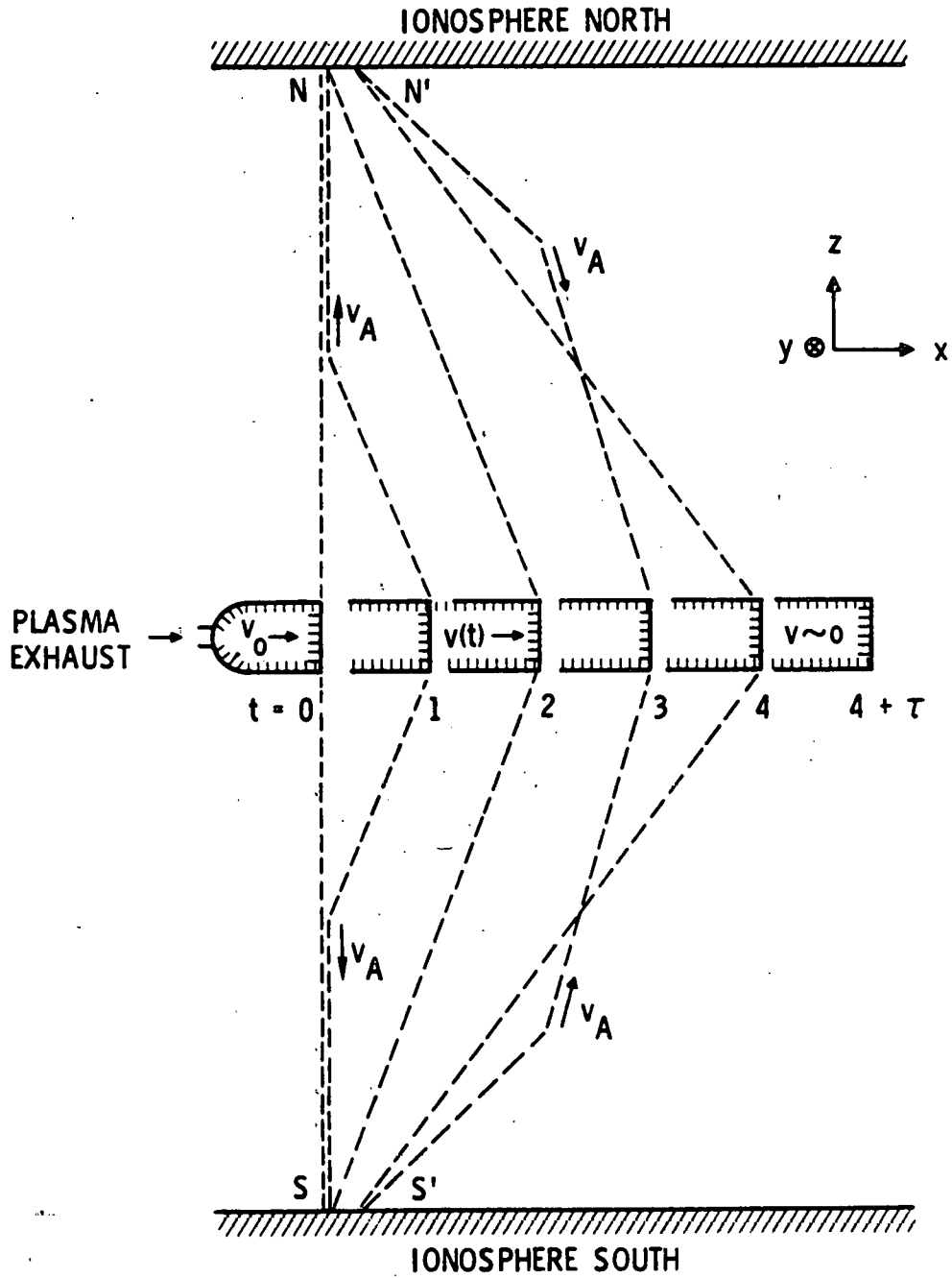


Fig. 26 Ar^+ Ion-Beam Wake in Magnetosphere

lines until the lines enter the ionosphere ("foot" of the field line) where the ambient ion density is high enough so that the conductivity can cause a cross-field current to flow and short out the polarization field. The shorting out process dissipates energy via Joule heating near the foot of the field lines and consequently drains energy from the propagating ion beam. Hence, according to the Chiu et al. model, the beam slows down and comes to rest relative to the geomagnetic field within 1000-2000 km. At that point, most of the 3.5 keV kinetic energy of the Ar^+ ions has been dissipated in the poleward regions of the ionosphere so that the ions constitute a relatively low energy or cold plasma. Of course, some of the ions will have energies ranging from a few eV up to the original 3.5 keV.

Holze and Liemohn,¹²¹ on the other hand, do not invoke the Alfvén wave mechanism and assume rather that after a period of initial expansion of the beam within the ambient plasma the polarization field is reduced to the point where the beam can no longer propagate ($V_p=0$). However, the ions will retain essentially all of their original kinetic energy, since there is no mechanism such as Joule heating in their model to dissipate this energy. They assume that since the initial LEO plane is inclined to the equatorial plane, a substantial component of the ion velocities will be parallel to the \vec{B} field lines so that the ions will disperse from the beam and move in spiral orbits along the field lines and become magnetically trapped as a relatively high energy or hot plasma (≈ 3.5 keV). Both the Chiu et al. model and the Holze-Liemohn model conclude that the bulk of the Ar^+ ions become trapped in the geomagnetic field. In the former model, most of the Ar^+ ions enter with very low energy, the balance being dissipated as heat in the lower and mid-latitude ionosphere. In the latter model, the ions become trapped with most of their energy intact.

A third picture, due to Curtis and Grebowsky,¹²³ claims that the Alfvén wave mechanism is not applicable and will not be effective in stopping the beam as an entity, or in dissipating its energy in the ionosphere because the Alfvén wave propagation speed is too slow relative to the initial beam propagation speed. Chiu et al.⁹³ have estimated the Alfvén wave propagation speed to be about one order of magnitude greater than that of the beam and therefore adequate to permit the Alfvén wave mechanism to effectively dissipate the beam energy. The Curtis and Grebowsky model assumed a slightly greater initial ion velocity than the Chiu et al. model (155 km/s vs 130 km/s, respectively), but this difference is probably not significant relative to their argument. The Curtis and Grebowsky model also assumed a much more divergent beam (22° relative to the beam axis) than the Chiu et al. model (10° relative to the beam axis). The importance of this opening angle difference is not clear.

In contrast to Holze and Liemohn who say that after an initial period of expansion the beam particles become individually trapped in the geomagnetic field, Curtis and Grebowsky claim that the polarization \vec{E} field is never reduced to the point where the beam propagation ceases and the individual ions disperse. They claim that the bulk of the beam propagates out of the geomagnetic field to outer space. Only the ions making up the outer sheath of the beam where the \vec{E} field is very weak are separated from the main beam and become magnetically trapped at essentially 3.5 keV.

In summary then, the three models suggest that:

Chiu et al.:⁹³ The beam is stopped relative to the magnetosphere within about 1000 to 2000 km. Most of the kinetic energy is dissipated in the ionosphere poleward from the equator. Most of the Ar⁺ ions are magnetically trapped with very low energies.

Holze & Liemohn:¹²¹ Essentially all of the Ar⁺ ions are magnetically trapped with their original kinetic energies.

Curtis & Grebowsky:¹²³ About 90% of the beam escapes to outer space intact. The remaining fraction of Ar⁺ ions are magnetically trapped with their original kinetic energy.

At present, there is no experimental data that can be invoked to decide which description is the closest to reality. Hence, until more appropriate experiments are performed and more theoretical work is undertaken, considerable caution is advised in interpreting the results of these models.

The four main potential impacts of Ar⁺ ions beam are:

1. Airglow emission enhancement due to interactions of the beam with ambient constituents, especially near LEO where the atmospheric density is greatest.
2. Alteration of auroral processes and response to changes in the solar wind.
3. Modification of the Van Allen Radiation Belts.
4. Generation of plasma instabilities that can cause interference with radio wave communication links involving satellites.

3.4.5.1 Artificial Airglow Emissions

Artificial Airglow is important because of possible interference with optical remote sensing systems. Space-based optical sensors are currently used for military, scientific, and earth-resource applications. For data interpretation purposes, such systems must take into consideration the background optical emissions from atmospheric constituents. The latter emissions, ranging from the far infrared (FIR) to the extreme ultraviolet (EUV), are collectively called airglow. COTV effluents could stimulate artificial airglow via two mechanisms:⁹³

1. Dissipation of the beam streaming energy in the form of thermal energy heats the ionosphere to the level of a few eV, which is comparable to the energy of natural photoelectrons that produce natural airglow. Its intensity, however, would be much greater than that of natural airglow because the artificially heated thermal electron flux would be orders of magnitude higher than the photoelectron flux.⁹³ It is not possible to assess the impact of the artificial airglow produced via this mechanism both because the exact level of thermal modification is not known and because the portion of the airglow spectrum is not extensively used, by present space-based sensing systems.⁹³

2. In the vicinity of LEO the energetic ion beam can have a direct impact on the relatively dense atmosphere. This impact will stimulate the near-UV part of the airglow spectrum (more properly auroral spectrum). This stimulation is more serious since this portion of the spectrum is utilized by present-day sensors. Chiu et al.⁹³ have estimated the intensity of this "artificial aurora" and given upper limits of the order of 10^6 to 10^7 Rayleighs for the altitude range of 350 km down to 150 km and for the wavelength range of 2500-5500Å. For comparison, the natural auroral intensities are of the order 10^5 Rayleighs at 150 km. If artificial intensities are indeed an order of magnitude greater than values arising in natural aurorae of moderate strength, then it may be necessary to employ mitigating strategies. A possible strategy would be to direct the beam essentially orthogonal to the \mathbf{B} field until the COTV is ≈ 2000 km from LEO; that is, avoid plane changes until after an altitude of 2000 km above LEO has been achieved.

3.4.5.2 Artificial Auroral Substorms

As discussed above, the Alfvén waves generated by the movement of the ion thruster beam through the ambient plasma propagates the polarization electric field down a magnetic field line to the denser regions of the ionosphere. Here, due to the relatively high conductivity, a cross-field electric current can flow like natural auroral currents created during magnetic storms. Two known consequences of the natural ionospheric currents are, first, the induction of current surges in terrestrial conductors such as pipelines and telephone and electric power transmission lines. Electrical currents in pipelines enhance corrosion,⁹³ while surges in telephone and transmission lines cause tripping of circuit breakers.¹²⁴⁻¹²⁶ The second effect is the dissipation of energy due to Joule heating (I^2R loss). The natural energy dissipation process is an important source of thermal energy in the auroral zones that together with the direct solar radiation energy input near the equator is believed to give rise to the thermospheric wind patterns.¹²⁷ Diurnal and seasonal variations in these two energy inputs cause corresponding temporal variations in the thermospheric wind patterns. The extent to which these wind patterns influence tropospheric weather processes is not known at present. However, modulation of auroral energy input processes, including the precipitation of high energy particles into the upper atmosphere by variations in the solar wind, has been suggested as a possible mechanism for the so-called "solar-terrestrial weather effect."

The main difference between the natural and "artificial auroral" processes is that the natural ones occur sporadically and over a large area in the high latitude regions. The natural energy dissipation rate is about 10^{17} ergs/s. The "artificial auroral" processes occur continuously and at mid-latitude regions. The corresponding energy dissipation rate is about 5×10^{15} ergs/s. Hence, averaged over a several-month period, the total energy dissipated is comparable.⁹³ The fact that the SPS-related processes occur in mid-latitude regions means that if electrical currents are induced in terrestrial conductors, they will occur in more populated areas than is the naturally occurring case and, therefore, could be a greater nuisance.

Based on a comparison with the natural heating processes of particle precipitating and Joule heating (roughly equal in magnitude), the artificial Joule-heating process could conceivably raise the temperature of the ambient

plasma to a few eV (10^3 - 10^4 K).⁸ Chiu et al.⁹³ used a simple, four-constituent, steady-state model to estimate the effect of heating the ambient plasma (including Ar^+ ions injected by the COTV) to 5 eV. The other constituents were H^+ , O^+ , and e^- . The results, shown in Fig. 27, indicate that the presence of the Ar^+ ions by themselves will not upset the normal distribution of plasmaspheric H^+ and O^+ , but the addition of thermal energy via Joule heating to 5 eV substantially increases the density of O^+ and Ar^+ ions at high altitudes.

3.4.5.3 Modification of the Van Allen Radiation Belts and Ring-Current Particles

The appearance of O^+ and Ar^+ ions (or any other heavy, relatively cold plasma constituents) in the region normally occupied by relativistic electrons can have serious consequences for the radiation dosage. The relativistic-electron distribution in the equatorial plane peaks at $1.8 R_E$ (the inner Van Allen Belt) and at $5 R_E$ (the outer Van Allen Belt). The minimum in between is located near $3 R_E$. This distribution is strongly correlated with magnetic storm activity.¹²⁸

Dynamical interactions between the magnetosphere and the solar wind are ultimately responsible for populating these high-energy-particle regions. Their populations are diminished by interactions with plasma turbulence that causes the magnetically trapped particles to be scattered into their loss-cones so that they mirror at altitudes where collisions with the denser neutral atmosphere are more probable than normal. Such collisions cause the

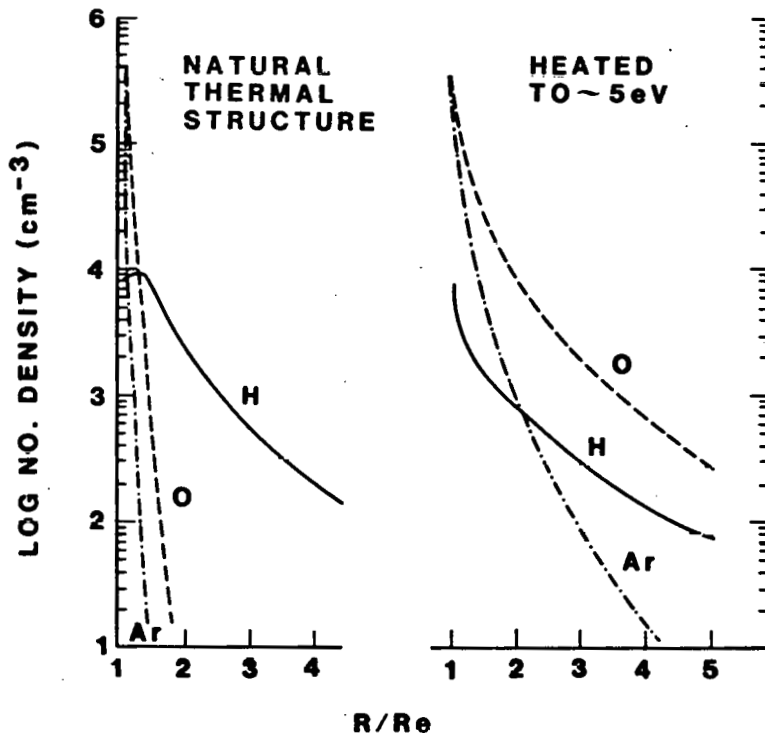


Fig. 27 Principal Components of Plasmasphere

trapped particles to deposit their energy into the ionosphere and thermosphere where they escape magnetic confinement. One of the mechanisms for loss of relativistic electrons from the radiation belts is a resonance between waves generated by gyrating ring-current protons and relativistic-electron gyrations. The removal of relativistic electrons by operation of this so-called ion-cyclotron instability occurs when the ring currents and radiation belts occupy the same region of space. This double occupancy occurs during the recovery phase of a magnetic storm.

As pointed out by Chiu et al.,⁹³ the presence of cold, heavy plasma components such as O^+ and Ar^+ are likely to damp out the ion-cyclotron instability and, therefore, prevent the natural removal or precipitation of relativistic electrons into the atmosphere, increasing their mean dosage with time. Based on the assumption that this instability is the principal mechanism for relativistic-electron precipitation, they estimate a twofold to threefold increase in dosage in the outer-radiation belt. There is, according to Chiu et al., no danger of a corresponding increase in ring-current-proton dosage, since the main mechanism for its limitation is not expected to be affected by the presence of cold, heavy ions. Also, the high atomic-numbered ions with high energy (HZE particle population) are not expected to be affected by the addition of the heavy, cold ions (O^+ and Ar^+).⁹³ On the other hand, the thermal electrons associated with the O^+ and Ar^+ ions probably (not proven) would cause the precipitation of the lower energy electrons (<100 keV) that populate the ring current. This precipitation is in contrast to the stabilizing influence that the O^+ and Ar^+ ions would have on the relativistic-electron population ($E > 1$ MeV) of the outer radiation belt. The rapid loss of ring-current electrons also contributes to the reduction of geomagnetic activity.

If a portion of the Ar^+ ions would escape from the beam with most of its original kinetic energy (3.5 keV), it would constitute a relatively hot plasma addition. Such an addition could have completely different consequences as suggested by Chiu et al.,⁹³ and elaborated upon by Curtis and Grebowsky¹²³ and by Holze and Liemohn.¹²¹ The energetic Ar^+ ions would add a ring-currentlike population that could interact resonantly with the radiation-belt electrons to remove them from their trapped orbits. This activity would effectively cause artificial aurorae and reduce relativistic-electron dosage. Hence, the state of the Ar^+ ions that become trapped in the geomagnetic field is important for determining their impact.

Additionally, as pointed out by Curtis and Grebowsky,¹²² if the bulk of the beam escapes the geomagnetic field altogether so that only about 10% of the ions are deposited in the magnetic field and if that portion becomes trapped at nearly its original kinetic energy; then virtually no energy would be dissipated in artificial mid-latitude auroral processes, no artificial mid-ionospheric currents would be generated, and no currents would be induced in terrestrial conductors. Hence, the ionosphere and plasmasphere would not be heated to a few eV causing the drastic compositional changes predicted by Chiu et al.⁹³

Finally, before leaving the subject of radiation belt modification, it should be mentioned that a new analysis relevant to this subject has recently been reported by Cladis, et al.¹²⁸ They investigated the changes in relativistic-electron populations in the outer trapping regions (near GEO)

during magnetic storms, based on data obtained from the OGO-5 satellite in 1968, but only recently reported in the literature.^{129,130} Though their analysis is too new to have been properly reviewed in the scientific literature or independently confirmed, it is nevertheless sufficiently important to be noted here.

Cladis, found that during typical magnetic storms, there is very little loss of relativistic flux in magnetic shells with $L < 4$ or 5.* In fact, there appears to be a substantial enhancement in that flux in the region from $L = 3$ to 4, suggesting the inward migration of flux during the early phases of a storm. However, outside of $L = 5$, the apparent loss of flux was very large. Although part of this loss could be explained in terms of inward movement, the remainder of the loss could not be explained in this manner and was thought to represent a true loss. As a storm subsides, they found that the flux increases throughout the trapping region from $L \lesssim 3$ to $L \lesssim 7$. This increase is caused by both the return of the old flux to its original position plus the injection of new electrons. They emphasized that because the decrease in relativistic-electron flux ($L \approx 5$) in the outer radiation belt precedes the establishment of a ring current, the so-called ion-cyclotron resonance, involving an interaction between the relativistic electrons and ring-current, ion-cyclotron waves, could not be entirely responsible for the loss of the relativistic electrons during the recovery phase. In contrast, Chiu et al.⁹³ assumed, in their analysis that the ion-cyclotron resonance was the major loss mechanism for the relativistic electrons. As a consequence, they concluded that since the addition of cold, heavy ions (O^+ and Ar^+) would suppress this resonant interaction, the injection of O^+ and Ar^+ would lead to the enhancement of the relativistic-electron population. If Cladis et al. are correct, then the suppression of resonant interaction would be inconsequential, at least during the recovery phase of a magnetic storm. However, they do not claim that the addition of the Ar^+ plasma from the COTVs is benign. Quite to the contrary, they suggest that a probable consequence of Ar^+ injection would be to increase the inner magnetospheric density and consequently to reduce its susceptibility to generation of magnetic and electric fluctuations. This consequence would tend to prevent the inward migration of relativistic electrons that naturally occurs during magnetic storms. Prevention of such inward migration would cause in turn a reduction of energetic electron fluxes inside the plasmasphere and an enhancement of the fluxes in the extreme outer regions. The ring current, the peak of the outer radiation belt, and the outer boundary of the trapping region would be moved outward by 1 or 2 integer L-values. While they are uncertain about the magnitude of the effects, they suggest that the electron flux at geosynchronous altitude ($L = 6.6$) would be increased by a factor between 1 and 5 by the presence of Ar^+ ions. Such an increase could decrease the lifetime of solar cells correspondingly. Hence, if the lifetime of solar cells in the normal GEO radiation environment is 40 years, it could be reduced to as little as 8 years.

*L is the magnetic-shell parameter. The intersection of a magnetic shell with the equatorial plane forms an approximate circle. The radius of the circle is very nearly equal to L Earth radii. Hence, GEO is located on the magnetic shell corresponding to an L-value of 6.6. It is 6.6 R_E from the center of the Earth. L is also an adiabatic invariant of the motion of a magnetically trapped charged particle.

To recapitulate, based on the assumption that the Ar^+ beam is stopped within the magnetosphere (which leads to a substantial increase in Ar^+ and O^+ density in the plasmasphere) and the assumption that the principal mechanism for removal of relativistic electrons from the outer radiation belt is the ion-cyclotron instability (which is suppressed by the presence of Ar^+ and O^+), Chiu, et al.⁹³ conclude that there would be a buildup of relativistic-electron flux in the outer belt ($4 \lesssim L \lesssim 6$). Cladis et al.,¹²⁸ on the other hand, claim that it is doubtful that the ion-cyclotron instability is important in removing relativistic electrons from the outer belt and that the consequences of adding Ar^+ would be to diminish the electron flux within the inner magnetosphere ($L \lesssim 5$) and increase it outside that region. At GEO ($L = 6.6$) the increase could be as much as a factor of 5. The sensitivity of Cladis et al.'s¹²⁸ results to the manner in which the Ar^+ ions are stopped within the magnetosphere has not been investigated. It is obviously very important to confirm their new findings.

3.4.5.4 Generation of Plasma Turbulence and Interference with Satellite Communication Systems

On one point, there seems to be general agreement, and that is, that the injection of the Ar^+ plasma beam will generate plasma turbulence. In fact, a recent experiment in an ionospheric simulation chamber at the Johnson Space Flight Center reported on by Chiu et al.⁹³ suggests that the Ar^+ plasma beam itself is much more turbulent than previously thought. Curtis and Grebowsky¹²³ suggest that even if only 10% of the energetic Ar^+ ions are trapped in the magnetosphere, they could lead to the generation of a substantial amount of plasma turbulence. While an enhancement of the radiation belts is not expected in the Curtis-Grebowsky picture, as discussed earlier, the generation of plasma turbulence is likely to scatter radiation belt particles out of their stably trapped orbits so that they will precipitate into the atmosphere and produce artificial auroral displays and radiowave interference. In addition, the injection of energetic Ar^+ ions into the ambient plasma predicted by the Curtis-Grebowsky model is likely to drive plasma instabilities that may produce sufficient scintillations to impair satellite-based communication systems. Chiu et al.⁹³ point out that many forms of plasma instabilities may be triggered by the plasma turbulence created by the interaction of the beam with the ambient plasma. In particular, they suggest that the electrostatic modes may generate density irregularities in the form of plasma striations aligned with the geomagnetic field. Such striations have been observed in barium release experiments. The irregularities are of the proper size, say Chiu et al.,⁹³ to be potential scattering centers for satellite communication signals. Such scattering would give rise to signal scintillations in agreement with Curtis and Grebowsky's¹²³ findings. However, the strength of such scintillations cannot be determined at this time. Experiments are required.

3.5 ENERGY DEPOSITION AND ABSORPTION

The construction and operation of the satellite power system would result in the deposition of energy as well as rocket effluent into the different regions of the atmosphere. In addition, a small fraction of the energy in the microwave beam is absorbed by the atmosphere. The question of energy

deposition from the argon plasma engines used by the COTV was discussed previously, as was the possibility of energy absorption from the microwave beam in the troposphere. The question of energy deposition and absorption in the stratosphere, mesosphere, and lower thermosphere is discussed briefly in this section.

One possible source of energy is the total energy content, thermal and streaming energy combined, of the exhaust from a large rocket such as the HLLV. An estimate of the amount of energy involved may be obtained by considering the energy balance for an accelerating rocket. The heat of combustion of the known fuel burned to produce the given mixture of exhaust products provides an estimate of the total energy production rate, from which is subtracted the rate of change of the rocket's potential and kinetic energies. When this calculation is carried out, the total energy deposition into the stratosphere amounts to 7.5×10^{-8} mW/cm² when averaged over the globe. This figure should be compared with the average energy input to the stratosphere due to solar UV absorption of 4 mW/cm². Any local energy effects due to the passage of a large rocket are quickly dissipated by the atmosphere. Similar conclusions apply to the mesosphere and lower thermosphere.

Estimates are available for the upper-atmosphere absorption of microwaves at the SPS frequency of 2.45 GHz. The fractional absorption of microwaves with a frequency of 2.45 GHz vertically incident on the atmosphere has been reported to be approximately 0.01, of which only 0.001 occurs in the stratosphere and mesosphere.¹³¹ The peak power in the microwave beam is expected to be approximately 23 mW/cm² with an average power level of about 5 mW/cm². Thus, a net continuous energy deposition in the stratosphere and mesosphere of approximately 0.005 mW/cm² would be expected per satellite. This figure is about 0.13% of the solar UV absorption input, and may be considered negligible. When averaged over the globe, even assuming the presence of sixty 5-GW rectennas, the average energy flux to the upper atmosphere would be only 6×10^{-5} mW/cm².

Microwave attenuation at 2.45 GHz above the troposphere is due almost entirely to absorption by molecular oxygen,¹³¹ with an absorption cross-section of approximately 1.3×10^{-31} m² (estimated from results presented in Ref. 131). Using this cross section together with data on the variation of atmospheric number density with altitude,¹³² the microwave attenuation values for both the stratosphere and mesosphere have been independently calculated and the results are in agreement with the total value given in Ref. 19 for altitudes above the troposphere. A total absorption rate of 0.0056 mW/cm² (0.14% of the solar input of 4 mW/cm²) is estimated in the stratosphere, and an absorption rate of 0.0001 mW/cm² (0.4% of the corresponding solar input of 0.025 mW/cm²) is estimated in the mesosphere.

These values are small in an absolute as well as a relative sense. The effect of absorption is to heat the air. The total temperature change depends on the length of time that an individual parcel of air spends in the microwave beam, and therefore on the beam diameter and wind speed. Assuming typical wind speeds and residence times and a beam diameter of 10 km, corresponding estimated temperature changes have been calculated and are totally negligible.

3.6 EFFECTS OF LARGE STRUCTURES ON THE SATELLITE ENVIRONMENT

It was necessary to set priorities early in the assessment program in order to limit the number of issues addressed to those judged to be most important. The workshops described earlier provided a considerable list of issues as well as valuable information needed to set priorities. It was the consensus at that time that the rocket effluent effects would be more important than the large structure effects. Hence, little if any new work was performed on the assessment of satellite structure effects on the upper atmosphere. Below we describe some of the more important issues related to structure effects.

Several authors, including Vondrak^{133,134} and Rosen¹³⁵ have suggested that large structures in the magnetosphere will serve as sinks for particles striking them. The COTV, depending upon which photocell option is chosen, will have a cross-sectional area of 1.3 to 2.9 km². Such a structure according to Rosen¹³⁴ could have a sweeping effect on the inner Van Allen Belt particles as it moves back and forth between LEO and GEO. However, the extent of that kind of a sweeping action, and the impact on the space environment is not clear at this time. Presumably it would have a net positive influence through the reduction of ionizing radiation levels.

The 60 SPS satellites will each cover an area of about 55 km² and be distributed along an equatorial line over the U.S. on the magnetic shell defined by an L value of about 6.6. One would expect therefore that flux tubes passing through these satellites and perhaps even the whole region of the magnetic shell could suffer some reduction of the energetic component of the plasma. This reduction, according to Vondrak,^{133,134} would be analogous to the "sweeping" of the Jovian radiation belts by the Galilean satellites.

The satellites are expected to become charged and, consequently, will accelerate ambient charged particles. This activity could lead to alteration of the ambient plasma and to enhanced radiation damage over and above that expected by unaccelerated particle bombardment. (The satellites will reside in the outer portion of the outer Van Allen Belt.) Project SCATHA (Spacecraft Charging at High Altitudes, see Garrett¹³⁶) is relevant to this issue. For example, SCATHA experiments have demonstrated that electric potentials established on spacecraft can accelerate and decelerate plasma components to tens of keV. These experiments have also indicated that klystron operations can be adversely influenced by the plasma environment, which may be modified by gaseous and particulate emissions in GEO.

The SPS satellite structure will dissipate most of the solar input energy as either reflected light or infrared radiation, since the proposed solar cells are expected to be, at most, 17-19% efficient. In addition, the DC-to-RF converters will dissipate approximately 1.2 GW of waste heat, probably at relatively high temperatures (>500°C). The effects of this radiation, as well as the effects of RF radiation on the astronomical community, could be important.¹³⁷

Satellite structural members will be exposed continuously to direct solar and cosmic radiation and are expected to be sources of photoelectrons and other charged and neutral particles. The ejected neutral particles could form relatively long-lived orbiting dust clouds or rings near GEO.^{134,138}

Finally, the structures will be sources of various neutral contaminant gases (as well as Ar^+ ions and electrons from attitude control and station keeping thrusters). Sources of neutral gases include: (a) chemically-fueled attitude control and stationkeeping thrusters used during eclipses of the sun by Earth, (b) leakage from pressurized chambers including living quarters, containment vessels, etc., and (c) satellite-surface erosion and outgassing of volatiles.¹³⁴

In addition to these satellite-structural sources, both chemical and ion propellants will be injected near GEO from thrusters required in space-construction operations and maneuvers. The neutral gases probably will have relatively low exhaust speed and be more randomly directed. Thus, they probably will be trapped and may form gas clouds in the GEO environment. The minimum gas release rates needed to produce significant effects have been evaluated by Vondrak.^{133,134} Garrett and Forbes also have performed some calculations relevant to this topic.¹³⁸ They suggest that it may be possible to measure an actual contaminant cloud (Xe^+ ions) during the 1979-1980 time frame by using the GEOS II satellite.

4 UPPER-LOWER ATMOSPHERIC COUPLING AND IMPLICATIONS FOR SPS IMPACTS ON CLIMATE AND WEATHER

It has already been stressed that as one moves to progressively higher altitudes the magnitude of possible perturbations grows because of the SPS rocket effluents. On the other hand, our understanding of the influence of phenomena in the upper atmosphere on weather-related phenomena in the troposphere diminishes with altitude. Consequently, our present ability to assess SPS impacts of upper atmospheric effects on the weather (short-term, local) or on the climate (long-term, global) is very limited.

Questions regarding coupling of upper and lower atmospheric processes are, of course, much older than the SPS concept, and interest in this subject has only recently been stimulated by concern over the influence that man-made changes in upper atmospheric properties might have on the climate. The most prominent issues have been those of CO₂ and fine-particle emissions by fossil fuel combustion and O₃ removal by NO_x emissions from high altitude aircraft and by aerosol propellants. Somewhat less dramatic, but of equal importance from a basic understanding point of view is the issue of how small changes in the sun's properties, such as exhibited by the sun-spot cycle and changes in the solar wind, are correlated with fluctuations in weather patterns. The so-called solar-terrestrial weather effect was quite thoroughly reviewed at the symposium/workshop held at Ohio State University in 1978.¹¹⁵

Three general mechanisms have been suggested that could couple upper and lower atmospheric processes. First, are the radiative mechanisms that involve changes in upper atmospheric composition and/or temperature that affect, in turn, heating at the surface. Two possible perturbations include change in transmission of visible solar radiation and change in the rate of emission or absorption of IR radiation. Second, are the dynamical mechanisms that involve relationships between upper and lower atmospheric circulation, pressure patterns, and temperature. Third, are those that influence atmospheric electricity. Each of these mechanisms and the implications for SPS are discussed briefly below:

Work on radiative mechanisms has been reviewed by Whitten et al.⁸⁰ They report that 10% increases in stratospheric water and increases of 1% in O₃ concentrations at all heights result in estimated changes of less than 0.1 K. Since SPS changes are predicted to be even smaller than these H₂O and O₃ changes, the corresponding surface temperature changes would be quite negligible. It was at first thought that corridor effects, i.e., buildups of H₂O and NO_x and corresponding reductions in O₃ in narrow latitude bands around the globe might lead to significant changes in surface temperatures. However, according to Whitten et al.'s estimates, the time scale for smoothing out local temperature changes is much shorter than that for buildup of corridors. Furthermore, their estimates indicate that significant corridors of H₂O, NO_x and reduced O₃ levels are unlikely to occur anyway.

Whitten et al.⁸⁰ have also reviewed recent work on dynamical coupling mechanisms. One such mechanism that was suggested by Hines,¹³⁹ involves the propagation of planetary pressure waves around the globe and their association with the large-scale pressure and circulation patterns in the troposphere. These waves are reflected in the upper atmosphere by the natural variation of temperature and pressure with altitude. If the properties that govern the

amplitude and phase of the reflected waves can be influenced by compositional changes in the upper atmosphere, then the pattern of the standing planetary waves may change as well with the consequence of changes in global circulation. Based upon investigations reviewed by Whitten et al., they conclude that the magnitudes of compositional changes they have predicted for the SPS are far below the threshold for significant effects on reflected planetary waves.

Markson^{100,101} has proposed a mechanism to describe the solar-terrestrial weather effect based upon the global electrical circuit that is driven by thunderstorm activity around the globe. Basically, the circuit consists of two equipotential surfaces, namely, the Earth's ground surface and the ionosphere, between which a small globally distributed current density flows in one direction during periods of fair weather and is returned through large local current flows in regions experiencing thunderstorms. Markson suggests that the modulation of ionizing radiation by interaction of the solar wind with the magnetosphere could affect the level of ionization and therefore the electrical conductivity above thunderstorms. The conductivity presumably influences the flow of electrical currents and as a result the nature and intensity of the storm activity. The suggestion that the solar wind can influence storm activity is particularly interesting because it is susceptible to modification by the compositional and dynamical changes of the magnetosphere discussed in Sec. 3.4. It is known, for example, that during magnetic storms, high-energy charged particles are precipitated into the high-latitude atmosphere where they produce ionizing radiation as a result of collisions with the dense neutral atmosphere. The radiation then alters the degree of ionization in the upper atmosphere. However, it is not known how change in ionization influences thunderstorms, nor is it known for certain how particle precipitation will be affected by COTV and POTV emissions.

In summary, based on present knowledge, it does not appear that the SPS will significantly influence tropospheric weather or climate through the upper-lower atmospheric coupling mechanisms discussed above. In view of the developing state of our knowledge regarding these mechanisms, it would be prudent to reexamine the question of large-scale rocket activity from time to time in the future. The atmospheric-electricity phenomenon should be examined more carefully, especially if the magnetospheric effects described in this report are borne out by future studies.

5 CONCLUSIONS

Effluents from the Earth-space transportation system are the major cause of SPS-related atmospheric effects. These effects include inadvertent weather modification, air quality degradation, compositional changes in the stratosphere and mesosphere, formation of noctilucent clouds, plasma density changes, airglow-emission enhancements, and changes in composition and dynamics of the plasmasphere and magnetosphere. In most cases, these effects have been difficult to assess because they involve processes that are incompletely understood by the scientific community or are speculative in nature, or because they involve extrapolations of known effects to unprecedented scales. Therefore, with few exceptions, the results should be regarded as tentative.

HLLV launches have been predicted to have a significant potential for inadvertent weather modifications on a short-term local scale. Under selected meteorological conditions such launches can affect convective patterns, alter cloud populations, and induce detectable precipitation. None of these effects are judged to be serious. Long-term, large-scale effects are unlikely, but have not been examined in detail.

Air quality impacts of HLLV launches are predicted to be very small, except possibly with respect to nitrogen dioxide. If a short-term, air-quality standard is set as anticipated, then ground-level concentrations of NO_2 resulting from rocket launches could exacerbate existing problems. However, the launches by themselves are not expected to cause an exceedance of the expected standard. The production of NO_2 can also lead to slight increases in acidity of precipitation on a local, intermittent basis. It seems unlikely that the enhancement would be great enough to cause significant environmental effects.

Stratospheric CO_2 and H_2O injections are estimated to be negligible from both the ozone depletion and greenhouse effect points of view. The expected fractional increase in the CO_2 concentration is only 1-2 parts in 10^5 . Small predicted reductions (0.01-0.02%) in the total ozone column (and therefore on the incidence of UV radiation at ground level) because of water injections would not be detectable.

The production of NO during reentry of the HLLV second stage is predicted theoretically to lead to an increase of about 0.02% in the globally-averaged, ozone-column density, with latitude-dependent variations of a factor of 2 around this value. The uncertainty in the calculations is such that no increase or even a small decrease is possible.

Repeated injections of H_2O in the mesosphere will result in an increase of about 8% (globally averaged) in the water vapor content at 80 km. Over a narrow region surrounding the launch latitude at the same height, the increase may be as much as 15%. At lower altitudes, the buildup is smaller and no corridor effects are expected. At higher altitudes, preliminary estimates suggest that the buildup in water vapor content will be greater, increasing the globally averaged natural content by as much as a factor of 100 or more at 120 km and above. On the short-term, smaller scale basis associated with individual rocket launches, significant increases in water vapor concentrations are also expected. On the same basis, small-scale ice clouds

(noctilucent clouds) are expected to be formed but are not expected to reach global scales and, therefore, are not expected to cause climatic impacts. Temperature effects, not fully understood at present, may alter the likelihood of cloud formation and possibly increase cloud persistence. Visual displays at low- to mid-latitudes may result near sunrise or sunset.

The injection of water vapor and hydrogen molecules, above 80 km will lead to an enhancement in the upward flux of hydrogen through the thermosphere. It has been estimated that the natural total flux of hydrogen (in all chemical forms) will be approximately doubled. The consequences of such an enhancement have not been examined in detail as yet. Possible consequences include an increase in escape rate of hydrogen to outer space and/or an increase in atmospheric density above 800 km.

Several mechanisms that are not completely independent of each other have been estimated to produce both large positive and negative changes in D- and E-region plasma densities on a short-term local scale. It is not clear how large or persistent the net plasma density changes will be or even what sign they will take. Long-term, global-scale D- and E-region effects appear to be quite small and probably unimportant although the predictions are uncertain at this time.

The picture is much clearer in the F-region. We can predict with considerable confidence the scale and, with somewhat less confidence, the duration, of short-term F-region holes caused by various engine burns of SPS space vehicles. Each POTV injection burn will result in an ionospheric hole 2 to 3 times as large as the continental U.S. HLLV circularization burns will produce holes one-tenth of that size. (POTV and HLLV burns occur once or twice per month and once or twice per day, respectively.) More speculatively, the periodic engine burns may lead to a chronic partial depletion in a ring-shaped global region centered around the launch latitude. Crude estimates suggest a 10% plasma reduction in this region superimposed on periodic, small-scale, but much more complete depletions. Confirmation with more detailed model calculations taking into account global transport processes is required. Probably the major consequences of this partially depleted ring will be perturbations of VLF, HF, and possibly VHF wave propagation. According to present information such perturbations are not expected to be large. However, sufficient uncertainties exist in the current data and theoretical modeling studies to warrant considerable caution in attempting to scale up radio wave propagation effects.

Another impact of potential importance is the enhancement of airglow emissions. The greatest enhancements observed to date are on the order of 10 kiloRayleighs for certain emissions in the visible and near infrared band. The significance of these effects in terms of their impacts on astronomical observations, remote sensing, and satellite-borne surveillance systems should be determined in the future with the aid of the corresponding communities.

The effect of POTV and COTV emissions in the magnetosphere is perhaps least understood at the present time and least supported by observations. This area of investigation could benefit by field and laboratory experiments and further improvements in theory. The masses and energies injected are unquestionably large compared to naturally occurring values and, therefore, cause concern. Some effects are summarized in Table 20. Heating by the argon

Table 20. Satellite Power System Magnetospheric Effects^a

<u>Effect</u>	<u>Cause</u>	<u>Mechanism</u>	<u>System/Activities Impacted</u>
1. Dosage Enhancement of Trapped Relativistic Electrons	O^+ and Ar^+ in magnetosphere due to exhaust and plasmasphere heating	Thermal heavy ions suppress ring-current ion cyclotron turbulence, which keeps electron dosage in balance in natural state	- Space equipment - Modification of human space activity
2. Artificial Ionospheric Current	Ionospheric electric field induced by argon beam	Beam induced Alfvén shocks propagate into ionosphere	- Powerline tripping - Pipeline corrosion (probably unimportant)
3. Modified Auroral Response to Solar Activity	Neutrals and heavy ions in large quantities	Rapid charge-exchange loss of ring-current particles	- May reduce magnetic storm interference with earth and space-based systems
4. Artificial Airglow	3.5 keV argon ions	Direct impact on atmosphere from LEO source	- Interference with optical earth-sensors
5. Plasma Density Disturbance on Small Spatial Scale	Plasma injection	Plasma instabilities	- Signal scintillation for space-based communications

^aFrom Ref. 93.

plasma beam and the addition of cold, heavy ions may lead to enhancement of dosage of trapped relativistic electrons in the outer Van Allen Belt and perhaps in the GEO environment. Production of artificial ionospheric electric currents similar to those associated with naturally occurring magnetic storms could be driven by the magnetosphere-argon-ion-beam interactions. Such currents could result in tripping circuit-breakers in long telephone and power lines and enhance pipeline corrosion. The presence of large quantities of neutrals (from the POTV) could lead to substantial depletion of energetic, trapped, charged particles and modification of auroral response to solar activity. The possibility also exists for the triggering of plasma instabilities that could cause satellite communications interference. Finally, an appreciable amount of airglow emission may be generated, especially near LEO, by the impingement of the dense Ar^+ beam on the thermosphere. Again, the significance of this artificial airglow should be judged by the remote sensing and astronomical communities.

The sun-weather effect that has been widely debated in the recent climate literature, if it is real, is likely to involve a coupling of the upper and lower atmosphere. Several mechanisms have been proposed to explain such a coupling. It has been conjectured that, if indeed modifications in the solar wind can influence climate through some such coupling scheme, then perhaps the SPS upper atmospheric effects that have been discussed herein could disturb the coupling scheme. In that sense, upper atmospheric SPS effects may be causally connected to possible climate effects. However, until the sun-weather effect is placed on a firmer footing, and until the SPS upper atmospheric effects themselves are better understood, the potential effect of SPS on climate and weather cannot be evaluated except in terms of the direct tropospheric effects discussed earlier in this section. Based on

present knowledge, significant influence of SPS-induced upper atmospheric effects on tropospheric weather is regarded unlikely.

One final effect that has been investigated in the current assessment is the inadvertent weather modification likely to be caused by the rectenna structure coupled with its release of waste heat. From the weather and climate points of view, the rectenna seems likely to have effects comparable with those arising from other nonindustrial land-use changes covering the same surface area; that is, local, detectable meteorological effects that probably have little consequence.

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APPENDIX A

ESTIMATE OF THE RATE OF CONVERSION OF NO₂ TO
HNO₃ IN WATER DROPLETS

A possible chemical mechanism that is capable of qualitatively explaining the observed general features of the chemistry of NO₂ and NO in aqueous solution was shown in Fig. 12, reproduced here for convenience as Fig. A.1. In order to use the given mechanism to quantitatively simulate the rate of production of HNO₃ in aqueous solution, reaction rate coefficients for all the reactions shown must be known; this information is not currently available. It is possible, however, to estimate an upper-bound rate of HNO₃ production in water droplets in the presence of gaseous NO₂ and NO, and the purpose of this appendix is to describe how this estimation may be done.

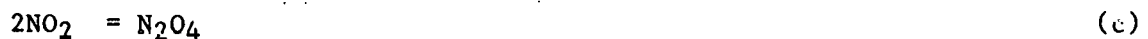
The most significant piece of experimental information available is the initial rate law for the decomposition of HNO₂ in the presence of NO, as determined by Abel and Schmid:^{A.1}

$$-\frac{d[\text{HNO}_2]}{dt} = k_{\text{exp}} \frac{[\text{HNO}_2]^4}{p_{\text{NO}}^2} \quad (\text{A.1})$$

with $k_{\text{exp}}=46$ at 25°C (concentrations in mol/L, pressures in atm, and time in min). The mechanism shown in Fig. A.1. is consistent with this rate law, and gives the following expression for k_{exp} :

$$k_{\text{exp}} = k_d K_a K_b^2 K_c \quad (\text{A.2})$$

This expression is derived by assuming that reactions (a) - (c) below represent rapid equilibria, as indicated in more detail in Fig. A.1, and that reaction (d) is the rate-determining step:



The quantities K_a - K_c in Eq. A.2 represent the equilibrium constants for reactions (a)-(c), respectively, and k_d represents the rate coefficient for reaction (d).

The quantity $K_e = K_a K_b^2 K_c$ represents the equilibrium constant for the net reaction obtained by adding reaction (c) to twice the sum of reactions (a) and (b):



The value of k_d may be estimated if K_e can be estimated, since the value of k_{exp} is known.

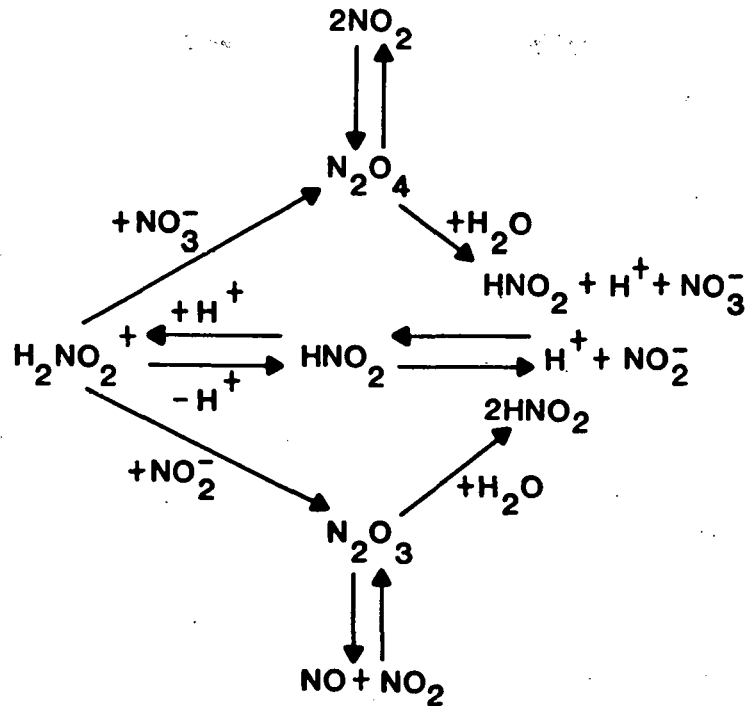


Fig. A.1 Possible Mechanism of the Aqueous Chemistry of Nitrogen Oxides

The value of K_e may be written in terms of the standard free energy change for reaction (e)

$$K_e = \exp(-\Delta G^\circ/RT) \quad (\text{A.3})$$

and the standard free energy change ΔG° may be written, using known free energies of formation^{A.2}

$$\Delta G^\circ = \Delta G_f^\circ(\text{N}_2\text{O}_4(\text{aq})) - 20.66 \text{ kcal} \quad (\text{A.4})$$

and $G_f^\circ(\text{N}_2\text{O}_4(\text{aq}))$ in turn may be expressed as

$$\Delta G_f^\circ(\text{N}_2\text{O}_4(\text{aq})) = -RT \ln K_c + 2\Delta G_f^\circ(\text{NO}_2(\text{aq})) \quad (\text{A.5})$$

A value for $\Delta G_f^\circ(\text{NO}_2(\text{aq}))$ does not seem to be available in the standard tabulations of thermodynamic properties. However, the equilibrium constants for both ozone and CO_2 between gas phases and aqueous phases are available:

$$\frac{[\text{CO}_2]}{P_{\text{CO}_2}} = 0.034 \text{ A.3} \quad (\text{A.6})$$

$$\frac{[\text{O}_3]}{P_{\text{O}_3}} = 0.0123 \text{ A.4} \quad (\text{A.7})$$

Since N lies between C and O in the periodic table, it is perhaps reasonable to estimate the corresponding equilibrium constant (or, equivalently, the standard free energy of formation of dissolved NO_2) by interpolation. This procedure was followed, using the known values of the molecular dipole moments to interpolate between the standard free energies of formation in aqueous solution, with the result that $\Delta G_f^\circ(\text{NO}_2(\text{aq}))$ is estimated to be +14.7 kcal/mol. Substitution into Eq. A.3 yields.

$$K_e = (3.74 \times 10^{-7}) K_c \quad (\text{A.8})$$

It will be convenient to write $K_c = \zeta K_c^{(g)}$ where $K_c^{(g)}$ is the equilibrium constant for reaction (c) but in the gas phase. K_c may be evaluated using known free energies of formation (A.2) and is equal to 8.85 at 25°C. Thus, using Eqs. A.2 and A.8, and the known value of k_{exp} ,

$$k_4 = (1.2 \times 10^7 \text{ min}^{-1}) / \zeta \quad (\text{A.9})$$

The rate of formation of HNO_3 in an aqueous solution of NO_2 may be written

$$\frac{d[\text{HNO}_3]}{dt} = k_4[\text{N}_2\text{O}_4]$$

or

$$\frac{d[\text{HNO}_3]}{dt} = k_4 K_c [\text{NO}_2]^2$$

or, using $K_c = \zeta K_c^{(g)}$,

$$\frac{d[\text{HNO}_3]}{dt} = k_4 \zeta K_c^{(g)} [\text{NO}_2]^2 \quad (\text{A.10})$$

and the numerical value of $k_4 \zeta K_c^{(g)}$ is, using Eq. A.9, the numerical value of $K_c^{(g)}$, and including the proper units, $k_4 \zeta K_c^{(g)} = 1.5 \times 10^6 \text{ mol}^{-1} \text{ s}^{-1}$. Given the rate law of Eq. A.10, the rate of acid formation in droplets in an atmosphere containing NO_2 may be estimated. Notice that if NO is also present in the atmosphere, the rate of HNO_3 production will be retarded, due to the reaction between dissolved NO and NO_2 to form dissolved N_2O_3 and the associated formation of HNO_2 , as shown in Fig. A.1. Thus the presence of NO, by competing for NO_2 , reduces the concentration of NO_2 in solution and, therefore, the rate of HNO_3 formation. Ignoring the presence of NO will, therefore, give an *upper bound* to the rate of HNO_3 formation, and this procedure is implicit in the adoption of Eq. A.10 as the appropriate rate law.

Denoting the atmospheric partial pressure in atmospheres of NO_2 by w , the concentration (mol/L) of unreacted NO_2 in aqueous solution by x and the aqueous concentration (mol/L) of nitric acid by y , the set of differential equations describing the approximate evolution of the system is as follows:

$$\frac{dw}{dt} = -\Lambda(w - \text{HNO}_2 x) \quad (\text{A.11})$$

$$\frac{dx}{dt} = \Lambda \left(\frac{V_{\text{air}}}{V_{\text{liq}}} \cdot \frac{1}{RT} \right) (w - H_{\text{NO}_2} x) - 2kx^2 \quad (\text{A.12})$$

$$\frac{dy}{dt} = kx^2 \quad (\text{A.13})$$

In writing these equations, the assumptions are made that gas-liquid exchange of NO_2 is a simple first-order process with rate constant (i.e., that the droplets are of uniform composition and that gas-phase diffusion is sufficiently fast that it is not a controlling factor), and that the only important chemical processes occurring are the rapid equilibrium of NO_2 and N_2O_4 in solution and the reaction of N_2O_4 with water to produce HNO_2 and HNO_3 . The quantity H_{NO_2} is the Henry's law constant for NO_2 (obtainable from ΔG_f° for NO_2 (aq) and NO_2 (gas)), and the rate constant for NO_2 chemical removal is written $2k$ since NO_2 disappears twice as fast as HNO_3 is produced, the extra NO_2 going into the production of HNO_2 . It is also assumed that HNO_3 remains in solution.

In order to solve analytically the x and y equations, the partial pressure of NO_2 (w) is kept fixed at its initial value. The solutions of Eqs. A.12 and A.13 are then found to be:

$$x(t) = \frac{\omega_- + \omega_+ a \exp(-\lambda t)}{1 - a \exp(-\lambda t)} \quad (\text{A.14})$$

in which

$$\omega_{\pm} = \frac{\lambda + \beta}{4k} \quad (\text{A.15})$$

$$\lambda = (\beta^2 + 8\alpha k)^{1/2} \quad (\text{A.16})$$

$$\alpha = \Lambda \left(\frac{V_{\text{air}}}{V_{\text{liq}}} \cdot \frac{1}{RT} \right) P_{\text{NO}_2} \quad (\text{A.17})$$

$$\beta = \Lambda \left(\frac{V_{\text{air}}}{V_{\text{liq}}} \cdot \frac{1}{RT} \right) H_{\text{NO}_2} \quad (\text{A.18})$$

$$a = \frac{x(0) - \omega_-}{x(0) + \omega_+} \quad (\text{A.19})$$

and

$$y(t) = y(0) + k \left[\omega_-^2 t + \frac{1}{\lambda} (\omega_- + \omega_+)^2 A + (\omega_+^2 - \omega_-^2) \ln B \right] \quad (\text{A.20})$$

with

$$A(t) = \frac{1}{1-a} - \frac{1}{1-a \exp(-\lambda t)} \quad (\text{A.21})$$

and

$$B(t) = \frac{1 - a}{1 - a \exp(-\lambda t)} \quad (\text{A.22})$$

Inserting the following numerical values:

$$H_{\text{NO}_2} = 52 \text{ L atm/mol (from } \Delta G_f^\circ (\text{NO}_2 \text{ (aq)}) = 14.7 \text{ kcal/mol)}$$

$$P_{\text{NO}_2} = 5 \times 10^{-7} \text{ atm (500 ppb)}$$

$$RT = 24.46 \text{ L atm mol}^{-1} \text{ (at } ^\circ 25 \text{ C)}$$

$$k = 1.5 \times 10^6 \text{ L mol}^{-1} \text{ s}^{-1}$$

$$\left(\frac{V_{\text{air}}}{V_{\text{liq}}} \right) = 1.8 \times 10^5 \text{ L/L}$$

$$\Lambda = 8.7 \times 10^{-4} \text{ s}^{-1} \text{ A.5}$$

and

$$x(0) = 0$$

one finds that $\lambda = 3.3 \times 10^2 \text{ s}^{-1}$; thus the NO_2 gas-liquid equilibrium is reached very rapidly and, accordingly, for times longer than a small fraction of a second, $x = \omega_- = \alpha/\beta$

or

$$[\text{NO}_2] = x = P_{\text{NO}_2}/H_{\text{NO}_2} = 9.6 \times 10^{-9} \text{ mol/L} \quad (\text{A.23})$$

Finally, one finds that

$$[\text{HNO}_3] = y = y(0) + k \left(\frac{P_{\text{NO}_2}}{H_{\text{NO}_2}} \right)^2 t \quad (\text{A.24})$$

$$= y(0) + (1.4 \times 10^{-10} \text{ mol L}^{-1} \text{ sec}^{-1})t \quad (\text{A.25})$$

This equation represents an upper bound to the nitric acid concentration in water droplets (for the specified volume of water per unit volume of air) exposed to 500 ppbv NO_2 and a somewhat larger NO concentration. The result is an upper bound due to (1) omission of the aqueous chemical effects of NO ; (2) the assumption of constant P_{NO_2} ; and (3) the assumption that all HNO_3 produced remains in solution.

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